A1-F18AC-SCM-000

1 September 1995

Change 16 - 15 May 2003

TECHNICAL MANUAL

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

NAVY MODEL

F/A-18A/B/C/D

161353 AND UP

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NATEC ELECTRONIC MANUAL

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Only those work packages/pages assigned to the manual are listed in this index. Insert Change 16, dated 15 May 2003. Dispose of superseded work packages/pages. Superseded classified work packages/pages shall be destroyed in accordance with applicable security regulations. If changed pages are issued to a work package, insert the changed pages in the applicable work package. The portion of text affected in a change or revision is indicated by change bars or the change symbol "R" in the outer margin of each column of text. Changes to illustrations are indicated by pointing hands, change bars, or MAJOR CHANGE symbols. Changes to diagrams may be indicated by shaded borders.

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ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

This WP supersedes TPDR WP, dated 1 November 2002.

1. The TPDRs listed below have been incorporated in this issue.

IDENTIFICATION NUMBER/ QA SEQUENCE NUMBER	LOCATION
None	

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ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

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WORK PACKAGE INDEX

ORGANIZATIONAL MAINTENANCE

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This WP supersedes WP001 01, dated 1 April 1997.

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INTRODUCTION

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

This WP supersedes WP002 00, dated 1 January 2000.

1. PURPOSE.

2. This manual provides the data required by the technician to do testing and troubleshooting of the system.

3. REQUISITION AND AUTOMATIC DISTRIBUTION OF NAVAIR TECHNICAL MANUALS.

4. Procedures to be used by Naval activities and other Department of Defense activities requiring NAVAIR technical manuals are defined in NAVAIR 00-25-100 and NAVAIRINST 5605.5.4A.

5. To automatically receive future changes and revisions to NAVAIR technical manuals, an activity must be established on the Automatic Distribution Requirements List (ADRL) maintained by the Naval Air Technical Data and Engineering Service Command (NATEC). To become established on the ADRL, contact your activity central technical publications librarian. If your activity does not have a library, you may establish your automatic distribution by contacting the Commanding Officer, NATEC, Attn: Distribution, NAS North Island, Bldg. 90, P.O. Box 357031, San Diego CA 92135-7031. Reconfirmation of these requirements is necessary once a year to remain on automatic distribution. Please use your NATEC assigned account number when referring to automatic distribution requirements.

6. If added or replacement copies of this manual are required with no attendant changes in the ADRL, they may be ordered by submitting a MILSTRIP requisition in accordance with NAVSUP 485 to Routing Identifier Code "NFZ". MILSTRIP requisitions can be submitted through your supply office, Navy message, or SALTS to DAAS (Defense Automated Address System), or through the DAAS or NAVSUP web sites. For assistance with a MILSTRIP requisition, contact the Naval Inventory Control Point (NAVICP) Publications and Forms Customer Service at DSN 442-2626 or (215) 697-2626, Monday through Friday, 0700 to 1600 Eastern Time.

7. MANUAL ISSUE DATE.

8. The date on the title page is the copy freeze date. No additions, deletions, or changes are made after the manual issue date except last minute safety of flight or required maintenance changes. Data collected after the manual issue date will be included in later changes or revisions of the manual.

9. EFFECTIVITIES.

10. Effectivity notes on manual title pages, work package title pages, and within a work package indicate the aircraft or software program to which the data applies. If no effectivity note appears on the work package title page, the work package has the same effectivity as shown on the manual title page. The effectivity notes may use:

a. Type, model, and series

NOTE

F/A-18D aircraft after bureau number 164967 was referred to as bureau number F/A-18D D-140. Now, F/A-18D aircraft after bureau number 164967 is 165409.

b. Bureau number (tail number)

c. Combination of type, model, series, and bureau numbers

d. Part number or serial number

e. Technical directive number

f. Configuration/identification number

11. The table below shows examples of effectivity notes and their meanings:

Effectivity Note Examples

Effectivity Note	Definition
160777 AND UP	Applicable to all F/A-18A, F/A-18B, F/A-18C and F/A-18D for bureau numbers listed.
F/A-18A, F/A-18B	Applicable to all F/A-18A and F/A-18B.
F/A-18C, F/A-18D	Applicable to all F/A-18C and F/A-18D.
F/A-18A	Applicable to all F/A-18A, but not F/A-18B, F/A-18C and F/A-18D.
F/A-18B	Applicable to all F/A-18B, but not F/A-18A, F/A-18C, and F/A-18D.
F/A-18C	Applicable to all F/A-18C, but not F/A-18A, F/A-18B, and F/A-18D.
F/A-18D	Applicable to all F/A-18D, but not F/A-18A, F/A-18B, and F/A-18C.
F/A-18A, F/A-18C	Applicable to all F/A-18A and F/A-18C, but not to F/A-18B and F/A-18D.
F/A-18B, F/A-18D	Applicable to all F/A-18B and F/A-18D, but not to F/A-18A and F/A-18C.
F/A-18A 160775, 160777 THRU 160782	Only applicable to some bureau numbers of F/A-18A. Not applicable to any F/A-18B, even if a F/A-18B bureau number is within the numbers listed.
F/A-18C 163427, 163430 THRU 163456	Only applicable to some bureau numbers of F/A-18C. Not applicable to any F/A-18D, even if a F/A-18D bureau number is within the numbers listed.
F/A-18B 160784 AND UP	Only applicable to some bureau numbers of F/A-18B. Not applicable to any F/A-18A, even if an F/A-18A bureau number is within the numbers listed.
F/A-18D 163434 THRU 163457	Only applicable to some bureau numbers of F/A-18D. Not applicable to any F/A-18C, even if a F/A-18C bureau number is within the numbers listed.

Effectivity Note Examples (Continued)

Effectivity Note	Definition
F/A-18B 160784 AND UP, F/A-18D	Applicable to some bureau numbers of F/A-18B. Not applicable to any F/A-18A, even if an F/A-18A bureau number is within the numbers listed. Also applicable to all F/A-18D aircraft.
F/A-18C, F/A-18D 163434 THRU 163457	Applicable to all F/A-18C aircraft. Applicable to some bureau numbers of F/A-18D.
F/A-18D D-140 AND UP OR F/A-18D 165409 AND UP	Applicable to all F/A-18D aircraft after bureau number 164967.
160775 THRU 160785 BEFORE F/A-18 AFC 772	Applicable to F/A-18A and F/A-18B for bureau numbers listed, before modification by technical directive.
161213 AND UP; ALSO 160775 THRU 160785 AFTER F/A-18 AFC 772	Applicable to aircraft modified during production; also applicable when affected aircraft have been modified by technical directive.
160775 THRU 160785; WHEN NO. 2 CONTROL PANEL P/N XXXX-X IS INSTALLED	Applicable to F/A-18A and F/A-18B for bureau numbers listed if panel P/N XXXX-X is installed. (Configuration before AVC)
161213 AND UP; ALSO 160775 THRU 160785; WHEN NO. 2 CONTROL PANEL P/N XXXX-Y (AVC-102) IS INSTALLED	Applicable to aircraft modified during production; also applicable to aircraft components modified to the production configuration by technical directive. (Configuration after AVC)
P/N MBEU65101-9, MBEU65101-10 & MBEU65105-3	Applicable to assemblies which are interchangeable between aircraft.
ENGINE NO. 215101 THRU 215109	Applicable to assemblies which are interchangeable between aircraft, but configurations can not be identified by part number.
CONFIG/IDENT NUMBER 84A	The CONFIG/IDENT Number is the program load identification number which identifies the software program loaded in specific programmable units. Refer to A1-F18AC-SCM-000 for CONFIG/IDENT Number tables.

12. **TECHNICAL DIRECTIVES.**

13. Technical directives are documents which provide instructions to add and record retrofit configuration modifications or inspection instructions to delivered aircraft, or aircraft components.

14. **AIRFRAME CHANGE (AFC) AND AIRBORNE SOFTWARE CHANGE (ASC) .** Technical directives which change configuration of aircraft structure or equipment installation, i.e. AFC, will list aircraft bureau numbers in effectivity notes and show before and after the AFC. Technical directives which change configuration of operational flight programs (OFP), i.e. ASC, will list the OFP CONFIG/IDENT NUMBER in effectivity notes and show the latest two authorized OFP programs. See AFC and ASC effectivity examples in Effectivity Note Example Table.

15. **AIRCRAFT COMPONENT CHANGES.** Technical directives which change configuration of aircraft components are listed below:

AAC	Aviation Armament Change for armament equipment
ACC	Aircrew System Change for aircrew survival equipment
AFC	Airframe Change for aircraft structure and equipment
ASC	Airborne Software Change for operational flight programs
AVC	Avionics Change for airborne electronic equipment, including wiring changes
AYC	Accessory Change for mechanical system
PPC	Power Plant Change for engines

16. Component changes will list part numbers in the effectivities. See AVC effectivity examples in Effectivity Note Example table.

17. RECORD OF APPLICABLE TECHNICAL DIRECTIVES.

18. The technical directives affecting this manual are listed in the Record of Applicable Technical Directives of each affected work package. Because an ASC directs all aircraft be modified within 30 days, ASC's are not listed. When all affected aircraft are modified, the before configuration is removed from the manual, and the technical directive entry is removed from the Record of Applicable Technical Directives.

19. TECHNICAL PUBLICATIONS DEFICIENCY REPORT (TPDR).

20. The TPDR (OPNAV FORM 4790/66) is the form for reporting errors and suspected omissions in the technical manuals. The TPDR WP lists the TPDRs that are included in the current issue of the manual.

21. TPDR reporting procedures are in OPNAVINST 4790.2 SERIES.

22. QUALITY ASSURANCE PROCEDURES.

23. Procedures or parts of procedures which require quality assurance inspection are identified by the letters (QA) after the applicable steps. When (QA) is assigned to a step or a heading which is immediately followed by substeps, the inspection requirement is applicable to all substeps.

24. When doing maintenance in any area, a visual inspection of the area will be made for cracks, corrosion and security of component installation before securing the area for flight.

25. TEST PROCEDURES.

26. Test procedures are done as part of malfunction isolation, during periodic inspection, or when correct system operation is to be verified.

27. Satisfactory completion of test procedures verifies correct system operation. Do steps in sequence. When doing system test procedures, make sure:

a. System Required Components identified in procedure are installed.

b. Related Systems Required identified in procedure are operative.

c. Steps are done in sequence.

d. Results are as shown in Normal Indication column, or do Remedy for Abnormal Indication.

e. Each malfunction is corrected before going to next step by repeating portion of test procedure which failed.

28. TROUBLESHOOTING.

29. **TROUBLESHOOTING PROCEDURES.** These procedures provide a series of steps with a NO-YES column. These steps lead to corrective action for the malfunction. Troubleshooting procedures list the data below for use as an aid when doing procedural steps:

a. Reference to a system schematic.

b. Reference to a component locator.

c. List of support equipment and materials required which will always be used in the procedure. Additional support equipment may be required.

d. An alphabetical list of components which could cause the malfunction.

30. Troubleshooting procedures (logic trees) are referenced from a test procedure Remedy for

Abnormal Indication column or from Fault Reporting Manual. Logic trees are written assuming the logic below:

a. If doing a test procedure, all steps testing functions before the failed step had normal indication.

b. For an abnormal indication, only one malfunction exists.

c. All replacement components are ready for installation.

31. **CONTINUITY TESTING.** When doing continuity tests during troubleshooting, the items listed below must be tested, as applicable.

a. Loose electrical connectors and bent, broken, or recessed pins.

b. Continuity between specific pins per procedural step or system schematic.

c. Shorts between conductor and shield.

d. Shorts between conductor and surrounding pins on connectors.

e. Shield continuity per diagrams/system schematics.

32. **TROUBLESHOOTING BEYOND BIT/SYSTEM TESTING.** This is required when any of the conditions listed below exist:

a. Malfunction was not detected by Built-In Test (BIT).

b. Malfunction was not detected by a functional test procedure.

c. When a troubleshooting procedure did not correct the malfunction.

d. When a troubleshooting procedure does not exist.

33. When any of the conditions listed in paragraph 28 exist, troubleshooting procedure/logic must then be determined. Use steps listed below to aid in determining procedure/logic:

a. Use referenced system schematic or select applicable system schematic for malfunction. Use schematic for troubleshooting beyond BIT analysis as listed below:

(1) Analyze interface of system components. Determine logic wiring and/or components which may cause the malfunction. Determine when an interfacing component could cause the malfunction.

(2) When malfunction can be caused by mission computer system signal interface, analyze mission computer system integrated functions and memory inspect suspected Input/Output REF CODES (A1-F18AC-FIM-100).

b. Review VIDS/MAF (OPNAV 4790/60) in Aircraft Discrepancy Book for related malfunctions.

(1) Analyze system/related system maintenance codes reported by Nose Wheelwell Digital Display Indicator.

(2) Determine if aircraft components that have been replaced could cause malfunction.

(3) When a repeat malfunction exists, analyze previous maintenance action completed for the malfunction.

(a) When component replacement is/was done, analyze component history as listed:

1) Determine where component came from.

2) Determine previous history of component (when available).

3) Determine if similar malfunction occurred on another aircraft.

4) Determine if replaced component could be causing existing malfunction.

5) Determine if replacing component again would correct malfunction.

(b) Determine if any rigging or control procedures that have been done could cause the malfunction.

(c) Determine when rigging/boresight procedures should be done to verify system operation for malfunction.

34. **TROUBLESHOOTING IMPROVEMENTS.** When a troubleshooting procedure did not correct a

malfunction and it is determined that additional or new troubleshooting is required, submit Technical Publications Deficiency Report (TPDR) providing the information listed below:

a. Fault descriptor for A1-F18( )-FRM-000.

b. Corrective action taken for malfunction.

c. Logic used to isolate malfunction.

d. Probable changes that could shorten troubleshooting time for malfunction.

35. DIAGRAMS.

36. System schematics are in A1-F18A( )-( )-500 series manuals.

37. ILLUSTRATED PARTS BREAKDOWN.

38. Each illustrated parts breakdown (IPB) in this manual has a parts list and illustration for the requisition, storage, authority for use and identification of parts. The illustration is integrated with, and supports, both the maintenance procedure and the parts list within each work package.

39. **PART NUMBER COLUMN.** Footnote symbols in the part number column are defined following the last part listed in each parts list (also see converted part numbers, this WP).

40. **INDENTION.** The first entry in the description column of each parts list is the figure title. This figure title identifies the parts list with the related maintenance procedure and is shown in the first indent. All parts data required to support the specific maintenance procedure is below the figure title in the second indent.

41. **COMMON NAMES.** The official nomenclature in the description column may not be the name commonly used for an item. If different from the official nomenclature, the common name is shown in parentheses in the description column immediately following the official nomenclature.

42. **COMMERCIAL AND GOVERNMENT ENTITY CODES.** Entity code or manufacturer's name and address are shown in the Description column in parentheses after the nomenclature for the item. These codes are per the Commercial and Government Entity (CAGE) Handbook H4/H8 Series. No code indicates the item is a government standard part.

43. **ATTACHING PARTS.** Attaching parts are identified by (AP) after the nomenclature of the item in the description column. Attaching parts are listed immediately following the part they attach.

44. **SPECIAL HANDLING.** Items requiring special handling; for example liquid oxygen components, magnetic control items or on-board oxygen generating system (OBOGS) are identified by the acronym LOX for liquid oxygen, MAG for magnetic control items and OXYGEN for on-board oxygen generating system (OBOGS) in the Description column, at the extreme right side.

45. **CONVERTED PART NUMBERS.** Some part numbers appear in the Part Number column which are different than the manufacturer's part number. These are converted part numbers. The unconverted manufacturer's part number is shown in the Description column following the manufacturer's code. Always use the part number in the Part Number column when ordering parts. If an item is not available under the listing in the Part Number column, it may be ordered using the unconverted part number found in the Description column or by using the number found on the part. Examples of special characters as they may appear in the Part Number and Description columns are shown below:

Part Number Column	Description Column
PORM	± (Plus or Minus)
DEG	° (Degree)
E	e (Lower case letter)
2	II (Roman Numeral)
0.001	.001 (Decimal)

46. **SUPERSEDED PARTS.** Superseded part numbers have been removed from the Part Number column and placed in the Description column of the superseding part (for example - supersedes 74A582090-1003). This indicates that the superseded part is usable if available through salvage, but should not be requisitioned or made.

47. **NEXT HIGHER ASSEMBLY.** Next higher assembly (NHA) data is not shown using indention. Next higher procurable assembly (NHPA) data is shown for part numbers that have a procurable

NHA. The NHPA and its assigned Source, Maintenance and Recoverability (SM&R) code are in parentheses as the last entry in the Description column. Requisition the NHPA when the part listed in the Part Number column is not available from supply. The components of assemblies that required disassembly during removal from aircraft, are footnoted in the part number column.

48. **UNITS PER ASSEMBLY (UPA) COLUMN.** This column lists the total number of each part required per assembly or subassembly and are not necessarily the total number used in the end item of equipment. The letters AR (As Required) are used for items; for example shims, when the requirement may vary.

49. **USABLE-ON CODES.** Applicable usable-on codes are identified on the final sheet of each parts list. No entry in the Use On column indicates parts are applicable to all configurations supported by this parts list.

50. **ALTERNATE OR EQUIVALENT PARTS.** An asterisk (*), in the Use On column, identifies alternate parts or equivalent parts that are interchangeable. When a letter code is followed by an asterisk in the Use On column, only the parts with the same letter code are interchangeable. An alternate part may be used when preferred part is not available. The asterisk is omitted for the preferred part(s). Equivalent parts are fully interchangeable. No equivalent part is preferred over another. All equivalent parts are identified by asterisks.

51. **SOURCE, MAINTENANCE AND RECOVERABILITY (SM&R) CODE COLUMN.** The codes used in this column are assigned per NAVSUPINST 4423.29 SERIES which contain definitions. A dash (-) is shown in the SM&R code column when no code has been assigned. The Aviation Supply Office P2300 series publication is to be used for the most current SM&R Code assignment information if the validity of any SM&R Code listed in an IPB is suspect. Refer to figure 1 for SM&R code explanations.

52. **PARTS LIST INDEX MANUAL, A1-F18AC-IPB-450.** This manual has a numerical index of part numbers and a reference designation index for use with aircraft organizational maintenance manuals. When reference designations or part numbers are known, the index locates specific maintenance instructions and parts data.

53. NAVY (AN) STANDARD/COMMON NAME NOMENCLATURE.

54. When an item has both Navy (AN) standard and common name nomenclature assigned, the common name nomenclature will be used in text and on illustrations. Full Navy (AN) standard nomenclature will be used in the Illustrated Parts Breakdown (IPB).

SOURCE			MAINTENANCE			
1ST POS	2ND POSITION		3RD POSITION		4TH POSITION	
MEANS OF ACQUIRING SUPPORT			USE: LOWEST LEVEL AUTHORIZED TO REMOVE/REPLACE THE ITEM.		REPAIR: LOWEST LEVEL WITH CAPABILITY AND RESOURCES TO PERFORM COMPLETE REPAIR ACTION	
P	A	ITEM: STOCKED	O	ORG/UNIT	O	ORG/UNIT
	B	ITEM: STOCKED, INSURANCE	2	MINESWEEPER	2	MINESWEEPER
	C	ITEM: STOCKED, DETERIORATIVE	3	SUBMARINES	3	SUBMARINES
	D	ITEM: SUPPORT, INITIAL ISSUE OF OUTFITTING & STOCKED ONLY FOR ADDITIONAL INITIAL ISSUE	4	AUX/AMPHIB	4	AUX/AMPHIB
	E	EQUIPMENT: SUPPORT, STOCKED FOR INITIAL ISSUE OR OUTFITTING OF SPECIFIED MAINTENANCE ACTIVITIES	5	DESTROYER, FFG	5	DESTROYER, FFG
	F	EQUIPMENT: SUPPORT, NON-STOCKED, CENTRALLY PROCURED ON DEMAND	6	CRUISER/CARRIER	6	CRUISER/CARRIER
	G	ITEM: STOCKED FOR SUSTAINED SUPPORT. UNECONOMICAL TO PRODUCE AT A LATER TIME	F	I/AFLOAT	F	I/AFLOAT
	H	ITEM: STOCKED, CONTAINS HAZMAT. HMIS/MSDS REPORTING REQUIRED				
	R	TERMINAL OR OBSOLETE, REPLACED			G	I/ASHORE AND AFLOAT
	Z	TERMINAL OR OBSOLETE, NOT REPLACED	G	I/ASHORE AND AFLOAT		
K	D	ITEM: DEPOT O/H & MAINTENANCE KITS				
	F	ITEM: MAINTENANCE KIT, PLACE AT O,F,H,L			H	I/ASHORE
	B	ITEM: IN BOTH DEPOT REPAIR & MAINT. KITS				
M	O	MFR OR FAB AT UNIT LEVEL	H	I/ASHORE		
	F	MFR OR FAB AT INTERMEDIATE/DS LEVEL			K	CONTRACTOR FACILITY
	H	MFR OR FAB AT INTERMEDIATE/GS LEVEL				
	L	MFR OR FAB AT SPECIALIZED REPAIR ACTIVITY (SRA)				
	G	MFR OR FAB AT INTERMEDIATE BOTH AFLOAT AND ASHORE	K	CONTRACTOR FACILITY	L	INTERMEDIATE SRA
	D	MFR OR FAB AT DEPOT MAINTENANCE LEVEL				
A	O	ITEM: ASSEMBLED AT ORG/UNIT				
	F	ITEM: ASSEMBLED AT INTERMEDIATE LEVEL - AFLOAT			D	DEPOT
	H	ITEM: ASSEMBLED AT INTERMEDIATE LEVEL - ASHORE	L	INTERMEDIATE SRA		
	L	ITEM: ASSEMBLED AT SRA			Z	NON-REPAIRABLE
	G	ITEM: ASSEMBLED AT INTERMEDIATE BOTH AFLOAT AND ASHORE				
	D	ITEM: ASSEMBLED AT DEPOT MAINTENANCE LEVEL				
X	A	ITEM: REQUISITION NEXT HIGHER ASSEMBLY	D	DEPOT	B	RECONDITION
	B	ITEM: NOT PROCURED OR STOCKED. AVAILABLE THRU SALVAGE. REQ. BY CAGE/PART NUMBER				
	C	INSTALLATION DRAWING, DIAGRAM, INSTRUCTION SHEET. IDENTIFY BY CAGE/PART NUMBER	Z	REF ONLY		
	D	NON-STOCKED. OBTAIN VIA LOCAL PURCHASE				

RECOVERABILITY		SERVICE OPTION	
5th POSITION		6th POSITION	
DISPOSITION: WHEN UNSERVICEABLE OR UNECONOMICALLY REPAIRABLE, CONDEMN OR DISPOSE.		ASSIGNED TO SUPPORT ITEMS TO CONVEY SPECIFIC INFORMATION TO THE SERVICE'S LOGISTICS COMMUNITY/OPERATING FORCES.	
O	ORG/UNIT	1	I-LEVEL 1ST DEGREE
F	I/AFLOAT	2	I-LEVEL 2ND DEGREE
G	I/ASHORE AND AFLOAT	3	I-LEVEL 3RD DEGREE
H	I/ASHORE	6	COMMERCIAL ITEM, ORGANICALLY MFR'D
K	DLR; CONTRACTOR FACILITY	8	NON-CONSUMABLE; 2ND DEGREE ENGINE I-LEVEL
L	INTERMEDIATE SRA LEVEL	9	NON-CONSUMABLE; 3RD DEGREE ENGINE I-LEVEL
D	DLR; CONDEMN OR DISPOSE AT DEPOT	E	END TO END TEST
		J	INTER-SERVICE DLR REPAIRABLE BELOW D-LEVEL
Z	NON-REPAIRABLE	P	PROGRESSIVE MAINTENANCE
		R	GOLD DISC REPAIR
A	NON-REPAIRABLE BUT REQUIRES SPECIAL HANDLING	T	TRAINING DEVICES

Figure 1. SM&R Code Explanation

ORGANIZATIONAL MAINTENANCE

HOW TO USE MANUAL

SOFTWARE CONFIGURATION MANUAL

This WP supersedes WP 002 01, dated 1 September 1995.

Reference Material

None

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Subject	Page No.
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Purpose	1

Record of Applicable Technical Directives

None

1. PURPOSE.

2. This manual provides data required to maintain software configuration and compatibility.

3. CONTENT.

4. This manual contains data required to identify software configurations and procedures for loading operational flight programs.

5. **PROGRAM LOAD VERSIONS.** Work package 003 00 documents aircraft software configuration by listing each applicable software version. Table 1 the table of use and applicability provides a historical record of the specific software version number for each programmable unit and related aircraft. Table 2 the current program load configurations provides a list of all currently applicable software versions for each programmable unit. The software version is identified by a program part number. Programmable units that display CONFIG/IDENT status on the aircraft displays also have two additional numbers which are used to determine the software version. These two numbers are the program load CONFIG/IDENT number and unit part number.

6. **PROGRAM LOAD CONFIG/IDENT NUMBER.** On programmable units with a program load CONFIG/IDENT number listed in table of use and applicability in WP003 00, the program load CONFIG/IDENT number can be called up and displayed on a Digital Display Indicator. The procedures for calling up the program load CONFIG/IDENT number are contained in WP004 00. On all programmable units, the software version can be determined by the program part number identified on the unit decal. Troubleshooting for program load status wrong or MC CONFIG caution displayed is also contained in WP004 00.

7. **COMPONENT LOCATOR.** WP005 00 contains component location information used with procedures in this manual.

8. **LOAD/VERIFICATION PROCEDURES.** Procedures for loading programs in the systems listed are contained in WP006 00.

a. Digital Data Computers No. 1 and No. 2 (MC1 and MC2)
b. Armament Computer CP-1342/AYQ-9(V) (SMS)
c. Command Launch Computer CP-1001( )/AWG (CLC)
d. ON F/A-18C AND F/A-18D Signal Data Computer CP-1726/ASQ-194 (SDC)
f. Computer-Power Supply CP-1325/APG-65 (CPS)
g. Control Converter C-10382/A (CSC)
h. Air Data Computer CP-1334( )/A (ADC)
i. Digital Map Computer CP-1802/ASQ-196 (DMC)
j. Data Transfer Interface Unit J-6008/A (DFIRS)
k. Radar Data Processor CP-2062/APG-73
l. Countermeasures Computer (RWR) AN/ALE-67
m. Countermeasures Dispenser AN/ALR-47
n. Airborne Self protect Jammer (ASPJ) AN/ALQ-126B
o. Countermeasures Set AN/ALQ-165

9. HOW TO USE MANUAL.

10. This manual is used to identify aircraft software configurations and programmable unit software versions, load software programs and to troubleshoot program load status wrong and MC CONFIG caution.

11. **IDENTIFYING SOFTWARE VERSIONS AND CONFIGURATIONS.** Identification of a software version is done by:

a. Comparing program part numbers on programmable units to program part numbers listed in table of use and applicability in WP003 00.

b. Comparing program load CONFIG/IDENT numbers, obtained using procedures in WP004 00, with program load CONFIG/IDENT numbers listed in table of use and applicability in WP003 00.

12. Aircraft software configuration is identified by comparing software version numbers with aircraft effectivities listed in table of use and applicability in WP003 00.

13. **PROGRAM LOADING.** Program loading of the systems listed is done by using the procedures contained in WP006 00.

a. Digital Data Computers No. 1 and No. 2 (MC1 and MC2)
b. Armament Computer CP-1342/AYQ-9(V) (SMS)
c. Command Launch Computer CP-1001( )/AWG (CLC)
d. ON F/A-18C AND F/A-18D, Signal Data Computer CP-1726/ASQ-194 (SDC)
f. Computer-Power Supply CP-1325/APG-65 (CPS)
g. Control Converter C-10382/A (CSC)
h. Air Data Computer CP-1334( )/A (ADC)
i. Digital Map Computer CP-1802/ASQ-196 (DMC)
j. Data Transfer Interface Unit J-6008/A (DFIRS)
k. Radar Data Processor CP-2062/APG-73
l. Countermeasures Computer AN/ALR-67
m. Countermeasures Dispenser AN/ALR-47
n. Airborne Self Protect Jammer AN/ALQ-126B
o. Countermeasures Set AN/ALQ-165

14. **Test Equipment Hookup.** Test equipment hookup and locator information is in WP007 00.

15. **TROUBLESHOOTING.** Troubleshooting for program load status wrong or MC CONFIG caution displayed is done using procedures contained in WP004 00 and the mux test connector 83J-G003 interconnect schematic WP008 00.

16. ABBREVIATIONS.

AC	armament computer
ADC	air data computer
AIC	intercommunication amplifier-control
AMLV	advanced memory loader verifier
ASPJ	airborne self protect jammer
CLC	command launch computer
CPS	controller processor section (FLIR)
CPS	computer power supply (radar)
CSC	control-converter
DCC	digital computer converter

DDI	digital display indicator (multipurpose display indicator)
DFIRS	deployable fault incident recording system
ECM	electronic countermeasures
DMC	digital map computer
FCC	flight control computer (RPYC)
GPS	global positioning system
INU	inertial navigation unit
MC	mission computer (digital data computer)
MDI	multipurpose display indicator
MLV	memory loader verifier
MLVS	memory loader-verifier set
MU	memory unit
MUX	multiplex bus
OFP	operational flight program
RDP	radar data processor
RPYC	roll pitch yaw computer (FCC)
RWR	radar warning receiver
SDC	signal data computer
SMS	armament computer

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

PROGRAM LOAD VERSIONS

Title	WP Number
Program Load Versions- F/A-18A AND F/A-18B	003 01
Program Load Versions- F/A-18C AND F/A-18D	003 02

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

PROGRAM LOAD VERSIONS

EFFECTIVITY: F/A-18A AND F/A-18B

Reference Material

Software Configuration Manual A1-F18AC-SCM-000
Program Load CONFIG/IDENT Verification WP004 00

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Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F18-MD-0187	-	Armament Computer, Inertial Nav Set, Mission Computer, Radar Computer Power Supply, Reloading of	1 Oct 81	-
F18 IASC 001	-	SMS OFP Change to Prevent Erroneous Double Weapon Releases (ECP078)	1 Nov 81	-
F18 IASC 002	-	Interim Airborne Software Change, Radar, Reloading of (ECP079)	15 Mar 82	-
F18 IASC 003	-	Interim Airborne Software Change, Mission Computer, Reloading of (ECP079)	15 Mar 82	-

Record of Applicable Technical Directives (Continued)

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F18-MD-0020	-	Right and Left ATS Failure Indications, Speed Sensing and Overspeed Protection, Addition of	1 Sep 82	-
F18-MD-0067	-	New Flight Control Software and Compatibility - Lot III	1 Sep 82	-
F18 IASC 005	-	Interim Airborne Software Change, Armament Computer, Reloading of (ECP044)	1 Sep 82	-
F18 IASC 006	-	Interim Airborne Software Change, Digital Data, Computer, Reloading of (ECP044)	1 Sep 82	-
F18 IASC 007	-	Interim Airborne Software Change, Computer Power Supply, Reloading of (ECP044)	1 Sep 82	-
F18 IASC 008	-	Interim Airborne Software Change, Integrated Flight Control, Roll-Pitch-Yaw Computer, Reloading of (ECP044)	1 Sep 82	-
F18 IASC 009	-	Interim Airborne Software Change, Inertial Navigation Group, Reloading of (ECP044)	1 Sep 82	-
F18 IASC 010	-	Interim Airborne Software Change, Maintenance Status Display and Recording System - Signal Data Recording Set BIT, Reprogramming of (ECP116)	1 Nov 82	-
F18 IASC 011	-	Interim Airborne Software Change, Computer Power Supply, Reloading of (ECP095)	1 Nov 82	-
F18 IASC 012	-	Interim Airborne Software Change, Inertial Navigation Group, Reloading of (ECP095)	1 Nov 82	-
F18 IASC 013	-	Interim Airborne Software Change, Armament Computer, Reloading of (ECP095)	1 Nov 82	-
F18 IASC 014	-	Interim Airborne Software Change, Digital Data Computer, Reloading of (ECP095)	1 Nov 82	-
F18 IASC 015	-	Interim Airborne Software Change, Digital Data Computer, Reloading of (ECP149)	1 May 83	-
F18 IASC 016	-	Interim Airborne Software Change, Integrated Flight Controls, Roll-Pitch-Yaw Computer CP-1330/ASW-44, Reprogramming of (ECP151)	15 Nov 83	-

Record of Applicable Technical Directives (Continued)

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F18 IASC 017	-	Interim Airborne Software Change, Computer Power Supply, CP-1325/APG-65; Reloading of (ECP155)	15 Nov 83	-
F18 IASC 018	-	Interim Airborne Software Change, Armament Computer CP-1342/AYQ-9(V); Reloading of (ECP155)	15 Nov 83	-
F18 IASC 019	-	Interim Airborne Software Change, Digital Data Computer, CP-1429 or 1539/AYK-14(V); Reloading of (ECP155)	15 Nov 83	-
F18 IASC 020	-	Interim Airborne Software Change, F/TF/A-18A Integrated Flight Controls, Roll-Pitch-Yaw Computer CP-1330/ASW-44, Reprogramming of (ECP142)	15 Sep 84	-
F18 IASC 021	-	Interim Airborne Software Change Inertial Navigation Unit CN-1561/ASN-130A, Reloading of (ECP169)	15 Sep 84	-
F18 IASC 022	-	Interim Airborne Software Change F/TF/A-18A Digital Data Computer, CP1539/ AYK-14(V): Reloading of (ECP169)	15 Sep 84	-
F18 IASC 023	-	Interim Airborne Software Change F/TF/A-18A Computer Power Supply, CP1325/APG-65: Reloading of (ECP169)	15 Sep 84	-
F18 IASC 024	-	Interim Airborne Software Change F/TF/A-18A Armament Computer, CP-1342/AYQ-9(V): Reloading of (ECP169)	15 Sep 84	-
F18 AFC 003	-	Air Cooling System-Restrictor Replacement for AN/AYK-14 Mission Computer (ECP 00029)	1 Jun 86	-
F18 IASC 025	-	Interim Airborne Software Change, F/A-18A/B Controller- Processor C-10661/AAS-38: Reloading of (ECP 00238)	1 Sep 86	-
F18 IASC 026	-	Interim Airborne Software Change, F/A-18A/B Digital Data Computer, CP-1539/ AYK-14(V): Reloading of (ECP 00243)	1 Sep 86	-
F18 IASC 027	-	Interim Airborne Software Change, F/A-18A/B Computer Power Supply, CP-1325/ APG-65: Reloading of (ECP 00243)	1 Sep 86	-

Record of Applicable Technical Directives (Continued)

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F18 IASC 028	-	Interim Airborne Software Change, F/A-18A/B Armament Computer, CP-1342/AYQ-9(V): Reloading of (ECP 00243)	1 Sep 86	-
F18 IASC 026 (REV A)	-	Interim Airborne Software Change, F/A-18A/B Digital Data Computer, CP-1539/ AYK-14(V): Reloading of (ECP 00243, Rev. A)	15 Apr 87	-
F18 IASC 027 (REV A)	-	Interim Airborne Software Change, F/A-18A/B Computer Power Supply, CP-1325/ APG-65: Reloading of (ECP 00243, Rev. A)	15 Apr 87	-
F18 IASC 028 (REV A)	-	Interim Airborne Software Change, F/A-18A/B Armament Computer, CP-1342/AYQ-9(V): Reloading of (ECP 00243, Rev. A)	15 Apr 87	-
F18 IASC 031	-	Interim Airborne Software Change, F/A-18 Digital Data Computer, CP-1539/AYK-14(V): Reloading of (ECP 00252)	15 Mar 88	-
F18 IASC 032	-	Interim Airborne Software Change, F/A-18 Computer Power Supply CP-1325/ APG-65: Reloading of (ECP 00252)	15 Mar 88	-
F18 IASC 034	-	Interim Airborne Software Change, F/A-18 Armament Computer CP-1342/AYQ-9(V): Reloading of (ECP 00252)	15 Mar 88	-
F/A-18 AFC 27	1 Aug 90	Improvement of Leading Edge Flap Design (ECP MDA-F/A-18-00044)	15 Mar 88	-
F18 IASC 033	-	Interim Airborne Software Change, F/A-18A/B Controller-Processor C-10661/AAS-38: Reloading of (ECP 00252)	1 Oct 88	-
F18 IASC 037 PT 2	-	Interim Airborne Software Change, F/A-18 Digital Data Computer CP-1539/AYK-14(V): Reloading of (ECP 00318)	1 Mar 90	-
F18 IASC 039 PT 2	-	Interim Airborne Software Change, F/A-18 Computer Power Supply CP-1325/APG-65: Reloading of (ECP 00318)	1 Mar 90	-
F18 IASC 038 PT 2	-	Interim Airborne Software Change, F/A-18 Armament Computer CP-1342/AVQ-9(V): Reloading of (ECP 00318)	1 Mar 90	-

Record of Applicable Technical Directives (Continued)

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F18 IASC 040	-	Interim Airborne Software Change, F/A-18 Control-Converter C-10382/A: Reloading of (ECP 00318)	1 Mar 90	-
F18 IASC 049 PT 2	-	Interim Airborne Software Change, F/A-18 13200988-02 Digital Data Computer CP-1539A/AYK-14(V): Reloading of (ECP 00383R1)	1 Sep 95	-
F18 IASC 050 PT 2	-	Interim Airborne Software Change, F/A-18 7959650-006. -007, -008 Armament Computer CP-1342/AYQ-9(V): Reloading of (ECP 00383R1)	1 Sep 95	-
F18 IASC 051 PT 2	-	Interim Airborne Software Change, F/A-18 Computer Power Supply: Reloading of (ECP 00383R1)	1 Sep 95	-
F18 IASC 060	-	Interim Airborne Software Change, AN/AAS-38A Detecting Set: Reloading of (ECP 00383R1)	1 Sep 95	-
F18 IASC 103	-	Interim Airborne Software Change, CP-1539A/AYK-14(V) Mission Computer, Reloading of (WUC 7415Y00)	15 Oct 00	-
F18 IASC 104	-	Interim Airborne Software Change, AM-6979/A Intercommunication Control Amplifier, Reloading of (WUC 64X1100)	15 Oct 00	-
F18 IASC 111	-	Interim Airborne Software Change, CP-2062 APG-73 Radar Processor, Reloading of	15 Oct 00	-
F18 IASC 116	-	Interim Airborne Software Change, C-10382/A Control-Converter, P/N 7959750-007 or 009, Reloading of	15 Oct 00	-
F/A-18 AVC JAX-AV-011	-	C-10661/AAS-38 Controller-Processor Assy, Modification of	1 Nov 93	-

1. INTRODUCTION.

2. This work package lists the program load CONFIG/IDENT numbers, program part numbers, unit part numbers, and aircraft effectivity related to a software configuration. The technical manuals affected by the software changes are listed with the date of publication which incorporates the change.

3. PROGRAM CHANGES.

4. There are two types of changes:

a. Program change only.

b. Program change with related component or aircraft wiring changes.

5. This manual includes before and after program load changes.

6. AIRCRAFT SOFTWARE CONFIGURATIONS.

7. Table of use and applicability identifies the software program loads (versions) that are required for a specific aircraft. The program loads identified by either a program load CONFIG/IDENT number, a program part number, or a unit part number. The table lists the items below:

a. programmable units

b. program load CONFIG/IDENT number (when applicable)

c. program part number (when applicable)

d. unit part number

e. aircraft effectivity

8. **PROGRAMMABLE UNITS.** Lists all programmable units. Programmable units include:

a. digital data computer no. 1

b. digital data computer no. 2

c. Armament Computer CP-1342/AYQ-9(V)

d. Computer-Power Supply CP-1325/APG-65

e. Inertial Navigation Unit CN-1561/ASN-130A or Inertial Navigation Group OA-8955/ASN-130

f. Control-Converter C-10382/A

g. Intercommunications Amplifier-Control AM-6979/A or Intercommunications Amplifier-Control AM-7360/A

h. Receiver-Transmitter RT-1250( )/ARC

i. Receiver-Transmitter Processor RT-1379/ASW

j. Air Data Computer CP-1334/A

k. Roll-Pitch-Yaw Computer CP-1330/ASW-44

l. Signal Data Recorder RO-508/ASM-612

m. Signal Data Converter CV-3493/ASM-612

n. Interconnecting Box J-3656/ASQ-173

o. Controller-Processor C-10661/AAS-38

9. **PROGRAM LOAD CONFIG/IDENT NUMBER.** Identifies the program that is loaded in applicable programmable units. The program load CONF/IDENT number is displayed on the configuration display using the CONFIG/IDENT verification procedure (WP004 00).

10. The program load CONFIG/IDENT number for digital data computers no. 1 and no. 2, Armament Computer CP-1342/AYQ-9(V), Computer-Power Supply CP-1325/APG-65, and Controller-Processor C-10661/AAS-38 consists of two groups of three characters. The first three characters are the program load identifier for the basic program load. The last three characters are the program identification (PID) number which identifies a specific version of the program load. The PID is also the auto load number required when using the Computer Memory Loader-Verifier Test Set AN/ASM-607(V)5.

11. **PROGRAM PART NUMBER.** The program part number is placarded on the program identification nameplate of programmable units that do not require a unit part number change when the program load is changed. The program part number is the same as the unit part number on programmable units that require a unit part number change when the program load is changed. It is used to identify the program that is loaded in the unit.

12. **UNIT PART NUMBER.** The unit part number is listed for all programmable units and identifies

the unit that is compatible with the corresponding program load. For replacement of programmable units, see applicable System Maintenance with IPB manual (A1-F18AC-( )-300).

13. Some units require a part number change when the program load is changed. Units requiring a part number change are:

a. Receiver-Transmitter RT-1250( )/ARC

b. Receiver-Transmitter Processor RT-1379/ASW

c. Air Data Computer CP-1334/A

d. Roll-Pitch-Yaw Computer CP-1330/ASW-44

e. Signal Data Recorder RO-508/ASM-612

f. Signal Data Converter CV-3493/ASM-612

14. The units that do not require a unit part number change when the program load is changed are:

a. Digital Data Computer No. 1

b. Digital Data Computer No. 2

c. Armament Computer CP-1342/AYQ-9(V)

d. Computer-Power Supply CP-1325/APG-65

e. Inertial Navigation Unit CN-1561/ASN-130A or Inertial Navigation Group OA-8955/ASN-130

f. Control-Converter C-10382/A

g. Intercommunication Amplifier-Control AM-6979/A or Intercommunication Amplifier-Control AM-7360/A

h. Interconnecting Box J-3656/ASQ-173

i. Controller-Processor C-10661/AAS-38

15. **AIRCRAFT EFFECTIVITY.** Lists aircraft bureau numbers, and when applicable, retrofit document number which the unit program load is usable on.

Table 1. Table of Use and Applicability

Programmable Units	Program Load CONFIG/IDENT Number	Program Part Number/Unit Part Number	Aircraft Effectivity
Air Data Computer CP-1334/A	—	4031000-906/ 4031000-906	161353 THRU 161519
	—	4031000-912/ 4031000-912	161520 THRU 162909 AFTER ECP 044
	—	4031000-913/ 4031000-913	163092 AND UP AFTER ECP 035
	—	4031000-914/ 4031000-914	163092 AND UP
	—	4031000-915/ 4031000-915	163092 AND UP AFTER ECP 178R1
Armament Computer CP-1342/AYQ-9(V)	3 120B	74A870620-1025/ 7959650-006	161353 THRU 163152 BEFORE F18 IASC 028 REV A
	3 85A+582	74A870620-1035/ 7959650-006, -007	161353 AND UP BEFORE F18 IASC 034
	3 87A-504	74A870620-1037/ 7959650-006, -007	161353 AND UP AFTER F18 IASC 034
	3 89A-522	74A870620-1037/ 7959650-007, -008	161353 AND UP AFTER F18 IASC 038 PT 2
	3 92A-527	74A870620-1037/ 7959650-006, -007, -008	161520 THRU 163175 AFTER F18 IASC 050 PT 2
	3 10A-532U or 10A-532V	74A870620-1037/ 7959650-006, -007, -008	161353 THRU 163175
Command Launch Computer CP-1001( )/AWG	5 ( )	5 ( )/ 704AS5961-2	161353 AND UP
Computer-Power Supply CP-1325/APG-65	3 101D	74A870619-1013/ 3525681-124, -135, -140	161353 THRU 161924 BEFORE F18 IASC 027 REV A
	3 102B	74A870619-1015/ 3525681-140	161925 THRU 161987; ALSO 161702 THRU 161924 AFTER F18 IASC 027 REV A

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/IDENT Number	Program Part Number/Unit Part Number	Aircraft Effectivity
	3 85A+063	74A870619-1019/ 3525681-140, -145	161702 AND UP BEFORE F18 IASC 032
	3 87X-073	74A870619-1020/ 3525681-145	161702 AND UP AFTER F18 IASC 032
	3 89X-873	74A870619-1023/ 3525681-140, -145	161702 AND UP AFTER F18 IASC 039 PT 2
	3 89X-875	74A870619-1025/ 3525681-140, -145	161702 AND UP AFTER F18 IASC 039 PT 2
	3 89X-877( )	74A870619-1025/ 3525681-140, -145	161702 THRU 163175 AFTER F18 IASC 051 PT 2
Control-Converter C-10382/A	3 1003	74A870624-1003/ 7959750-003	161353 THRU 161528
	3 1005	74A870624-1005/ 7959750-005	161702 AND UP
	3 89X-002	74A870624-1007/ 7959750-005	161353 THRU 163175 AFTER F18 IASC 40
	3 15C-002	74A870624-1025/ 7959750-007/009	162394 THRU 163175 AFTER F18 IASC 116
Controller-Processor C-10661/AAS-38 (FLIR)	3 84 030	3061270-1/ 3061270-1	1 161353 AND UP BEFORE F18 IASC 025
	3 85X-032	7 3061270-1/ 3061270-1 OR 7 3061270-2/ 6 3061270-2	1 161353 AND UP AFTER F18 IASC 025
	3 87X-033	7 3061270-1/ 3061270-1 OR 7 3061270-2/ 6 3061270-2	1 161353 AND UP AFTER F18 IASC 025

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/IDENT Number	Program Part Number/Unit Part Number	Aircraft Effectivity
	3 87X-034	7 3061270-1/ 3061270-1 OR 7 3061270-2/ 6 3061270-2	1 161353 AND UP AFTER F18 IASC 033
	3 87X-035	3061270-2/ 3061270-2	2 161353 AND UP
	3 88X-036	3061270-2/ 3061270-2	2 161353 AND UP
	3 89X-035	3061270-2/ 3061270-2	4 161353 AND UP
	3 89X-037	3061270-2/ 3061270-2	4 161353 AND UP
	3 89X+038	3061270-2/ 3061270-2	4 161353 AND UP AFTER F18 IASC 60
Digital Data Computer No. 1	3 85A+119	74A870618-1023/ 13200988-01, -02	161353 THRU 163175 BEFORE F18 IASC 031
	3 87X-145	74A870618-1025/ 13200988-01, -02	161353 THRU 163175 AFTER F18 IASC 031
	3 87X+147	74A870618-1026/ 13200988-02	161353 THRU 163175 AFTER F18 ASC 031
	3 87X%153	74A870618-1026/ 13200988-02	161353 THRU 163175
	3 89A-161	74A870618-1028/ 13200988-02	161353 THRU 163175 AFTER F18 IASC 037 PT 2
	3 92A-177( )	74A870618-1028/ 13200988-02	161520 THRU 163175 AFTER F18 IASC 049 PT 2
	3 10A-185U or 10A-185V	74A870618-1028/ 13200988-02	161353 THRU 163175
	3 10A+187U/V	74A870618-1028/ 13200988-02	161519 THRU 163123 AFTER F18 IASC 103

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/IDENT Number	Program Part Number/Unit Part Number	Aircraft Effectivity
	3 12A-195U	74A870618-1028/ 13200988-02	161519 THRU 163123
Digital Data Computer No. 2	3 85A+120	74A870618-1023/ 13200988-01, -02	161353 THRU 163175 BEFORE F18 IASC 031
	3 87X-146	74A870618-1025/ 13200988-01, -02	161353 THRU 163175 AFTER F18 IASC 031
	3 87X+148	74A870618-1026/ 13200988-02	161353 THRU 163175 AFTER F18 ASC 031
	3 87X % 154	74A870618-1026/ 13200988-02	161353 THRU 163175
	3 89A-162	74A870618-1028/ 13200988-02	161353 THRU 163175 AFTER F18 IASC 037 PT 2
	3 92A-178( )	74A870618-1028/ 13200988-02	161520 THRU 163175 AFTER F18 IASC 049 PT 2
	3 10A-186U or 10A-186V	74A870618-1028/ 13200988-02	161353 THRU 163175
	3 10A+188U/V	74A870618-1028/ 13200988-02	161519 THRU 163123 AFTER F18 IASC 103
	3 12A-196U	74A870618-1028/ 13200988-02	161519 THRU 163123
EGI Receiver CN-1694(V)4/ASN-172(V)	3 95H-LEMU	800990-2	
Inertial Navigation Unit CN-1561/ASN-130A	3 300	74A870617-1007/ 879010-1-00300	161702 THRU 162413 BEFORE F18 IASC 021
	3 84B02	74A870617-1010/ 879010-1-84B02 or 74A870617-1010/ 879010-2-84B02	161702 THRU 163175 AFTER F18 IASC 021
Inertial Navigation Group OA-8955/ ASN-130	3 042	774R870001-1003/ 874500-3, -4	161213 THRU 161251 AFTER F18 IASC 012

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/IDENT Number	Program Part Number/Unit Part Number	Aircraft Effectivity
	3 200	74A870617-1006/ 874910-1, -2	161520 THRU 161528; ALSO 161353 THRU 161519 AFTER F18 IASC 009
Intercommunication Amplifier-Control AM-6979/A	—	74A870633-1001/ 5016400-1	161353 THRU 161528
	—	74A870633-1001/ 5016400-2, -4, -6	161353 THRU 161987
	—	74A870633-1002/ 5016400-4, -6	161702 AND UP
	—	74A870633-1019/ 5016400-4, -6	161519 THRU 163123 AFTER F18 IASC 104
Interconnecting Box J-3656/ASQ-173 (LDT)	3 102 or 103	71320600-019/ 71320600-019	161353 AND UP
Radar Data Processor CP-2062/APG-73	3 15C-701V OR 15C+704U	74A870666-1037/ 3525046-110	162394 THRU 163175
Receiver-Transmitter RT-1250A/ARC	—	622-6321-001/ 622-6321-001	161353 AND UP
Receiver-Transmitter RT-1250/ARC	—	622-4016-001/ 622-4016-001	161353 AND UP
Receiver-Transmitter RT-1824(C)/ARC	3 15C-034 OR 15C-037	822-1133-001	162394 THRU 163175
Receiver-Transmitter-Processor RT-1379A/ ASW	—	622-5663-002/ 622-5663-002	161353 AND UP
Roll-Pitch-Yaw Computer CP-1330/ASW-44	—	897E518G101 OR 897E518G128/ 897E518G101, OR 897E518G128	161353 THRU 161519 BEFORE F/A-18 AFC 27
	—	897E518G110/ 897E518G110	161520 THRU 161528 BEFORE F18 IASC 008, F18 IASC 016, AND F/A-18 AFC 27

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/IDENT Number	Program Part Number/Unit Part Number	Aircraft Effectivity
	3 75	897E518G115 OR 897E518G120/ 897E518G115, OR 897E518G120	161520 THRU 161528 AFTER F18 IASC 008 BUT BEFORE F18 IASC 016 AND F/A-18 AFC 27
	3 75	897E518G123 OR 897E518G129/ 897E518G123 OR 897E518G129	161520 THRU 161528 AFTER F18 IASC 016 BUT BEFORE F/A-18 AFC 27
	3 75	897E518G303/ 897E518G303	161702 THRU 161924 BEFORE F18 IASC 016 AND F18 IASC 020
	3 86	897E518G304/ 897E518G304	161925 THRU 161987 BEFORE F18 IASC 020, ALSO 161702 THRU 161924 AFTER F18 IASC 016 BUT BEFORE F18 IASC 020
	3 86	897E518G307/ 897E518G307	161702 THRU 161924 AFTER F18 IASC 016 BUT BEFORE F18 IASC 020
	3 99	897E518G309 Version 8.3.3/ 897E518G309	8 162394 AND UP; ALSO 161702 THRU 161987 AFTER F18 IASC 020 AND 161353 THRU 161528 AFTER F/A-18 AFC 27
	3 113	897E518G310 Version 8.5/ 897E518G310	8 161353 AND UP AFTER ECP 324
Roll-Pitch-Yaw Computer CP-1330A/ASW-44	3 91C*004 or 117	936E918G5 Version 10.7	
Signal Data Recorder RO-508/ASM-612	—	3839010-5/ 3839010-5	161353 THRU 161528 BEFORE F18 IASC 010
	—	3839010-6/ 3839010-6	161702 AND UP; ALSO 161353 THRU 161528 AFTER F18 IASC 010

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/IDENT Number	Program Part Number/Unit Part Number	Aircraft Effectivity
Signal Data Converter CV-3493/ASM-612	—	3761022-1/ 3761022-1	161353 AND UP

LEGEND

1 FLIR POD CUM 8 THROUGH 123.
2 FLIR POD CUM 124 AND UP.
3 REFER TO WP004 00 TO DETERMINE PROGRAM LOAD CONFIG/IDENT NUMBER FOR UNITS INSTALLED ON AIRCRAFT.
4 FLIR POD CUM 124 AND UP; ALSO CUM 8 THROUGH 123 AFTER F/A-18 AVC JAX-AV-011.
5 CLASSIFIED PROGRAM CONTACT TYCOM FOR PROGRAM IDENTIFICATION.
6 3061270-2 MAY BE USED ON POD 8 THROUGH 123 ONLY IF 3061270-1 IS NOT AVAILABLE OR AFTER F/A-18 AVC JAX-AV-011.
7 PROGRAM CONFIG/IDENT 87X-034 IS PREFERRED SOFTWARE LOAD.
8 MUST BE USED AS A SET, DO NOT MIX PER AIRCRAFT.

Table 2. Current Program Load Configurations

Programmable Units	Program Load CONFIG/IDENT Number	Program Part Number/Unit Part Number	Aircraft Effectivity
Air Data Computer CP-1334/A	—	4031000-912/ 4031000-912	161353 THRU 162909 AFTER ECP 044
	—	4031000-913/ 4031000-913	163092 AND UP AFTER ECP 035
	—	4031000-915/ 4031000-915	163092 AND UP AFTER ECP 178R1
Armament Computer CP-1342/AYQ-9(V)	[2] 92A-527U	74A870620-1037/ 7959650-006, -007, -008	161520 THRU 163175 AFTER F18 IASC 050 PT 2
	[3] 10A-532U or 10A-532V	74A870620-1037/ 7959650-006, -007, -008	161353 THRU 163175
Command Launch Computer CP-1001( )/ AWG	[2] ( )	[4] ( )/ 704AS5961-2	161353 AND UP
Computer-Power Supply CP-1325/APG-65	[2] 89X-875	74A870619-1025/ 3525681-140 / -145	161353 AND UP AFTER F18 IASC 039 PT 2
	[2] 89X-877U	74A870619-1025/ 3525681-140 / -145	161353 THRU 163175 AFTER F18 IASC 051 PT 2
Control-Converter C-10382/A	[2] 89X-002	74A870624-1007/ 7959750-005	161353 AND UP AFTER F18 IASC 40
	[2] 15C-002	74A870624-1025/ 7959750-007/009	162394 THRU 163175 AFTER F18 IASC 116
Controller-Processor C-10661/AAS-38 (FLIR)	[2] 85X-032	[5] 3061270-1/ 3061270-1	[1] 161353 AND UP
	[2] 87X-033	[5] 3061270-1/ 3061270-1	[1] 161353 AND UP
	[2] 87X-034	[5] 3061270-1/ 3061270-1	[1] 161353 AND UP
	[2] 89X-035	3061270-2/ 3061270-2	[3] 161353 AND UP
	[2] 89X+038	3061270-2/ 3061270-2	[3] 161353 AND UP AFTER F18 IASC 060

Table 2. Current Program Load Configurations (Continued)

Programmable Units	Program Load CONFIG/IDENT Number	Program Part Number/Unit Part Number	Aircraft Effectivity
Digital Data Computer No. 1	2 10A+187U/V	74A870618-1028/ 13200988-02	161519 THRU 163123 AFTER IASC 103
	2 12A-195U	74A870618-1028/ 13200988-02	161519 THRU 163123
Digital Data Computer No. 2	2 10A+188U/V	74A870618-1028/ 13200988-02	161519 THRU 163123 AFTER IASC 103
	2 12A-196U	74A870618-1028/ 13200988-02	161519 THRU 163123
EGI Receiver CN-1694(V)4/ASN-172(V)	2 95H-LEMU	800990-2	
Inertial Navigation Unit CN-1561/ASN-130A	2 84B02	74A870617-1010/ or 74A870617-1010/ 879010-2-84B02	161353 THRU 163175 AFTER F18 IASC 021 REV A
Intercommunication Amplifier-Control AM-6979/A	—	74A870633-1002/ 5016400-4,-6	161353 AND UP
	—	74A870633-1019/ 5016400-4,-6	161519 THRU 163123 AFTER IASC 104
Interconnecting Box J-3656/ASQ-173 (LDT)	2 102 or 103	71320600-019/ 71320600-019	161353 AND UP
Radar Data Processor CP-2062/APG-73	2 15C-701U/V	74A870666-1037/ 3525046-110	162394 THRU 163175 AFTER F18 IASC 111
Receiver-Transmitter RT-1250A/ARC	—	622-6321-001/ 622-6321-001	161353 AND UP
Receiver-Transmitter RT-1250/ARC	—	622-4016-001/ 622-4016-001	161353 AND UP

Table 2. Current Program Load Configurations (Continued)

Programmable Units	Program Load CONFIG/IDENT Number	Program Part Number/Unit Part Number	Aircraft Effectivity
Receiver-Transmitter RT-1824(C)/ARC	2 15C-034 OR 15C-037	822-1133-001	162394 THRU 163175
Receiver-Transmitter-Processor RT-1379A/ ASW	—	622-5663-002/ 622-5663-002	161353 AND UP
Roll-Pitch-Yaw Computer CP-1330/ASW-44	2 113	897E518G310 Version 8.5/ 897E518G310	6 161353 AND UP AFTER ECP 324
Roll-Pitch-Yaw Computer CP-1330A/ASW-44	2 91C*004 or 117	936E918G5 Version 10.7	
Signal Data Recorder RO-508/ASM-612	—	3839010-5/ 3839010-5	161353 THRU 161528 BEFORE F18 IASC 010
	—	3839010-6/ 3839010-6	161702 AND UP; ALSO 161353 THRU 161528 AFTER F18 IASC 010
Signal Data Converter CV-3493/ASM-612	—	3761022-1/ 3761022-1	161353 AND UP

LEGEND

1 FLIR POD CUM 8 THRU 123.
2 REFER TO WP004 00 TO DETERMINE PROGRAM LOAD CONFIG/IDENT NUMBER FOR UNITS INSTALLED ON AIRCRAFT.
3 FLIR POD CUM 124 AND UP; ALSO CUM 8 THROUGH 123 AFTER F/A-18 AVC JAX-AV-011.
4 CLASSIFIED PROGRAM CONTACT TYCOM FOR PROGRAM IDENTIFICATION.
5 3061270-2 MAY BE USED ON POD 8 THROUGH 123 ONLY IF 3061270-1 IS NOT AVAILABLE OR AFTER F/A-18 AVC JAX-AV-011.
6 MUST BE USED AS A SET, DO NOT MIX PER AIRCRAFT.

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

PROGRAM LOAD VERSIONS

EFFECTIVITY: F/A-18C AND F/A-18D

Reference Material

Software Configuration Manual A1-F18AC-SCM-000
Program Load CONFIG/IDENT Verification WP004 00

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Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F18 IASC 032	-	Interim Airborne Software Change, F/A-18 Computer Power Supply CP-1325/APG-65: Reloading of (ECP 00252)	1 Jun 88	-
F18 IASC 034	-	Interim Airborne Software Change, F/A-18 Armament Computer CP-1342/AYQ-9(V): Reloading of (ECP 00252)	1 Jun 88	-
F18 IASC 035	-	Interim Airborne Software Change, F/A-18 Flight Incident Recording and Aircraft Monitoring Set (FIRAMS) - Signal Data Computer CP-1726/ASQ-194 (SDC): Reloading of (ECP 00252)	1 Jun 88	-

Record of Applicable Technical Directives (Continued)

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F18 IASC 033	-	Interim Airborne Software Change, F/A-18A/B Controller - Processor C-10661/ AAS-38: Reloading of (ECP 00252)	1 Oct 88	-
F18 IAVC 4038	12 Mar 88	Signal Data Computer, CP-1726/ASQ-194, Modification of (ECP MDA F/A-18-00178R1C1)	15 Feb 89	-
F18 IASC 037 PT 1	-	Interim Airborne Software Change, F/A-18 Digital Data Computer CP-1699/AYK-14(V): Reloading of (ECP-00318)	1 Jan 90	-
F18 IASC 039 PT 1	-	Interim Airborne Software Change, F/A-18 Computer Power Supply CP-1325/APG-65: Reloading of (ECP 00318)	1 Jan 90	-
F18 IASC 038 PT 1	-	Interim Airborne Software Change, F/A-18 Armament Computer CP-1342/AVQ-9(V): Reloading of (ECP 00318)	1 Jan 90	-
F18 IASC 040	-	Interim Airborne Software Change, F/A-18 Control- Converter C-10382/A : Reloading of (ECP 00318)	1 Jan 90	-
F18 IASC 041	-	Interim Airborne Software Change, F/A-18 Memory Unit MU-860B/ASQ-194 : Reloading of (ECP 00383)	1 Sep 95	-
F18 IASC 045	-	Interim Airborne Software Change, F/A-18 Digital Map Computer CP-1802/ASQ-196 : Reloading of (ECP 00318)	1 Jan 90	-
F18 AVC 4324	-	CP-1342/AYQ-9(V) Armament Computer, Modification of (ECP 407)	15 Feb 93	-
F/A-18 AFC 136	1 Jul 92	CP-1330/ASW-44 Integrated Flight Controls, Roll-Pitch-Yaw Computer, Removal and Installation of (ECP 00324)	15 Feb 93	-
F/A-18 AFC 126	-	Deployable Flight Incident Recorder Set (ECP MDA-F/A-18-00321R1C1)	15 Feb 93	-
F18 IASC 047	-	Interim Airborne Software Change, Digital Map Set AN/ASQ-196, Reloading of, (ECP 00383)	15 Jul 93	-

Record of Applicable Technical Directives (Continued)

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F18 IASC 049	-	Interim Airborne Software Change, 13203471-20 Digital Data Computer CP-1699A/AYK-14(V), 13217903-01 or -06 Digital Data Computer CP-2060/AYK-14(V) or CP-2060A/AYK-14(V), Reloading of, (ECP 00383)	15 Jul 93	-
F18 IASC 050	-	Interim Airborne Software Change, 7959650 Armament Computer CP-1342/AYQ-9(V), Reloading of, (ECP 00383)	15 Jul 93	-
F18 IASC 051 PT 1, Rev A	-	Interim Airborne Software Change, Radar CP-1325/APG65, Reloading of, (ECP 00383)	15 Jul 93	-
F18 IASC 051 PT 1, Rev B	-	Interim Airborne Software Change, Radar CP-1325/APG65, Reloading of, (ECP 00383R1)	1 Sep 95	-
F18 IASC 052	-	Interim Airborne Software Change, Signal Data Computer CP-1726/ASQ-194, Reloading of, (ECP 00383)	15 Jul 93	-
F18 IASC 053	-	Interim Airborne Software Change, Roll-Pitch-Yaw Computer CP-1330A/ASW-44, Reloading of, (ECP 00383)	15 Jul 93	-
F18 AVC JAX-AV-011	-	C-10661/AAS-38 Controller-Processor Assy, Modification of	1 Nov 93	-
F/A-18 IASC 060	-	Interim Airborne Software Change, AN/AAS-38A Detecting Set, Reloading of, (ECP 00383R1)	15 Feb 94	-
F18 IASC 061	-	Interim Airborne Software Change, CP-2215/AYK-14V and CP-2216/AYK-14V Digital Data Computer, Reloading of, (ECP 00466)	1 Sep 95	-
F18 IASC 062	-	Interim Airborne Software Change, CP-1342/AYQ-9V Armament Computer, Reloading of, (ECP 00466)	1 Sep 95	-
F18 IASC 063	-	Interim Airborne Software Change, CP-2062/APG-73 Radar Data Processor, Reloading of, (ECP 00466)	1 Sep 95	-

Record of Applicable Technical Directives (Continued)

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F18 IASC 064	-	Interim Airborne Software Change, CP-1726/ASQ-194 Signal Data Computer, Reloading of, (ECP 00466)	1 Sep 95	-
F18 IASC 065	-	Interim Airborne Software Change, CP-1334A/A Air Data Computer, Reloading of, (ECP 00466)	1 Sep 95	-
F18 IASC 066	-	Interim Airborne Software Change, C-10382/A Control-Converter, Reloading of, (ECP 00466)	1 Sep 95	-
F18 IASC 067	-	Interim Airborne Software Change, IP-1556/A Digital Display Indicator (DDI), Reloading of, (ECP 00466)	1 Sep 95	-
F18 IASC 073	-	Interim Airborne Software Change, CP-1325/Apg-65 Radar Computer Power Supply, Reloading of, (ECP 00466)	1 Sep 95	-
F18 IASC 078	-	Interim Airborne Software Change, CP-2215/AYK-14, CP-2216/AYK-14, CP-2359/AYK-14 and CP-2360/AYK-14 Digital Data Computer , Reloading of	15 Mar 98	-
F18 IASC 079	-	Interim Airborne Software Change, F/A-18 CP-1342/AYQ-9(V), Armament Computer, Reloading of	15 Mar 98	-
F18 IASC 080	-	Interim Airborne Software Change, CP-1325/APG-65 Radar Computer Power Supply, Reloading of	15 Mar 98	-
F18 IASC 081	-	Interim Airborne Software Change, AM-7360/A Amplifier Control-Intercommunication, Modification of	15 Mar 98	-
F18 IASC 082	-	Interim Airborne Software Change, AN/AAS-38/-38A Detecting Set, Reloading of	15 Mar 98	-
F18 IASC 083	-	Interim Airborne Software Change, AN/AAS-38B Detecting Set, Reloading of	15 Mar 98	-
F18 IASC 084	-	Interim Airborne Software Change, CP-2062/APG-73 Radar Data Processor, Reloading of	15 Mar 98	-

Record of Applicable Technical Directives (Continued)

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F18 IASC 085	-	Interim Airborne Software Change, CP-2218/AYK-22(V) Armament Computer, Reloading of	15 Mar 98	-
F18 IASC 091	-	Interim Airborne Software Change, CP-2218/AYK-22(V) Armament Computer, Reloading of	15 Sep 98	-
F18 IASC 092	-	Interim Airborne Software Change, CP-1325/APG-65 Radar Computer Power Supply, Reloading of	15 Sep 98	-
F18 IASC 093	-	Interim Airborne Software Change, CP-2062/APG-73 Radar Data Processor, Reloading of	15 Sep 98	-
F18 IASC 094	-	Interim Airborne Software Change, R-2484/APG-73 Radar Receiver (BIST), Reloading of	15 Sep 98	-
F18 IASC 095	-	Interim Airborne Software Change, CP-1726/ASQ-194 Signal Data Computer, Reloading of	15 Sep 98	-
F18 IASC 096	-	Interim Airborne Software Change, RT-1763/APX-111(V) Combined Interrogator Transponder Set (CIT), Reloading of	1 May 99	-
F18 IASC 100	-	Interim Airborne Software Change, C-10382/A Control-Converter, P/N 7959750-007 or -009, Reloading of	15 Sep 98	-
F18 IASC 101	-	Interim Airborne Software Change, AN/AAS-38/-38A Detecting Set, Reloading of	15 Sep 98	-
F18 IASC 102	-	Interim Airborne Software Change, AN/AAS-38B Detecting Set, Reloading of	15 Sep 98	-
F18 IASC 111	-	Interim Airborne Software Change, CP-2062 APG-73 Radar Processor, Reloading of	15 Oct 00	-

Record of Applicable Technical Directives (Continued)

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F18 IASC 113	-	Interim Airborne Software Change, CP-1802 ASQ-196 Digital Map Computer, Installation of	15 Oct 00	-
F18 IASC 114	-	Interim Airborne Software Change, AM-7360/A Amplifier Control Intercommunication (ACI), Reloading of	15 Oct 00	-
F18 IASC 115	-	Interim Airborne Software Change, IP-1556 Digital Display Indicator (DDI), P/N 129000-59, Reloading of	15 Oct 00	-
F18 IASC 116	-	Interim Airborne Software Change, C-10382/A Control-Converter, P/N 7959750-007 or 009, Reloading of	15 Oct 00	-
F/A-18 AFC 175 PT 2	-	Miniaturized Airborne Global Positioning System (GPS) Receiver (MAGR), Incorporation of (ECP MDA-F/A-18-0405)	1 Jan 00	-
F18 AFC 185	-	Have Quick/Sincgars, Incorp of (ECP MDA-F/A-18-00292R1A3 /R2)	15 Feb 94	-
F/A-18 AFC 236	-	AN/APX-111(V) Combined Interrogator/ Transponder (CIT) Identification Friend or Foe (IFF) System, Retrofit of (ECP MDA-F/A-18-0520R1)	1 Jan 00	-
F/A-18 AFC 258	11 Jan 00	Crash Survivable Flight Incident Recorder (CSFIRS), Installation of (ECP MDA-F/A-18-0573)	15 Apr 00	-

1. INTRODUCTION.

2. This work package lists the CONFIG/IDENT numbers, program part numbers, unit part numbers, and aircraft effectivity related to a software configuration. The technical manuals affected by the software changes are listed with the date of publication which incorporates the change.

3. PROGRAM CHANGES.

4. There are two types of changes:

a. Program change only.

b. Program change with related component or aircraft wiring changes.

5. This manual includes before and after program load changes.

6. AIRCRAFT SOFTWARE CONFIGURATIONS.

7. Table of use and applicability identifies the software program loads (versions) that are required for a specific aircraft. The program loads are identified by either a program load CONFIG/IDENT number, a program part number, or a unit part number. The table lists the items below:

a. programmable units.

b. program load CONFIG/IDENT number (when applicable).

c. program part number (when applicable).

d. unit part number.

e. aircraft effectivity.

8. **PROGRAM LOAD CONFIG/IDENT NUMBER.** Identifies the program that is loaded in applicable programmable units. The number is displayed on the configuration display using the CONFIG/IDENT verification procedure (WP004 00).

9. The program load CONFIG/IDENT number for all programmable units except the Interconnecting Box J-3656/ASQ-173, Roll-Pitch-Yaw Computer CP-1330/ASW-44, some Control Converter C-10382/A, Inertial Navigation Unit CN-1521/ASN-130A or CN-1649/ASN-139, and Command Launch Computer CP-1001( )/AWG consists of two groups of three characters. The first three characters are the program load identifier for the basic program load. The last three characters are the program identification (PID) number which identifies a specific version of the program load. The PID is also the auto load number required when using the Computer Memory Loader-Verifier Test Set AN/ASM-607(V)5 or Advanced Memory Loader-Verifier Test Set AN/ASM-687.

10. **PROGRAM PART NUMBER.** The program part number is placarded on the program identification nameplate of programmable units that do not require a unit part number change when the program load is changed. The program part number is the same as the unit part number on programmable units that require a unit part number change when the program load is changed. It is used to identify the program that is loaded in the unit.

11. **UNIT PART NUMBER.** The unit part number is listed for all programmable units and identifies the unit that is compatible with the corresponding program load. For replacement of programmable units, see applicable System Maintenance with IPB manual (A1-F18AC-( )-300 , A1-F18AE-( )-300, A1-F18AG-( )-300, or A1-F18AH-( )-300.

12. Some units require a part number change when the program load is changed. Units requiring a part number change are:

a. Receiver-Transmitter RT-1250( )/ARC.

b. Receiver-Transmitter RT-1556/ARC.

c. Receiver-Transmitter Processor RT-1379A/ASW.

d. Air Data Computer CP-1334/A.

e. Roll-Pitch-Yaw Computer CP-1330/ASW-44.

13. **AIRCRAFT EFFECTIVITY.** Lists aircraft bureau numbers, and when applicable, retrofit document number which the unit program load is usable on.

Table 1. Table of Use and Applicability

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
Air Data Computer CP-1334/A	—	4031000-914/ 4031000-914	163427 THRU 163483
	—	4031000-915/ 4031000-915	163427 THRU 164279 AFTER ECP 178R1C1
Air Data Computer CP-1334A/A	5 91X-010	74A870656-1001/ 4031000-920	164627 AND UP AFTER ECP 206R2
	5 93X-022	74A870656-1004/ 4031000-920	164627 AND UP AFTER F18 IASC 065
Armament Computer CP-1342/AYQ-9(V)	5 91C-653( )	74A870620-1053/ 7959650-108	3 163429 AND UP AFTER F18 IASC 050 PT 1
	5 09C-663( )	74A870620-1069/ 7959650-108/-109	3 163429 AND UP AFTER F/A-18 AVC 4324 AND F18 IASC 062
	5 11C-676U	74A870620-1093/ 7959650-108, -109	163429 THRU 165206 AFTER F18 IASC 079
	5 15C-621U or 15C-696V	74A870620-1093/ 7959650-108, -109	163429 THRU 165206
	5 17C-608U Boot Ver 12	74A870620-1093/ 7959650-108, -109	163429 THRU 165206
Armament Computer CP- 2218/AYK-22(V)	5 11C-507U/V	74A870686 -1009/ 82370-01	165207 THRU 165416 AFTER F18 IASC 085
	5 13C-512U/V	74A870686 -1011/ 82370-01	165207 THRU 165532 AFTER F18 IASC 091
	5 15C+530U or 15C-532V	74A870686 -1009/ 82370-01	165207 AND UP
	5 17C-551U	74A870686 -1011/ 82370-01	165207 AND UP
Command Launch Computer CP-1001( )/AWG	5 005	9 ( )/ 704AS5961-2, -3	3 163427 AND UP
	5 007	9 ( )/ 704AS5961-2, -3	3 163427 AND UP

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
Computer-Power Supply CP-1325/APG-65	5 87X-073	74A870619-1020/ 3525681-145, -150, -155	163499 THRU 163782; ALSO 163427 THRU 163498 AFTER F18 IASC 032
	5 89X-873( )	74A870619-1024/ 3525681-145, -150, -155	163985 THRU 164897; ALSO 163427 THRU 163782 AFTER F18 IASC 039 PT 1
	5 91C-813( )	74A870619-1028/ 3525681-155	163427 THRU 164897 AFTER F18 IASC 051 PT 1, REV A
	5 91C-814( )	74A870619-1034/ 3525681-155	161353 THRU 164279; ALSO 164627 THRU 164897 BEFORE AFC 211 AND F18 IASC 051 PT 1, REV B
	5 09C-802( )	( )/ 3525681-155	161353 THRU 164279; ALSO 164627 THRU 164897 BEFORE AFC 211 AND AFTER F18 IASC 073
	5 11C-800U	74A870619-1037/ 3525681-155	163429 AND UP AFTER F18 IASC 080
	5 13C-800U	74A870619-1041/ 3525681-155	163429 THRU 164888 AFTER F18 IASC 092
Control-Converter C-10382/A	5 1005	74A870624-1005/ 7959750-005	163427 THRU 164279
	5 89X-002	74A870624-1007/ 7959750-005	163427 THRU 164279 AFTER F18 IASC 040 PT 1
	5 91C-005	74A870624-1009/ 7959750-007	164627 AND UP
	5 93C-001	74A870624-1015/ 7959750-007	164627 AND UP AFTER F18 IASC 066
	5 93C-007	74A870624-1025/ 7959750-007	163985 THRU 165532 AFTER F18 IASC 100

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
	5 15C-002	74A870624-1025/ 7959750-007/009	163427 AND UP AFTER F18 IASC 116
Controller-Processor C-10661( )/AAS-38 (FLIR)	5 85X-032	10 3061270-1/ 3061270-1	1 163427 AND UP
	5 87X-033	10 3061270-1/ 3061270-1	1 163427 AND UP
	5 87X-034	10 3061270-1/ 3061270-1	1 163427 AND UP AFTER F18 IASC 033
	5 89X-035	3061270-2/ 3061270-2	7 163427 AND UP
	5 91X+038	3061270-2/ 3061270-2	7 163427 AND UP AFTER F18 IASC 060 REV A
	5 11X-001	3061270-2	163427 AND UP AFTER F18 IASC 082
	5 13X-001	3061270-2	163427 THRU 165532 AFTER F18 IASC 101
Controller-Processor C-10661( )/AAS-38B (FLIR)	5 11X-103	260582	163427 AND UP AFTER F18 IASC 083
	5 13X-101	260582	163427 THRU 165532 AFTER F18 IASC 102
Data Transfer Interface Unit J-6008/A (DFIRS)	5 91C-011	74A870654-1007/ 136787-1	164725 AND UP; ALSO 164627 THRU 164724 AFTER F/A-18 AFC 126
Digital Computer Converter CP-1805/AAR-50 (NFLR)	5 89C-004	74A870649-1000/ 6096500-110	2 163985 AND UP
	5 89C-005	74A870649-102/6096500-110	163985 AND UP

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
	5 91C-002	74A870649-103/6096500-110	163985 AND UP AFTER F18 IASC 059
Digital Data Computer No. 1	5 91C+291( )	74A870618-1037/ 13203741-19, -20 or 13217903-06, ,-07, or 13221275-01	163427 AND UP AFTER F18 IASC 049 PT 1, REV B
	5 09C-215( )	74A870618-1054/ 13221275-01, or 13221952-01	8 163429 AND UP AFTER F18 IASC 061
	5 11C-235U	74A870618-1061/ 13221275-01	163429 AND UP AFTER F18 IASC 078
	5 11C-249U	74A870618-1073/ 13225797-01	163429 AND UP AFTER F18 IASC 078
	5 13C+271U	74A870618-( )/ 13225797-01	163429 AND UP
	5 15C-261U	74A870618-( )/ 13225797-01	163429 AND UP
	5 17C-203U	74A870618-( )/ 13225797-01	163429 AND UP
Digital Data Computer No. 2	5 91C+292( )	74A870618-1037/ 13203471-19, -20, 13217903-01, -06, -07, or 13221275-01	163427 AND UP AFTER F18 IASC 049 PT 1 REV B
	5 09C-216( )	74A870618-1054/ 13221275-01, or 13221952-01	8 163429 AND UP AFTER F18 IASC 061 PT 1
	5 11C-236U	74A870618-1061/ 13221952-01	163429 AND UP AFTER F18 IASC 078
	5 11C-250U	74A870618-1073/ 13225798-01	163429 AND UP AFTER F18 IASC 078
	5 13C+272U	74A870618-( )/ 13225797-01	163429 AND UP
	5 15C+262U	74A870618-( )/ 13225797-01	163429 AND UP

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
	5 17C-204U	74A870618-( )/ 13225797-01	163429 AND UP
Digital Display Indicator	5 89C-011	74A870654-1005/ 129000-29, -29A, -39, -49	163427 AND UP
	5 91C-015	74A870654-1007/ 129000-59	163985 AND UP
	5 09C-017	74A870654-1009/ 129000-59	163985 AND UP AFTER F18 IASC 067
	5 15C-022	74A870654-1009/ 129000-59	163985 AND UP AFTER F18 IASC 115
Digital Map Computer CP-1802/ASQ-196	5 89C-011	74A870650-1005/ 8506200-911, -912, -913, -914	164220 AND UP; ALSO 163985 THRU 164219 AFTER F18 IASC 047
	5 13C-006	74A870650-1005/ 8506200-911, -912, -913, -914	163985 AND UP AFTER F18 IASC 113
Digital Memory Unit MU-928/ASQ-196	DM0305	74A870650-1003/ 8505300-912	163427 AND UP
	DM0607	74A870650-1005/ 8505300-912, -913	163427 AND UP
Inertial Navigation Unit CN-1561/ASN-130A	5 84B02	74A870617-1010/ 879010-01-84B02 or 879010-02-84B02	163427 THRU 164068
Inertial Navigation Unit CN-1649/ASN-139	5 90X-L92U	886401/ 886401-1	163427 AND UP
	5 90X-L94U	886401/ 886401-2	163427 AND UP
	5 90X-L99U	886401/ 886401-3	163427 AND UP
Intercommunication Amplifier-Control AM-7360/A	—	74A870633-1006/ 5150100-1	F/A-18C 163427 THRU 164013, F/A-18D 163434 THRU 164009

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
	—	74A870633-1007/ 5150100-1	F/A-18C 163427 THRU 164279
	—	74A870633-1008/ 5150100-2	163427 THRU 164279 AFTER ECP 178R1C2
	—	74A870633-1009/ 5150100-2	164627 AND UP AFTER ECP 292
	—	74A870633-1011/ 5150100-2	164627 AND UP AFTER F18 IASC 058
	—	74A870633-1016/ 5150100-2	163429 AND UP AFTER F18 IASC 081
	—	74A870633-1018/ 5150100-2	163429 AND UP AFTER F18 IASC 114
Interconnecting Box J-3656/ASQ-173 (LDT)	5 102 or 103	71320600-019/ 71320600-019	163427 AND UP
Memory Unit MU-860B/ASQ-194	5 87D-002	Y799998F-002/ 791700-4	4 163427 THRU 163726
	5 87D-003	74A870646-1002/ 791700-4	4 163427 AND UP
	5 87D-004	74A870646-1002/ 791700-7	163427 AND UP AFTER F18 IASC 041
MIDS Radio Terminal RT-1765(C)/USQ-140(V)(C)	5 E6D or 45D or E76	P600A58-1	163429 THRU 165687
Radar Receiver R-2484/APG-73	5 48Z	74A870683-1005/ 3525026-110	164627 AND UP
	5 68	74A870683-1007/ 3525026-110	164627 THRU 165532 AFTER F18 IASC 094
Radar Data Processor CP-2062/APG-73	5 09C-704( )	74A870666-1011/ 3525046-110	164627 AND UP
	5 11C-700U	74A870666-1027/ 3525046-110	164627 THRU 165416 AFTER F18 IASC 084
	5 13C-702U/V	74A870666-1037/ 3525046-110	164627 THRU 165532 AFTER F18 IASC 093

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
	5 15C+704U	74A870666-1037/ 3525046-110	164627 THRU 165687 AFTER F18 IASC 111
Radio Receiver (GPS) R-2512A/U	—	613-8379-007	164945 AND UP; ALSO 163429 THRU 164912 AFTER F/A-18 AFC 175 PT 2
Radio Receiver-Transmitter (CIT) RT-1763/APX-111(V)	5 11C-007	74A870657-TBD/ 1007101G-10	165222 AND UP
	5 13C-004U/V	74A870685-1003	165222 AND UP; ALSO 163985 THRU 165221 AFTER F/A-18 AFC 236 AND FA18 IASC 096
	5 15C-003U or 15C-002	74A870657-TBD/ 1007101G-10	165222 AND UP
	5 17C-003U	74A870685-1003	165222 AND UP
Receiver-Transmitter RT-1250/ARC	—	622-4016-001/ 622-4016-001	163427 AND UP
Receiver-Transmitter RT-1250A/ARC	—	622-6321-001/ 622-6321-001	163427 AND UP
Receiver-Transmitter RT-1556( )/ARC	—	622-9878-001/ 622-9878-001	164898 AND UP; ALSO 163427 THRU 164897 AFTER F/A-18 AFC 185
Receiver-Transmitter RT-1824(C)/ARC	5 15C-038	822-1133-001	163429 THRU 165687
Receiver-Transmitter-Processor RT-1379A/ASW	—	622-5663-002/ 622-5663-002	163427 AND UP
Roll-Pitch-Yaw Computer CP-1330/ASW-44	5 99	897E518G309 Version 8.3.3/ 897E518G309	6 163427 THRU 163510
	5 113	897E518G310 Version 8.5/ 897E518G310	6 163427 THRU 163510 AFTER F/A-18 AFC 136

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
Roll-Pitch-Yaw Computer CP-1330A/ASW-44	5 ▶ 107	936E918G1 Version 10.1/ 936E918G1	11 ▶ 163699 THRU 163773
		936E918G2 Version 10.1/ 936E918G2	11 ▶ 163699 AND UP

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
	5 91C*002 or 112	936E918G4 Version 10.3	6 163699 AND UP AFTER F18 IASC 053
	5 91C*004 or 117	936E918G6 Version 10.5.1	6 163699 AND UP
	5 91C*004 or 117	936E918G6 Version 10.7	
Signal Data Computer CP-1726/ASQ-194	5 87D-307	74A870645-1007/ 791660-5	163427 THRU 163431
		74A870645-1007/ 791660-6	163433 THRU 163498; ALSO 163427 THRU 163431 AFTER F/A-18 AVC 4038
		74A870645-1007/ 791660-8	163499 THRU 163510; ALSO 163427 THRU 163498 AFTER F18 IASC 035
		74A870645-1007/ 791660-10	163427 THRU 163782 AFTER F18 IASC 035
		74A870645-1007/ 791660-11	163427 AND UP AFTER F18 IASC 035
	5 91C-327	74A870645-1009/ 791660-8	163499 THRU 163510; ALSO 163427 THRU 163498 AFTER F18 IASC 035 AND IASC 052 REV A
		74A870645-1009/ 791660-10	163699 THRU 163782; ALSO 163427 THRU 163510 AFTER F18 IASC 052 REV A
		74A870645-1009/ 791660-11	163985 AND UP; ALSO 163427 THRU 163782 AFTER F18 IASC 052 REV A
		74A870645-1009/ 791660-12	163429 THRU 164724 AFTER F18 IASC 052 REV B

Table 1. Table of Use and Applicability (Continued)

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
	5 09C-326	74A870645-1011/ 791660-8	163499 THRU 163510; ALSO 163427 THRU 163498 AFTER F18 IASC 035 AND IASC 064
		74A870645-1011/ 791660-10	163699 THRU 163782; ALSO 163427 THRU 163510 AFTER F18 IASC 064
		74A870645-1011/ 791660-11	163985 AND UP; ALSO 163427 THRU 163782 AFTER F18 IASC 064
		74A870645-1011/ 791660-12	163429 THRU 164724 AFTER F18 IASC 064
	5 13C-390	74A870645-1015/ 791660-12	163429 THRU 165532 AFTER F18 IASC 095
Voice/Data Recorder RO-646/ASH-39(V) (CSFIRS)	—	552474-30-01/ 173877-01-01	163429 THRU 164279 AFTER F/A-18 AFC 258

LEGEND

1 FLIR POD CUM 8 THRU 123.

2 NFLR POD CUM 6 THRU 25.

3 HARM BLOCK 3 AND UP OPERATES ONLY WITH DIGITAL DATA COMPUTER CONFIG/IDENT 89C AND UP INSTALLED.

4 PROGRAM LOAD CONFIG/IDENT NUMBER 87D-003 IS PREFERRED CONFIGURATION OVER PROGRAM LOAD CONFIG/IDENT NUMBER 87D-002 IN MEMORY UNIT.

5 REFER TO WP004 00 TO DETERMINE PROGRAM LOAD CONFIG/IDENT NUMBER FOR WRA'S INSTALLED ON AIRCRAFT.

6 MUST BE USED AS A SET, DO NOT MIX PER AIRCRAFT.

7 FLIR POD CUM 124 AND UP; ALSO CUM 8 THRU 123 AFTER F/A-18 AVC JAX-AV-011.

8 DIGITAL DATA COMPUTER CONFIG/IDENT 09C AND UP IS LOADABLE ONLY ON TYPE 8 (XN8) OR NEWER DIGITAL DATA COMPUTERS.

9 CLASSIFIED PROGRAM CONTACT TYCOM FOR PROGRAM IDENTIFICATION.

10 PROGRAM LOAD CONFIG/IDENT NUMBER 87X-034 IS PREFERRED CONFIGURATION.

11 MIXING OF 936E918G1 AND 936E918G2 PER AIRCRAFT IS ALLOWED.

Table 2. Current Program Load Configurations

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
Air Data Computer CP-1334/A	—	4031000-915/ 4031000-915	163427 THRU 164279
Air Data Computer CP-1334A/A	[1] 91X-010	74A870656-1001/ 4031000-920	164627 AND UP AFTER ECP 206R2
	[1] 93X-022	74A870656-1004/ 4031000-920	164627 AND UP AFTER F18 IASC 065
Armament Computer CP-1342/AYQ-9(V)	[1] 15C-621U or 15C-696V	74A870620-1093/ 7959650-108, -109	163429 THRU 165206
	[1] 17C-608U Boot Ver 12	74A870620-1093/ 7959650-108, -109	163429 THRU 165206
Armament Computer CP- 2218/AYK-22(V)	[1] 15C+530U or 15C-532V	74A870686 -1009/ 82370-01	165207 AND UP
	[1] 17C-551U	74A870686 -1011/ 82370-01	165207 AND UP
Command Launch Computer CP-1001( )/AWG	[1] 005	[4] ( )/ 704AS5961-3	163427 AND UP
	[1] 007	[4] ( )/ 704AS5961-3	163427 AND UP
Computer-Power Supply CP-1325/APG-65	[1] 11C-800U	74A870619-1037/ 3525681-155	163429 AND UP AFTER F18 IASC 080
	[1] 13C-800U	74A870619-1041/ 3525681-155	163429 THRU 164888 AFTER F18 IASC 092
Control-Converter C-10382/A	[1] 93C-007	74A870624-1025/ 7959750-007	163985 THRU 165532 AFTER F18 IASC 100
	[1] 15C-002	74A870624-1025/ 7959750-007/009	163427 AND UP AFTER F18 IASC 116
Controller-Processor C-10661( )/AAS-38 (FLIR)	[1] 11X-001	3061270-2	163427 AND UP AFTER F18 IASC 082
	[1] 13X-001	3061270-2	163427 THRU 165532 AFTER F18 IASC 101

Table 2. Current Program Load Configurations (Continued)

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
Controller-Processor C-10661( )/AAS-38B (FLIR)	1 11X-103	260582	163427 AND UP AFTER F18 IASC 083
	1 13X-101	260582	163427 THRU 165532 AFTER F18 IASC 102
Data Transfer Interface Unit J-6008/A (DFIRS)	1 91C-011	74A870654-1007/ 136787-1	164725 AND UP; ALSO 164627 THRU 164724 AFTER F/A-18 AFC 126
Digital Computer Converter CP-1805/AAR-50 (NFLR)	1 89C-005	74A870649-102/6096500-110	163985 AND UP
	1 91C-002	74A870649-104/6096500-110	163985 AND UP AFTER F18 IASC 059
Digital Data Computer No. 1	1 15C-261U	74A870618-( )/ 13225797-01	163429 AND UP
	1 17C-203U	74A870618-( )/ 13225797-01	163429 AND UP
Digital Data Computer No. 2	1 15C+262U	74A870618-( )/ 13225797-01	163429 AND UP
	1 17C-204U	74A870618-( )/ 13225797-01	163429 AND UP
Digital Display Indicator	1 09C-017	74A870654-1009/ 129000-59	163985 AND UP AFTER F18 IASC 067
	1 15C-022	74A870654-1009/ 129000-59	163985 AND UP AFTER F18 IASC 115
Digital Map Computer CP-1802/ASQ-196	1 89C-011	74A870650-1005/ 8506200-911, -912, -913, -914	164220 AND UP; ALSO 163985 THRU 164219 AFTER F18 IASC 047
	1 13C-006	74A870650-1005/ 8506200-911, -912, -913, -914	163985 AND UP AFTER F18 IASC 113
Digital Memory Unit MU-928/ASQ-196	DM0305	74A870650-1003/ 8505300-911, -914	163427 AND UP

Table 2. Current Program Load Configurations (Continued)

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
	DM0607	74A870650-1005/ 8505300-913	163427 AND UP AFTER F18 IASC 067
Inertial Navigation Unit CN-1561/ASN-130A	1 84B02	74A870617-1010/ 879010-01-84B02 or 879010-02-84B02	163427 THRU 164068 AFTER F18 IASC 021
Inertial Navigation Unit CN-1649/ASN-139	1 90X-L94( )	886401/ 886401-2	163427 AND UP
	1 90X-L99( )	886401/ 886401-3	163427 AND UP
Intercommunication Amplifier-Control AM-7360/A	—	74A870633-1009/ 5150100-2	164627 AND UP AFTER ECP 292
	—	74A870633-1011/ 5150100-2	164627 AND UP
	—	74A870633-1016/ 5150100-2	163429 AND UP AFTER F18 IASC 081
	—	74A870633-1018/ 5150100-2	163429 AND UP AFTER F18 IASC 114
Interconnecting Box J-3656/ASQ-173 (LDT)	1 102 or 103	71320600-019/ 71320600-019	163427 AND UP
Memory Unit MU-860B/ASQ-194	1 87D-004	74A870646-1002/ 791700-4	163427 AND UP
MIDS Radio Terminal RT-1765(C)/USQ-140(V)(C)	1 E6D or 45D or E76	P600A58-1	163429 THRU 165687
Radar Receiver R-2484/APG-73	1 48Z	74A870683-1005/ 3525026-110	164627 AND UP
	1 68	74A870683-1007/ 3525026-110	164627 THRU 165532 AFTER F18 IASC 094
Radar Data Processor CP-2062/APG-73	1 13C-702U/V	74A870666-1037/ 3525046-110	164627 THRU 165532 AFTER F18 IASC 093
	1 15C+704U	74A870666-1037/ 3525046-110	164627 THRU 165687 AFTER F18 IASC 111

Table 2. Current Program Load Configurations (Continued)

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
Radio Receiver (GPS) R-2512A/U	—	613-8379-007	164945 AND UP; ALSO 163429 THRU 164912 AFTER F/A-18 AFC 175 PT 2
Radio Receiver-Transmitter (CIT) RT-1763/APX-111(V)	[1] 15C-003U or 15C-002	74A870657-TBD/ 1007101G-10	165222 AND UP
	[1] 17C-003U	74A870685-1003	165222 AND UP
Receiver-Transmitter RT-1250/ARC	—	622-4016-001/ 622-4016-001	163427 AND UP
Receiver-Transmitter RT-1250A/ARC	—	622-6321-001/ 622-6321-001	163427 AND UP
Receiver-Transmitter RT-1556( )/ARC	—	622-9878-002/ 622-9878-002	164898 AND UP; ALSO 163427 THRU 164897 AFTER F/A-18 AFC 185
Receiver-Transmitter RT-1824(C)/ARC	[1] 15C-038	822-1133-001	163429 THRU 165687
Receiver-Transmitter-Processor RT-1379A/ASW	—	622-5663-002/ 622-5663-002	163427 AND UP
Roll-Pitch-Yaw Computer CP-1330/ASW-44	[1] 113	897E518G310 Version 8.5/ 897E518G310	[2] 163427 THRU 163510 AFTER F/A-18 AFC 136
Roll-Pitch-Yaw Computer CP-1330A/ASW-44	[1] 91C*002 or 112	936E918G4 Version 10.3	[2] 164754 AND UP AFTER F18 IASC 053
	[3] 91C*004 or 117	936E918G6 Version 10.5.1	[2] 163699 AND UP
	[3] 91C*004 or 117	936E918G6 Version 10.7	

Table 2. Current Program Load Configurations (Continued)

Programmable Units	Program Load CONFIG/ IDENT Number	Program Part Number/ Unit Part Number	Aircraft Effectivity
Signal Data Computer CP-1726/ASQ-194	[1] 91C-322	74A870645-1009/ 791660-8	163499 THRU 163510; ALSO 163427 THRU 163498 AFTER F18 IASC 035 AND F18 IASC 052
		74A870645-1009/ 791660-10	163699 THRU 163782; ALSO 163427 THRU 163510 AFTER F18 IASC 052
		74A870645-1009/ 791660-11	163985 AND UP; ALSO 163427 THRU 163782 AFTER F18 IASC 052
		74A870645-1009/ 791660-12	163429 THRU 164724 AFTER F18 IASC 052
	[1] 13C-390	74A870645-1015/ 791660-12	163429 THRU 165532
Voice/Data Recorder RO-646/ASH-39(V) (CSFIRS)	—	552474-30-01/ 173877-01-01	163429 THRU 164279 AFTER F/A-18 AFC 258

LEGEND

[1] REFER TO WP004 00 TO DETERMINE PROGRAM LOAD CONFIG/IDENT NUMBER FOR WRA'S INSTALLED ON AIRCRAFT.

[2] MUST BE USED AS A SET, DO NOT MIX PER AIRCRAFT.

[3] WITH ROLL-PITCH-YAW COMPUTER CP-1330/A/ASW-44 AND DIGITAL DATA COMPUTER CONFIG/IDENT 11C, REFER TO WP004 00 TO DETERMINE PROGRAM LOAD CONFIG/IDENT NUMBER FOR WRA'S INSTALLED ON AIRCRAFT.

[4] CLASSIFIED PROGRAM. CONTACT TYCOM FOR PROGRAM IDENTIFICATION.

[5] MIXING OF 936E918G1 AND 936E918G2 PER AIRCRAFT IS ALLOWED.

[6] FLIR POD CUM 124 AND UP; ALSO CUM 8 THRU 123 AFTER F/A-18 AVC JAX-AV-011.

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

PROGRAM LOAD CONFIG/IDENT VERIFICATION

Reference Material

Line Maintenance Procedures A1-F18AC-LMM-000
Line Maintenance Access Doors A1-F18AC-LMM-010
Software Configuration Manual A1-F18AC-SCM-000
Program Load Versions WP003 00
Component Locator WP005 00

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Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 126	-	Deployable Flight Incident Recorder Set (ECP MDA-F/A-18-00321R1C1)	15 Feb 92	-

1. INTRODUCTION.

2. This work package provides procedures to display program load CONFIG/IDENT numbers of programmable units (Table 1, Table 2, and Table 3) and to replace the unit when the program load CONFIG/IDENT number is not correct (Table 4). Fault isolation procedures for MC CONFIG caution are provided using the status of the customer identification display (Table 5 and Table 6). Unit address 28 (MC1) MI addresses are also provided (Table 7).

Table 1. CONFIG/IDENT Verification - F/A-18A AND F/A-18B

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components None **Related Systems Required** Armament Computer Avionics Cooling System Control-Converter C-10382/A Digital Data Computer No. 1 Digital Data Computer No. 2 Electrical System Flight Control System Forward Looking Infrared System Inertial Navigation System Laser Detector Tracker System Multipurpose Display Group Radar System **Support Equipment Required** None **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. WITH DIGITAL DATA COMPUTER CONFIG/IDENT 92A AND UP (WP003 00), the number of CAUTIONS on the Digital Display Indicator has been limited to 21. When this procedure requires the presence of a CAUTION or requires that the CAUTION does not exist and 21 CAUTIONS are already displayed, hydraulic power must be applied (A1-F18AC-LMM-000) to provide space for the required CAUTION indication.		
a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set and hold 1 and 2 switches to B ON for 3 seconds.	Switches remain on (latched).	1. If switches unlatch in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000).

Table 1. CONFIG/IDENT Verification - F/A-18A AND F/A-18B (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
		2. If both switches do not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00). 3. If one switch does not remain on, replace GND PWR Control Panel Assembly (A1-F18AC-420-300, WP023 00).
c. On left and right Digital Display Indicators (LDDI and RDDI), set power switch to NIGHT or DAY and allow 2 minute warmup. Adjust BRT and CONT controls for best display.	LDDI and RDDI have display and center pushbutton switch on bottom row is labeled MENU.	1. No display on LDDI: ON F/A-18A do table 1 (A1-F18AC-745-200, WP006 00). ON F/A-18B do table 1 (A1-F18AC-745-200, WP007 00). 2. No display on RDDI: ON F/A-18A, do table 2 (A1-F18AC-745-200, WP006 00). ON F/A-18B, do table 2 (A1-F18AC-745-200, WP007 00). 3. STANDBY is displayed: ON F/A-18A, do table 1 (A1-F18AC-745-200, WP004 00). ON F/A-18B, do table 1 (A1-F18AC-745-200, WP005 00). 4. BRT or CONT controls do not affect display. Replace Digital Display Indicator (A1-F18AC-745-300, WP004 00).
d. To verify the program load CONFIG/IDENT number of the Radar system, do substeps below: (1) On SNSR pod control box panel assembly (SNSR panel), set RADAR switch to OPR. (2) If this is the last system to be verified, go to step k.		

Table 1. CONFIG/IDENT Verification - F/A-18A AND F/A-18B (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
e. To verify the program load CONFIG/IDENT number of Inertial Navigation system, do substeps below:		
(1) On SNSR panel, set INS switch to NORM.		
(2) If this is the last system to be verified, go to step k.		
f. To verify the program load CONFIG/IDENT of the Armament Computer (SMS), do substeps below:		
(1) On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000).
		2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
(2) If this is the last system to be verified, go to step k.		
g. To verify the program load CONFIG/IDENT number of the Laser Detector Tracker (LTD) system, do substeps below:		
(1) On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000).
(2) On SNSR panel, set LST/CAM switch to ON.		2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
(3) On LDDI, do substeps below:		
(a) Press MENU pushbutton switch.	LDDI has MENU display (figure 2, detail A).	Replace Digital Display Indicator (A1-F18AC-745-300, WP004 00).

Table 1. CONFIG/IDENT Verification - F/A-18A AND F/A-18B (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(b) Press LST pushbutton switch.	LDDI has LST control display (detail B).	Replace Digital Display Indicator (A1-F18AC-745-300, WP004 00).
(c) Press LST pushbutton switch.	LST pushbutton legend is boxed (detail B).	Replace Digital Display Indicator (A1-F18AC-745-300, WP004 00).
(4) If this is the last system to be verified, go to step k.		
h. To verify the program CONFIG/IDENT of the Forward Looking Infrared (FLIR) System do substeps below:		
(1) On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-300, WP006 00).
(2) On SNSR panel, set FLIR switch to STBY.		
(3) If this is the last system to be verified, go to step k.		
(4) If this is the last system to be verified, go to step k.		
i. To verify the program load CONFIG/IDENT number of the Flight Control (FCCA and FCCB) system, do substeps below:		
(1) On GND PWR control panel assembly, set and hold 4 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-300, WP006 00).
(2) If this is the last system to be verified, go to step k.		

Table 1. CONFIG/IDENT Verification - F/A-18A AND F/A-18B (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
j. To verify the program load CONFIG/IDENT number of the Electronic Countermeasures System (ALQ-126B), do substeps below:		
(1) On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
(2) On the ECM Control Panel, set the ECM switch to STBY.	On LH advisory and threat warning indicator panel, STBY illuminates for 3 to 4 minutes and then goes off.	1. If light did not come on, do table 1 (A1-F18AC-760-200, WP016 00). 2. If light did not go off, do table 2 (A1-F18AC-760-200, WP016 00).
(3) If this is the last system to be verified, go to step m.		
k. To verify the program load CONFIG/IDENT number of the Radar Warning Receiver System (ALR-67), do substeps below:		
(1) On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).

Table 1. CONFIG/IDENT Verification - F/A-18A AND F/A-18B (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) On Control-Indicator press and release POWER ON switch.	1. POWER ON light comes on.	Do table 1, (A1-F18AC-760-200, WP033 00).
	2. BIT, OFFSET, DISPLAY, and SPECIAL lights come on.	Replace Control-Indicator C-10250/ALR-67(V) (A1-F18AC-760-300, WP056 00).
	3. Forward Azimuth Indicator has status and emitter display.	1. No display on Azimuth Indicator. Do table 2, (A1-F18AC-760-200, WP033 00). 2. Priority display cycles A to N continuously. Do substeps below: a. Remove Radar Receiver R-2055A/ALR-67(V) (A1-F18AC-760-300, WP054 00). b. On Radar Receiver, if circuit breaker CB1 is in OFF position (tripped), do step c. If CB1 is in ON position, do table 5, (A1-F18AC-760-200, WP034 00). c. Reset CB1 and install Radar Receiver R-2055A/ALR-67(V) (A1-F18AC-760-300, WP054 00). If malfunction still exists do table 5, (A1-F18AC-760-200, WP034 00). 3. Azimuth Indicator displays flashing B. Do table 5, (A1-F18AC-760-200, WP033 00).
	4. On F/A-18B, rear Azimuth Indicator has status and emitter display.	1. No display on rear Azimuth Indicator. Do table 5, (A1-F18AC-760-200, WP033 00). 2. Rear Azimuth Indicator displays flashing B. Do table 5, (A1-F18AC-760-200, WP033 00).
(3) If this is the last system to be verified, go to step m.		

Table 1. CONFIG/IDENT Verification - F/A-18A AND F/A-18B (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
l. To verify the program load CONFIG/IDENT number of digital data computer no. 1 (MC1), digital data computer No. 2 (MC2), and Control-Converter C-10382/A (CSC), do step m.		
m. On RDDI, do substeps below:		
(1) Press and release MENU pushbutton switch until BIT pushbutton appears.	RDDI has MENU display (figure 1, detail A).	Replace Right Digital Display Indicator (A1-F18AC-745-300, WP004 00).
(2) Press BIT pushbutton switch.	RDDI has BIT control display (detail B).	Replace Right Digital Display Indicator (A1-F18AC-745-300, WP004 00).
(3) Observe BIT status display for MC1 and MC2.	BIT status of MC1 and MC2 indicates GO (detail B).	Replace applicable Digital Data Computer No. 1 (A1-F18AC-741-300, WP003 00) or Digital Data Computer No. 2 (A1-F18AC-741-300, WP004 00).
(4) Observe BIT status of systems to be verified when turn on tests are complete - SF TEST not displayed.	BIT status of systems which have been selected to verify program load CONFIG/IDENT number indicates GO, DEGD, OH, or DEGD OH (detail B).	Do applicable system troubleshooting if system to be verified BIT status is NO GO, NOT RDY, or RESTRT.
(5) Press MAINT pushbutton switch.	RDDI has maintenance BIT control display (detail D).	Replace Right Digital Display Indicator (A1-F18AC-745-300, WP004 00).
(6) Press CONFIG pushbutton switch.	1. RDDI has configuration display (detail G).	Replace Right Digital Display Indicator (A1-F18AC-745-300, WP004 00).
	2. Customer identifier (USN) is displayed (detail G), on configuration display and MC CONFIG caution is not displayed (detail E).	1. If MC CONFIG is displayed and USN customer identifier is not displayed, do table 5. 2. If MC CONFIG is displayed and USN customer identifier is displayed, do table 6.
(7) Record program load identification numbers.		

Table 1. CONFIG/IDENT Verification - F/A-18A AND F/A-18B (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE When a system is off or unable to communicate with the Mission Computer, program load CONFIG/IDENT number number displayed is the program load CONFIG/IDENT number of the unit which was the last to successfully communicate with the Mission Computer and may not be the program load CONFIG/IDENT number of the unit installed. When MC1, MC2, ALQ-126B, ALR-67, RDR, SMS, INS, FCCA, or FCCB have incompatible basic program loads or incompatible program identification extensions, a S/W CONFIG caution appears on the RDDI caution line (figure 1, detail E) and a line is displayed through the CONFIG/IDENT number which is not compatible. Determine that BIT status indicates MUX communication before verifying program load CONFIG/IDENT numbers.		
n. Determine if the latest software/firmware is installed.	1. All program load CONFIG/ IDENT numbers are correct except MC1 and MC2 and no S/W CONFIG caution displayed (table 2, WP003 00	Do table 4 (line through CONFIG/IDENT indicates system with wrong load status).
	2. MC1 or MC2 program load CONFIG/IDENT numbers are correct (table 2, WP003 00) and S/W CONFIG and MC CONFIG cautions are not displayed on caution line.	1. Reload Digital Data Computer No. 1 or Digital Data Computer No. 2 that has the wrong program load CONFIG/IDENT number (line through program load CONFIG/ IDENT number indicates system with wrong load status) (WP006 00). 2. If reload is attempted and correct program load CONFIG/ IDENT number is still wrong, replace applicable Digital Data Computer No. 1 (A1-F18AC-741-300, WP003 00) or Digital Data Computer No. 2 (A1-F18AC-741-300, WP004 00).
o. On SNSR panel, set RADAR, FLIR, LST/CAM and INS switches to OFF.		
p. On LDDI and RDDI set power switch to OFF.		

Table 1. CONFIG/IDENT Verification - F/A-18A AND F/A-18B (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
q. On the ECM Control Panel set the DISPENSER switch to OFF.		
r. On the Control-Indicator press the POWER on switch to turn power OFF.	1. Control-Indicator buttons disappear. 2. Forward Azimuth Indicator display disappears.	
s. Remove electrical power (A1-F18AC-LMM-000).		

Table 2. CONFIG/IDENT Verification - F/A-18C AND F/A-18D

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components None **Related Systems Required** Air Data Computer CP-1334A/A Armament Computer Avionics Cooling System Control-Converter C-10382/A Data Transfer Interface Unit J-6008/A - 164725 AND UP; ALSO 164627 THRU 164724 AFTER F/A-18 AFC 126 Digital Data Computer No. 1 Digital Data Computer No. 2 Digital Map Computer CP-1802/ASQ-196 - 163985 AND UP Electrical System Flight Control System Forward Looking Infrared System Inertial Navigation System Laser Detector Tracker System Multipurpose Display Group Navigation Forward Looking Infrared System - 163985 AND UP WITH NFLR INSTALLED Radar System Signal Data Computer CP-1726/ASQ-194 **Support Equipment Required** None **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. The number of CAUTIONS on the Digital Display Indicator has been limited to 21. When this procedure requires the presence of a CAUTION or requires that the CAUTION does not exist and 21 CAUTIONS are already displayed, hydraulic power must be applied (A1-F18AC-LMM-000) to provide space for the required CAUTION indication.		
a. Apply electrical power (A1-F18AC-LMM-000).		

Table 2. CONFIG/IDENT Verification - F/A-18C AND F/A-18D (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. On GND PWR control panel assembly, set and hold 1 and 2 switches to B ON for 3 seconds.	Switches remain on (latched).	1. If switches unlatch in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If both switches do not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00). 3. If one switch does not remain on, replace GND PWR Control Panel Assembly (A1-F18AC-420-300, WP023 00).
c. On left and right Digital Display Indicators (LDDI and RDDI), set power switch to NIGHT or DAY and allow 2 minute warmup. Adjust BRT and CONT controls for best display.	LDDI and RDDI have display and center pushbutton switch on bottom row is labeled MENU (figure 1, detail A).	1. No display on LDDI: ON F/A-18C 163427 THRU 163782 , do table 1 (A1-F18AC-745-200, WP006 00). ON F/A-18D 163434 THRU 163778, do table 1 (A1-F18AC-745-200, WP007 00). ON F/A-18C 163985 AND UP, do table 1 (A1-F18AG-745-200, WP006 00). ON F/A-18D 163986 AND UP, do table 1 (A1-F18AG-745-200, WP007 00). 2. No display on RDDI: ON F/A-18C 163427 THRU 163782, do table 2 (A1-F18AC-745-200, WP006 00). ON F/A-18D 163434 THRU 163778, do table 2 (A1-F18AC-745-200, WP007 00). ON F/A-18C 163985 AND UP , do table 2 (A1-F18AG-745-200, WP006 00). ON F/A-18D 163986 AND UP, do table 2 (A1-F18AG-745-200, WP006 00).

Table 2. CONFIG/IDENT Verification - F/A-18C AND F/A-18D (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
		3. STANDBY is displayed: ON F/A-18C 163427 THRU 163782 do table 2, (A1-F18AC-745-200, WP004 00). ON F/A-18D 163434 THRU 163778, do table 2, (A1-F18AC-745-200, WP005 00). ON F/A-18C 163985 AND UP, do table 2 (A1-F18AG-745-200, WP004 00). ON F/A-18D 163986 AND UP, do table 2 (A1-F18AG-745-200, WP005 00). 4. BRT or CONT controls do not affect display. ON 163427 THRU 163782, replace Digital Display Indicator (A1-F18AC-745-300, WP004 00). ON163985 AND UP, replace Digital Display Indicator (A1-F18AG-745-300, WP004 00).
d. To verify the program load CONFIG/IDENT number of the Radar (RDR) system do substeps below: (1) On SNSR pod control box panel assembly (SNSR panel), set RADAR switch to STBY. (2) If this is the last system to be verified, go to step l.		
e. To verify the program load CONFIG/IDENT number of the Inertial Navigation Unit (INS), do substeps below: (1) On SNSR panel, set INS switch to NORM. (2) If this is the last system to be verified, go to step l.		
f. To verify the program load CONFIG/IDENT number of the Armament Computer (SMS), do substeps below:		

Table 2. CONFIG/IDENT Verification - F/A-18C AND F/A-18D (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
(2) If this is the last system to be verified, go to step l.		
g. To verify the program load CONFIG/IDENT number of the Forward Looking Infrared System (FLIR), do substeps below:		
(1) On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
(2) On SNSR panel, set FLIR switch to STBY.		
(3) If this is the last system to be verified, go to step l.		
h. To verify the program load CONFIG/IDENT number of the Laser Detector Tracker System (LDT), do substeps below:		
(1) On GND PWR control panel assembly set, and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
(2) On SNSR panel:		

Table 2. CONFIG/IDENT Verification - F/A-18C AND F/A-18D (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
ON 163427 THRU 163782, set LST/CAM switch to ON. ON 163985 AND UP, set LST/NFLR switch to ON. (3) On LDDI, do substeps below:		
(a) Press and release MENU pushbutton switch until LST pushbutton appears.	Indicator has MENU display (figure 2, detail A).	ON 163427 THRU 163782, replace Left Digital Display Indicator (A1-F18AC-745-300, WP004 00). ON 163985 AND UP, replace Left Digital Display Indicator (A1-F18AG-745-300, WP004 00).
(b) Press LST pushbutton switch.	Indicator has LST control display (figure 2, detail B).	ON 163427 THRU 163782, replace Left Digital Display Indicator (A1-F18AC-745-300, WP004 00). ON 163985 AND UP, replace Left Digital Display Indicator (A1-F18AG-745-300, WP004 00).
(c) Press LST pushbutton switch.	LST pushbutton legend is boxed (figure 2, detail B).	ON 163427 THRU 163782, replace Left Digital Display Indicator (A1-F18AC-745-300, WP004 00). ON 163985 AND UP, replace Left Digital Display Indicator (A1-F18AG-745-300, WP004 00).
(4) If this is the last system to be verified, go to step l. i. ON 163985 AND UP WITH NFLR INSTALLED, to verify the program load CONFIG/IDENT number of the Navigation Forward Looking Infrared System (NFLR), do substeps below:		
(1) On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000).

Table 2. CONFIG/IDENT Verification - F/A-18C AND F/A-18D (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
		2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
(2) On SNSR panel, set LST/NFLR switch to ON.		
(3) Allow navigation infrared receiving set (NFLR) five minutes to cooldown.		
(4) If this is the last system to be verified, go to step l.		
j. To verify the program load CONFIG/IDENT number of the Flight Control System (FCCA, FCCB) do substeps below:		
(1) On GND PWR control panel assembly, set and hold 4 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000).
		2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
(2) If this is the last system to be verified, go to step l.		

Table 2. CONFIG/IDENT Verification - F/A-18C AND F/A-18D (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
j1. To verify the program load CONFIG/IDENT number of the Countermeasures Dispensing System (ALE-47), do substeps below:		
(1) On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
(2) On the ECM Control Panel, set the DISPENSER switch to ON.		
(3) If this is the last system to be verified, go to step l.		
j2. To verify the program load CONFIG/IDENT number of the Electronic Countermeasures System (ALQ-126B), do substeps below:		
(1) On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
(2) ON F/A-18C; F/A-18D 163434 THRU 163778, on ECM control panel assembly, set the ECM mode switch to STBY.	In cockpit, on LH advisory and threat warning indicator panel, STBY comes on for 3 to 4 minutes and then goes out.	1. If STBY light did not come on, do table 1 (A1-F18AE-760-200, WP016 00). 2. If STBY light did not go off, do table 2 (A1-F18AE-760-200, WP016 00).

Table 2. CONFIG/IDENT Verification - F/A-18C AND F/A-18D (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) ON F/A-18D 163986 AND UP, on ECM control panel assembly, set the ECM mode switch to STBY.	In cockpit, on LH advisory and threat warning indicator panel, STBY comes on for 3 to 4 minutes and then goes out.	1. If STBY light did not come on, do table 1 (A1-F18AE-760-200, WP016 01). 2. If STBY light did not go off, do table 2 (A1-F18AE-760-200, WP016 00).
(4) If this is the last system to be verified, go to step l.		
j3. To verify the program load CONFIG/IDENT number of the Electronic Countermeasures System (ALQ-165), do substeps below:		
(1) On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
(2) ON F/A-18C; F/A-18D 163434 THRU 163778, on ECM control panel assembly, set the ECM mode switch to STBY.	In cockpit, on LH advisory and threat warning indicator panel, STBY comes on for 3 to 4 minutes and then goes out.	1. If STBY light did not come on, do table 1 (A1-F18AE-760-200, WP019 00). 2. If STBY light did not go off, do table 1 (A1-F18AE-760-200, WP018 00).
(3) ON F/A-18D 163986 AND UP, on ECM control panel assembly, set the ECM mode switch to STBY.	In cockpit, on LH advisory and threat warning indicator panel, STBY comes on for 3 to 4 minutes and then goes out.	1. If STBY light did not come on, do table 2 (A1-F18AE-760-200, WP016 01). 2. If STBY light did not go off, do table 1 (A1-F18AE-760-200, WP018 00).
(4) If this is the last system to be verified, go to step l.		
j4. To verify the program load CONFIG/IDENT number of the Radar Warning Receiver System (ALR-67), do substeps below:		

Table 2. CONFIG/IDENT Verification - F/A-18C AND F/A-18D (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
(2) On ALR-67 Control- Indicator, press the POWER ON switch.	1. Control-Indicator buttons light up with descriptions. 2. Power button displays ALR-67 in red and ON in green.	Do table 1, (A1-F18AE-760-200, WP045 00).
(3) If this is the last system to be verified, go to step l.		
k. To verify the program load CONFIG/IDENT number of the numbers for the listed systems, go to step l. (1) digital data computer no. 1 MC1) (2) digital data computer no. 2 (MC2) (3) Control-Converter C-10382/A (CSC) (4) LDDI (5) RDDI (6) Signal Data Computer CP-1726/ASQ-194 (SDC) (7) Memory Unit MU-860B/ ASQ-194 (MU) (8) Air Data Computer CP-1334A/A (ADC) (9) ON 163985 AND UP, Digital Map Computer CP-1802/ASQ-196 (DMS) (10) ON 164725 AND UP; ALSO 164627 THRU 164724 AFTER F/A-18 AFC 126, Data Transfer Interface Unit J-6008/A (DFIRS).		
l. On RDDI, do substeps below:		

Table 2. CONFIG/IDENT Verification - F/A-18C AND F/A-18D (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Press and release MENU pushbutton switch until BIT pushbutton appears.	Indicator has MENU display (figure 1, detail A).	Press and release MENU on LDDI. 1. If LDDI Displays MENU, ON 163427 THRU 163782, replace right Digital Display Indicator (A1-F18AC-745-300, WP004 00). ON 163985 AND UP, replace right Digital Display Indicator (A1-F18AG-745-300, WP004 00). 2. If LDDI does not display MENU, make sure MC1 and MC2 do not have same MC2 OFP loaded by reloading both MC1 and MC2 (WP006 00).
(2) Press BIT pushbutton switch.	Indicator has BIT Top level display (figure 1, detail C).	ON 163427 THRU 163782, replace Right Digital Display Indicator(A1-F18AC-745-300, WP004 00). ON 163985 AND UP, replace Right Digital Display Indicator (A1-F18AG-745-300, WP004 00).
(3) Press CONFIG pushbutton switch.	1. Indicator has configuration display (figure 1, detail F).	ON 163427 THRU 163782, replace Right Digital Display Indicator (A1-F18AC-745-300, WP004 00). ON 163985 AND UP, replace Right Digital Display Indicator (A1-F18AG-745-300, WP004 00).
	2. Customer identifier (USN) is displayed (detail F), on configuration display and MC CONFIG caution is not displayed (figure 1, detail E).	1. If MC CONFIG is displayed and USN customer identifier is not displayed, do table 5. 2. If MC CONFIG is displayed and USN customer identifier is displayed, do table 6.
(4) Record program load identification numbers.		

Table 2. CONFIG/IDENT Verification - F/A-18C AND F/A-18D (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE When MUX communication has not been established with a system since MC1 turn on the program load CONFIG/IDENT number of that system is not displayed on the configuration display and CONFG advisory is displayed. CONFG advisory indicates that all system configurations have not been checked for compatibility. When MC1, MC2, RDR, SMS, INS, FCSA, FCSB, SDC, MU, ADC, CSC, ALQ-126B, ALQ-165, ALE-47, ALR-67, and ON 163985 AND UP, DMS, have incompatible basic program loads or incompatible program identification extensions, a S/W CONFIG caution appears on the DDI caution line (figure 1, detail E) and a line is displayed through the program load CONFIG/IDENT number which is not correct.		
m. Determine if latest software is installed.	1. MC1 or MC2 program load CONFIG/IDENT numbers are correct (table 2, WP003 00) and S/W CONFIG and MC CONFIG cautions are not displayed on caution line.	1. Reload Digital Data Computer No. 1 or Digital Data Computer No. 2 that has the wrong program load CONFIG/IDENT number (line through program load CONFIG/IDENT number indicates system with wrong load status) (WP006 00).

Table 2. CONFIG/IDENT Verification - F/A-18C AND F/A-18D (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
		2. If reload is attempted and correct program load CONFIG/IDENT number is still wrong, replace applicable Digital Data Computer No. 1 (A1-F18AE-741-300, WP003 00) or Digital Data Computer No. 2 (A1-F18AE-741-300, WP004 00).
	2. All program load CONFIG/IDENT numbers are correct (except MC1 and MC2 and no S/W CONFIG caution displayed) (table 2, WP003 00).	Do table 4 (line through program load CONFIG/IDENT number indicates system with wrong load status).
n. On SNSR panel, do substeps below: (1) Set RADAR, FLIR and INS switches to OFF. (2) ON 163427 THRU 163782, set LST/CAM switch to OFF. (3) ON 163985 AND UP, set LST/NFLR switch to OFF. o. On the ECM Control Panel set the DISPENSER switch to OFF. p. On the ECM Control Panel set the ECM switch to OFF.		
q. On the Control-Indicator, press the POWER ON switch to turn power off.	1. Control-Indicator buttons disappear. 2. Forward Azimuth Indicator display disappears.	
r. On LDDI and RDDI, set power switch to OFF. s. Remove electrical power (A1-F18AC-LMM-000).		

Table 3. CONFIG/IDENT Verification - Command Launch Computer

Procedure	Normal Indication	Remedy for Abnormal Indication
Command Launch Computer CP-1001( )/AWG **Related Systems Required** Armament Computer Avionics Cooling System Digital Data Computer No. 1 Digital Data Computer No. 2 Electrical System Multipurpose Display Group **Support Equipment Required** None **Materials Required** None **NOTE** For Component Locator, refer to WP005 00.		
a. Open door 14R (A1-F18AC- LMM-010).		
b. On Armament Computer CP-1342/AVQ-9(V), set ARMAMENT switches to 64 for station 2 or 8.		
c. On Armament Computer CP-2218/AYK-22(V), set Weapon Insertion Panel switches to 64 for station 2 or 8 (A1-F18AH-740-200, WP007 00).		
d. Apply electrical power (A1-F18AC-LMM-000).		
e. On GND PWR control panel assembly, set and hold 1, 2 and 3 switches to B ON for 3 seconds.	Switches remain on (latched).	1. If switches unlatch in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000).

Table 3. CONFIG/IDENT Verification - Command Launch Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
		2. If all switches do not remain on, do Ground Power Switching System Test(A1-F18AC-420-200, WP006 00). 3. If one switch does not remain on, replace GND PWR Control Panel Assembly (A1-F18AC-420-300, WP023 00).
f. On left and right Digital Display Indicators (LDDI and RDDI), set power switch to NIGHT or DAY and allow 2 minute warmup. Adjust BRT and CONT controls for best display.	LDDI and RDDI have display and center pushbutton switch on bottom row is labeled MENU.	1. No display on LDDI: ON F/A-18C 163427 THRU 163782, do table 1 (A1-F18AC-745-200, WP006 00). ON F/A-18D 163434 THRU 163778, do table 1 (A1-F18AC-745-200, WP007 00). ON F/A-18C 163985 AND UP , do table 1 (A1-F18AG-745-200, WP006 00). ON F/A-18D 163986 AND UP, do table 1 (A1-F18AG-745-200, WP007 00). 2. No display on RDDI: ON F/A-18C 163427 THRU 163782, do table 2 (A1-F18AC-745-200, WP006 00). ON F/A-18D 163434 THRU 163778, do table 2 (A1-F18AC-745-200, WP007 00). ON F/A-18C 163985 AND UP , do table 2 (A1-F18AG-745-200, WP006 00). ON F/A-18D 163986 AND UP, do table 2 (A1-F18AG-745-200, WP006 00).

Table 3. CONFIG/IDENT Verification - Command Launch Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
		3. STANDBY is displayed: ON F/A-18C 163427 THRU 163782, do table 1 (A1-F18AC-745-200, WP004 00). ON F/A-18D 163434 THRU 163778, do table 1 (A1-F18AC-745-200, WP005 00). ON F/A-18C 163985 AND UP, do table 2 (A1-F18AG-745-200, WP004 00). ON F/A-18D 163986 AND UP, do table 2 (A1-F18AG-745-200, WP005 00). 4. BRT or CONT controls do not affect display. Replace Digital Display Indicator: 161353 THRU 163782 (A1-F18AC-745-300, WP004 00). 163985 AND UP (A1-F18AG-745-300, WP004 00).
g. WITH DIGITAL DATA COMPUTER CONFIG/IDENT 09C AND UP to verify the program load CONFIG/IDENT number of the Command Launch Computer CP-1001( )/AWG (CLC), do step k.		
h. On master arm control panel, press and release A/G switch.	A/G indicator light comes on.	ON F/A-18C, do table 1 (A1- F18AE-740-200, WP037 00). ON F/A-18D 163434 THRU 163778, do table 2 (A1-F18AE-740-200, WP037 00). ON F/A-18D 163986 AND UP, do table 3 (A1-F18AE-740-200, WP037 00).
i. On LDDI, do substeps below:		
(1) Press and release MENU pushbutton switch until STORES pushbutton selection is displayed.	MENU display appears on LDDI (figure 1, detail A).	Replace left Digital Display Indicator : 163427 THRU 163782 (A1-F18AC-745-300, WP004 00). 163985 AND UP (A1-F18AG-745-300, WP004 00).

Table 3. CONFIG/IDENT Verification - Command Launch Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Press STORES pushbutton switch.	Stores display appears with HARM pushbutton label displayed on LDDI.	Replace left Digital Display Indicator : 163427 THRU 163782 (A1-F18AC-745-300, WP004 00). 163985 AND UP (A1-F18AG-745-300, WP004 00).
(3) Press HARM pushbutton switch.	HARM display appears on LDDI with TOO pushbutton label displayed on LDDI.	Troubleshoot using AGM-88 HARM Armament Computer/ Command Launch Computer Interface Schematic and AGM-88 HARM TOO Mode Interface Schematic (A1-F18AE-740-500, WP063 00 and WP064 00).
(4) Press TOO pushbutton switch.	TOO mode display appears on LDDI with CLASS pushbutton label displayed on LDDI.	Troubleshoot using AGM-88 HARM Armament Computer/ Command Launch Computer Interface Schematic and AGM-88 HARM TOO Mode Interface Schematic (A1-F18AE-740-500, WP063 00 and WP064 00).
(5) Press CLASS pushbutton switch.	HARM CLASS selection display appears on LDDI with PAGE pushbutton label displayed on LDDI.	Troubleshoot using AGM-88 HARM Armament Computer/ Command Launch Computer Interface Schematic and AGM-88 HARM TOO Mode Interface Schematic (A1-F18AE-740-500, WP063 00 and WP064 00).
(6) Press and release PAGE pushbutton switch to last page of CLASS selection display.	Last page of CLASS selection display appears on LDDI with SID pushbutton label displayed on LDDI.	Troubleshoot using AGM-88 HARM Armament Computer/ Command Launch Computer Interface Schematic and AGM-88 HARM TOO Mode Interface Schematic (A1-F18AE-740-500, WP063 00 and WP064 00).
(7) Press SID pushbutton switch.	TOO mode display appears on LDDI with CLC 007 or CLC 010 displayed at top center of display.	Troubleshoot using AGM-88 HARM Armament Computer/ Command Launch Computer Interface Schematic and AGM-88 HARM TOO Mode Interface Schematic (A1-F18AE-740-500, WP063 00 and WP064 00).

Table 3. CONFIG/IDENT Verification - Command Launch Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
j. Go to step m.		
k. On LDDI, do substeps below:		
(1) Press and release MENU pushbutton switch until BIT pushbutton appears.	Indicator has MENU display (figure 1, detail A).	Press and release MENU on RDDI. 1. If RDDI Displays MENU, ON 163427 THRU 163782, replace left Digital Display Indicator (A1-F18AC-745-300, WP004 00). ON 163985 AND UP, replace left Digital Display Indicator (A1-F18AG-745-300, WP004 00). 2. If RDDI does not display MENU, make sure MC1 and MC2 do not have same MC2 OFP loaded by reloading both MC1 and MC2 (WP006 02).
(2) Press BIT pushbutton switch.	Indicator has BIT Top level display (detail C).	ON 163427 THRU 163782, replace Digital Display Indicator (A1-F18AC-745-300, WP004 00). ON 163985 AND UP, replace Digital Display Indicator (A1-F18AG-745-300, WP004 00).
(3) Press CONFIG pushbutton switch.	1. Indicator has configuration display (detail F).	ON 163427 THRU 163782, replace Digital Display Indicator (A1-F18AC-745-300, WP004 00). ON 163985 AND UP, replace Digital Display Indicator (A1-F18AG-745-300, WP004 00).
	2. Customer identifier (USN) is displayed (detail F), on configuration display and MC CONFIG caution is not displayed (detail E).	1. If MC CONFIG is displayed and USN customer identifier is not displayed, do table 5. 2. If MC CONFIG is displayed and USN customer identifier is displayed, do table 6.
(4) Record CLC program load CONFIG/IDENT number.		

Table 3. CONFIG/IDENT Verification - Command Launch Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE When MUX communication has not been established with a system since MC1 turn on the program load CONFIG/IDENT number of that system will not be displayed on the configuration display and CONFG advisory will be displayed. CONFG advisory indicates that all system configurations have not been checked for compatibility. When MC1, MC2, RDR, SMS, INS, FCSA, FCSB, SDC, MU, ADC, CSC, and ON 163985 AND UP, DMS, have incompatible basic program loads or incompatible program identification extensions, a S/W CONFIG caution will appear on the DDI caution line (figure 1, detail E) and a line will be drawn through the program load CONFIG/IDENT number which is not correct.		
l. Determine if latest CLC software/firmware is installed.	1. MC1 or MC2 program load CONFIG/IDENT numbers are correct (table 2, WP003 00) and S/W CONFIG and MC CONFIG cautions are not displayed on caution line.	1. Reload Digital Data Computer No. 1 or Digital Data Computer No. 2 that has the wrong program load CONFIG/IDENT number (line through program load CONFIG/IDENT number indicates system with wrong load status)(WP006 00). 2. If reload is attempted and correct program load CONFIG/IDENT number is still wrong, replace applicable Digital Data Computer No. 1 (A1-F18AE-741-300, WP003 00) or Digital Data Computer No. 2 (A1-F18AE-741-300, WP004 00).
	2. CLC program load CONFIG/IDENT number is correct (table 2, WP003 00).	Do table 4 (line through program load CONFIG/IDENT number indicates system with wrong load status).
m. On LDDI and RDDI, set power switch to OFF.		
n. Remove electrical power (A1-F18AC-LMM-000).		
o. Close door 14R (A1- F18AC-LMM-010).		

Table 4. Program Load CONFIG/IDENT Wrong

Support Equipment Required

None

Materials Required

None

Malfunction is caused by one of the items below:

Air Data Computer C-1334( )/A - F/A-18C AND F/A-18D
Armament Computer
Command Launch Computer CP-1001/AWG
Computer Power Supply CP-1325/APG-65 - F/A-18A AND F/A-18B; ALSO 163427 THRU 1642797; ALSO 164627 THRU 164897 BEFORE F/A-18 AFC-211
Control-Converter C-10382/A
Controller Processor C-10661( )/AAS-38
Data Transfer Interface Unit J-6008/A (DFIRS) - 164725 AND UP; ALSO 164627 THRU 164724 AFTER F/A-18 AFC 126
Digital Computer-Converter CP-1805/AAR50 - F/A-18C AND F/A-18D
Digital Display Indicator IP-1553/A - 163985 AND UP
Digital Map Computer CP-1802/ASQ-196 - F/A-18C AND F/A-18D
Interconnecting Box J-3656/ASQ-173
Inertial Navigation Unit CN-1561/ASN-130A or Inertial Navigation Unit CN-1649/ASN-139
Memory Unit MU-806/ASQ-194
Radar Data Processor CP-2062/APG-73 164898 AND UP; ALSO 164627 THRU 164897 AFTER F/A-18 AFC-211
Roll-Pitch-Yaw Computer CP-1330/ASW-44 (FCCA)
Roll-Pitch-Yaw Computer CP-1330/ASW-44 (FCCB)
Signal Data Computer CP-1726/ASQ-194 - F/A-18C AND F/A-18D

Procedure	No	Yes
a. Air Data Computer C-1334( )/A indicates wrong program load CONFIG/IDENT number do substeps below:		
(1) WITH DIGITAL DATA COMPUTER CONFIG/IDENT 91C AND UP, do substeps below:		
(a) Reload Air Data Computer C-1334A/A (WP006 00)	-	-
(b) If load status still wrong replace Air Data Computer C-1334A/A (A1-F18AC-560-300, WP003 00)	-	-
(2) Replace Air Data Computer C-1334( )/A (A1-F18AC-560-300, WP003 00)	-	-
b. Armament Computer indicates wrong program load CONFIG/IDENT number do substeps below:		
(1) Reload Armament Computer (WP006 00)	-	-

Table 4. Program Load CONFIG/IDENT Wrong (Continued)

Procedure	No	Yes
(2) If load status still wrong, replace Armament Computer CP-1342/AYQ-9(V) (A1-F18AC-740-300, WP006 00 or A1-F18AE-740-300, WP006 00) or Armament Computer CP-2218/AYK-22(V) (A1-F18AH-740-300, WP003 00)	-	-
c. Command Launch Computer CP-1001( )/AWG indicates wrong program load CONFIG/IDENT number do substeps below:		
(1) Reload Command Launch Computer CP-1001( )/AWG (WP006 00)	-	-
(2) If load status still wrong replace Command Launch Computer CP-1001( )/AWG (A1-F18AC-740-300, WP010 00 or A1-F18AE-740-300, WP011 00)	-	-
d. Computer Power Supply CP-1325/APG-65 indicates wrong program load CONFIG/IDENT number do substeps below:		
(1) WITH DIGITAL DATA COMPUTER CONFIG/IDENT 89C do substeps below:		
(a) Reload Computer Power Supply CP-1325/APG-65 (WP006 00)	-	-
(b) If load status still wrong replace Computer-Power Supply CP-1325/APG-65 (A1-F18AC-742-300, WP005 00)	-	-
(2) WITH DIGITAL DATA COMPUTER CONFIG/IDENT 91C AND UP do substeps below:		
(a) Make sure Computer Power Supply CP-1325/APG-65 part number is 3525681-155 then reload OFP (WP006 00)	-	-
(b) If load status still wrong replace Computer-Power Supply CP-1325/APG-65 (A1-F18AC-742-300, WP005 00)	-	-
(3) Replace Computer-Power Supply CP-1325/APG-65 (A1-F18AC-742-300, WP005 00)	-	-
e. Control-Converter C-10382/A indicates wrong program load CONFIG/IDENT number do substeps below:		
(1) WITH DIGITAL DATA COMPUTER CONFIG/IDENT 91C AND UP do substeps below:		
(a) Reload Control-Converter C-10382/A (WP006 00)	-	-
(b) If load status still wrong replace Control-Converter C-10382/A (A1-F18AE-741-300, WP005 00)	-	-
(2) Replace Control-Converter C-10382/A (A1-F18A( )-741-300, WP005 00	-	-
f. Controller-Processor C-10661( )/AAS-38 indicates wrong program load CONFIG/IDENT number do substep below:		

Table 4. Program Load CONFIG/IDENT Wrong (Continued)

Procedure	No	Yes
(1) Replace Controller-Processor C-10661( )/AAS-38 (A1-F18AC-744-300, WP009 00)....	-	-
g. ON 164725 AND UP; ALSO 164627 THRU 164724 AFTER F/A-18 AFC 126, Data Transfer Interface Unit J-6008/A (DFIRS) indicates wrong program load CONFIG/ IDENT number do substeps below:		
(1) Reload Data Transfer Interface Unit J-6008/A (DFIRS) (WP006 00)	-	-
(2) If load status still wrong replace Data Transfer Interface Unit J-6008/A (A1-F18AE-580-300, WP014 00)	-	-
h. Digital Computer-Converter CP-1805/AAR50 indicates wrong program load CONFIG/ IDENT number do substep below:		
(1) Replace Digital Computer-Converter CP-1805/AAR50 (A1-F18AG-746-300, WP004 00)	-	-
i. ON 163985 AND UP, Digital Display Indicator IP-1553 indicates wrong program load CONFIG/IDENT number do substep below:		
(1) Replace Digital Display Indicator IP-1553 (A1-F18AG-745-300, WP004 00).....	-	-
j. Digital Map Computer CP-1802/ASQ-196 indicates wrong program load CONFIG/IDENT number do substeps below:		
(1) WITH DIGITAL DATA COMPUTER CONFIG/IDENT 91C AND UP, do substeps below:		
(a) Reload Digital Map Computer CP-1802/ASQ-196 (WP006 00)	-	-
(b) If load status still wrong replace Digital Map Computer CP-1802/ASQ-196 (A1-F18AG-731-300, WP003 00)	-	-
(2) replace Digital Map Computer CP-1802/ASQ-196 (A1-F18AG-731-300, WP003 00)...	-	-
k. Inertial Navigation Unit CN-1561/ASN-130A or Inertial Navigation Unit CN-1649/ASN-139 indicates wrong program load CONFIG/IDENT number do substep below:		
(1) Replace Inertial Navigation Unit CN-1561/ASN-130A or Inertial Navigation Unit CN-1649/ASN-139 (A1-F18AC-730-300, WP004 00)	-	-
l. Interconnecting Box J-3656/ASQ-173 indicates wrong program load CONFIG/IDENT number do substep below:		
(1) Replace Interconnecting Box J-3656/ASQ-173 (A1-F18AC-743-300, WP004 00)	-	-
m. Memory Unit MU-806/ASQ-194 indicates wrong program load CONFIG/IDENT number do substep below:		
(1) Replace Memory Unit MU-806/ASQ-194 (A1-F18AE-580-300, WP005 00)	-	-

Table 4. Program Load CONFIG/IDENT Wrong (Continued)

Procedure	No	Yes
n. ON 164898 AND UP, Radar Data Processor CP-2062/APG-73 indicates wrong program load CONFIG/IDENT number do substeps below:		
(1) Reload Radar Data Processor CP-2062/APG-73 (WP006 00)	-	-
(2) Replace Radar Data Processor CP-2062/APG-73 (A1-F18AH-742-300, WP004 00)	-	-
o. Roll-Pitch-Yaw Computer CP-1330/ASW-44 (FCCA) indicates wrong program load CONFIG/IDENT number do substep below:		
(1) Replace Roll-Pitch-Yaw Computer CP-1330/ASW-44 (FCCA) (A1-F18AC-570-300, WP003 00)	-	-
p. Roll-Pitch-Yaw Computer CP-1330/ASW-44 (FCCB) indicates wrong program load CONFIG/IDENT number do substep below:		
(1) Replace Roll-Pitch-Yaw Computer CP-1330/ASW-44 (FCCB) (A1-F18AC-570-300, WP003 00)	-	-
q. ON F/A-18C/D, Signal Data Computer CP-1726/ASQ-194 indicates wrong program load CONFIG/IDENT number do substeps below:		
(1) Reload Signal Data Computer CP-1726/ASQ-194 (WP006 00)	-	-
(2) If load status still wrong, replace Signal Data Computer CP-1726/ASQ-194 (A1-F18AE-580-300, WP003 00)	-	-

Table 5. MC CONFIG Caution and No USN Customer Identifier

Support Equipment Required

NOTE

Alternate item type designations or part numbers are listed in parentheses.

Part Number or Type Designation	Nomenclature
77BN (77AN)	Multimeter

Materials Required

None

NOTE

Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (A1-F18AC-741-500, WP013 00 or A1-F18AE-741-500, WP013 00) may be used as an aid when doing this procedure.

For component locator, refer to WP005 00.

Malfunction is caused by one of the items below:

Aircraft Wiring
Digital Data Computer No. 1

Procedure	No	Yes
NOTE The question used in logic tree "Does continuity exist" means to test for the items listed below: 1. Pin to pin test per procedure step. 2. Shorts to ground. 3. Shorts between surrounding pins on connectors. 4. Shorts between shield and conductors. 5. Shield continuity.		
a. Do substeps below:		
(1) Open door 13L (A1-F18AC-LMM-010).		
(2) Disconnect 83P-E001C from digital data computer no. 1.		
(3) Does continuity exist from:		

Table 5. MC CONFIG Caution and No USN Customer Identifier (Continued)

Procedure	No	Yes
83P-E001C pin 46 to aircraft ground 83P-E001C pin 47 to aircraft ground 83P-E001C pin 48 to aircraft ground 83P-E001C pin 49 to aircraft ground 83P-E001C pin 50 to aircraft ground 83P-E001C pin 51 to aircraft ground?	b	c
b. Replace Digital Data Computer No. 1 (A1-F18A( )-741-300, WP003 00) and do step d	-	-
c. Repair defective pin (A1-F18A( )-WRM-000) and do step d	-	-
d. If disconnected, removed, or opened during this procedure, make sure the items listed below are connected, installed, or closed:		
(1) 83P-E001C		
(2) Door 13L	-	-

Table 6. MC CONFIG Caution and USN Customer Identifier Displayed

Support Equipment Required

NOTE

Alternate item type designations or part numbers are listed in parentheses.

Part Number or Type Designation	Nomenclature
77BN (77AN)	Multimeter

Materials Required

None

NOTE

Digital Data Computer No. 1 and No. 2 Cautions, Advisory and Maintenance Codes Schematic (A1-F18AC-741-500, WP013 00 or A1-F18AE-741-500, WP013 00) may be used as an aid when doing this procedure.

For component locator, refer to WP005 00.

For memory inspect displays, see figure 3, this WP.

Malfunction is caused by one of the items below:

Aircraft Wiring
Digital Data Computer No. 1
Digital Data Computer No. 2

Procedure	No	Yes

NOTE

The question used in logic tree "Does continuity exist" means to test for the items listed below:

1. Pin to pin test per procedure step.
2. Shorts to ground.
3. Shorts between surrounding pins on connectors.
4. Shorts between shield and conductors.
5. Shield continuity.

a. Do substeps below:

(1) Open door 14R (A1-F18AC-LMM-010).

Table 6. MC CONFIG Caution and USN Customer Identifier Displayed (Continued)

Procedure	No	Yes
(2) Disconnect 83P-F002C from digital data computer no. 2.		
(3) Open door 13L (A1-F18AC-LMM-010).		
(4) Disconnect 83P-E001C from digital data computer no. 1.		
(5) Does continuity exist from: 83P-F002C pin 46 to aircraft ground 83P-F002C pin 47 to aircraft ground 83P-F002C pin 48 to aircraft ground 83P-F002C pin 49 to aircraft ground 83P-F002C pin 50 to aircraft ground 83P-F002C pin 51 to aircraft ground 83P-E001C pin 90 to aircraft ground 83P-E001C pin 91 to aircraft ground?	c	b
b. Repair defective pin (A1-F18A( )-WRM-000) and do step h	-	-
c. Does continuity exist from 83P-F002C pin 90 to 91?	d	e
d. Repair defective wiring from 83P-F002C pin 90 to 91 (A1-F18A( )-WRM-000) and do step h	-	-
e. Do substeps below:		
(1) Connect 83P-F002C to digital data computer no. 2 (Door 14R).		
(2) Connect 83P-E001C to digital data computer no. 1 (Door 13L).		
(3) Apply electrical power (A1-F18AC-LMM-000).		
(4) On GND PWR control panel assembly, set and hold 1 switch to A ON and 2 switch to B ON for 3 seconds.		
(5) On left and right Digital Display Indicator (LDDI and RDDI), set power switch to DAY or NIGHT and allow 2 minute warmup. Adjust BRT and CONT controls for best display.		
(6) On RDDI:		
(a) Press and release MENU pushbutton switch until BIT pushbutton appears.		
(b) Press BIT pushbutton switch.		
(c) Press MI pushbutton switch.		
(7) On Electronic Equipment Control C-10380/ASQ, adjust BRT control for best display then do substeps below:		

Table 6. MC CONFIG Caution and USN Customer Identifier Displayed (Continued)

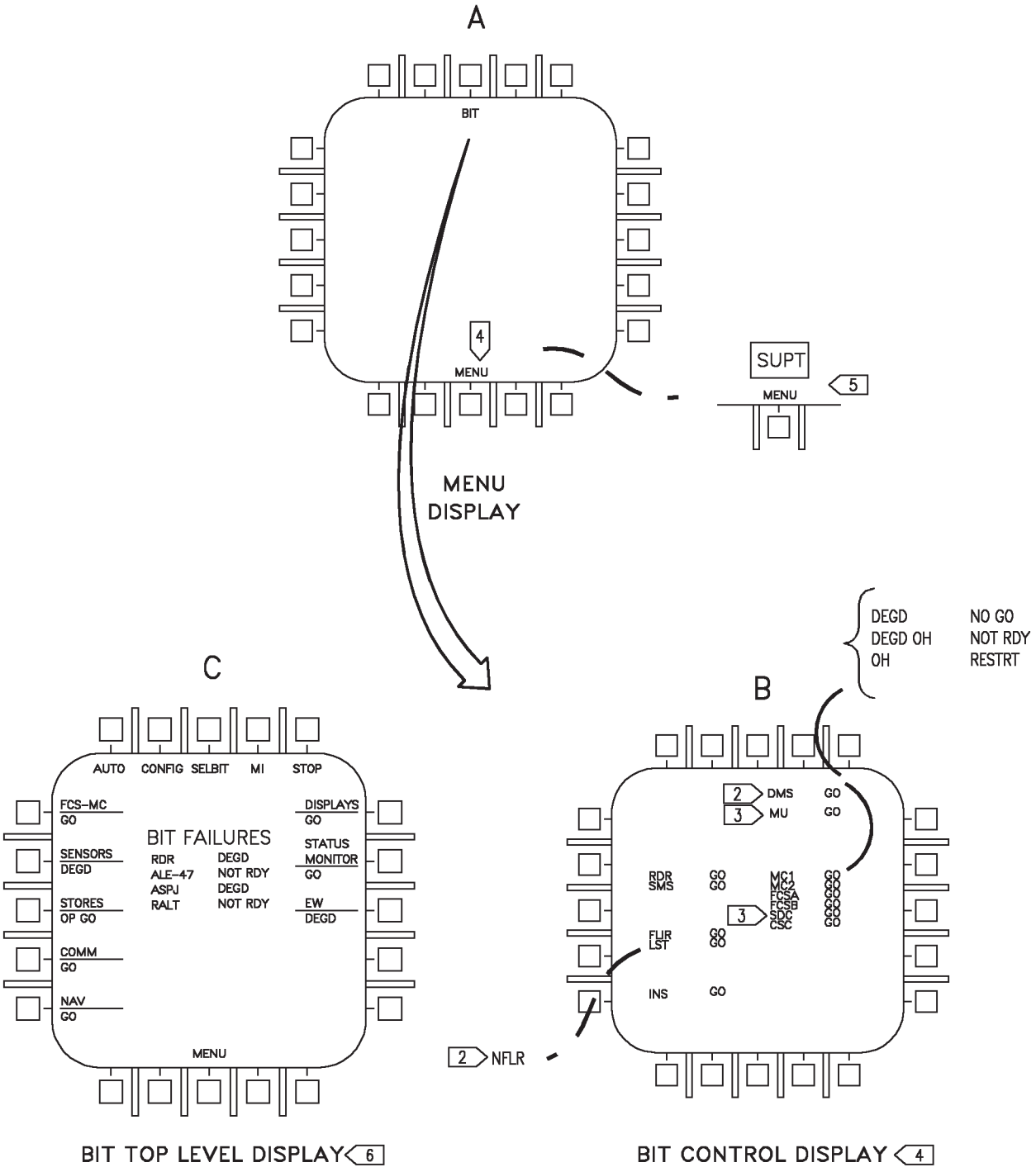
Procedure	No	Yes
(a) Press UNIT select switch.		
(b) Press keyboard switches 2 and 8. Verify 28 is displayed on scratch pad display and press ENT.		
(c) On RDDI, verify unit address is 28.		
(d) Press ADDR select switch.		
NOTE WITH DIGITAL DATA COMPUTER CONFIG/IDENT 09C AND UP, ref code addressing is eight octal digits. When ref code address is less than eight digits, a 0 (zero) is displayed before the address. Example - address 7004444 is displayed as 07004444. WITH DIGITAL DATA COMPUTER CONFIG/IDENT 89A AND UP; OR 91C, ref code addressing is six octal digits. When ref code address is less than six digits, a 0 (zero) is displayed before the address. Example - address 4444 is displayed as 004444.		
(e) Enter ref code I91S15 by pressing keyboard switches to enter ref code MI address (Table 7, this WP).		
(f) Verify that ref code MI address is displayed on scratch pad display and then press ENT.		
(g) On RDDI, verify ADDR readout is same as address entered.		
NOTE On RDDI DATA readout is six octal digits. When an X is indicated in an octal digit location in this procedure, that digit is ignored.		
(8) On RDDI, does DATA readout display 0XXXXX?.....	f	g
f. Replace Digital Data Computer No. 1 (A1-F18A( )-741-300, WP003 00) and do step h.....	-	-
g. Replace Digital Data Computer No. 2 (A1-F18A( )-741-300, WP004 00) and do step h.....	-	-
h. If disconnected, removed, or opened during this procedure, make sure the items listed below are connected, installed, or closed:		
(1) 83P-E001C		
(2) 83P-F002C		
(3) Door 14R		

Table 6. MC CONFIG Caution and USN Customer Identifier Displayed (Continued)

Procedure	No	Yes
(4) Door 13L.....	-	-

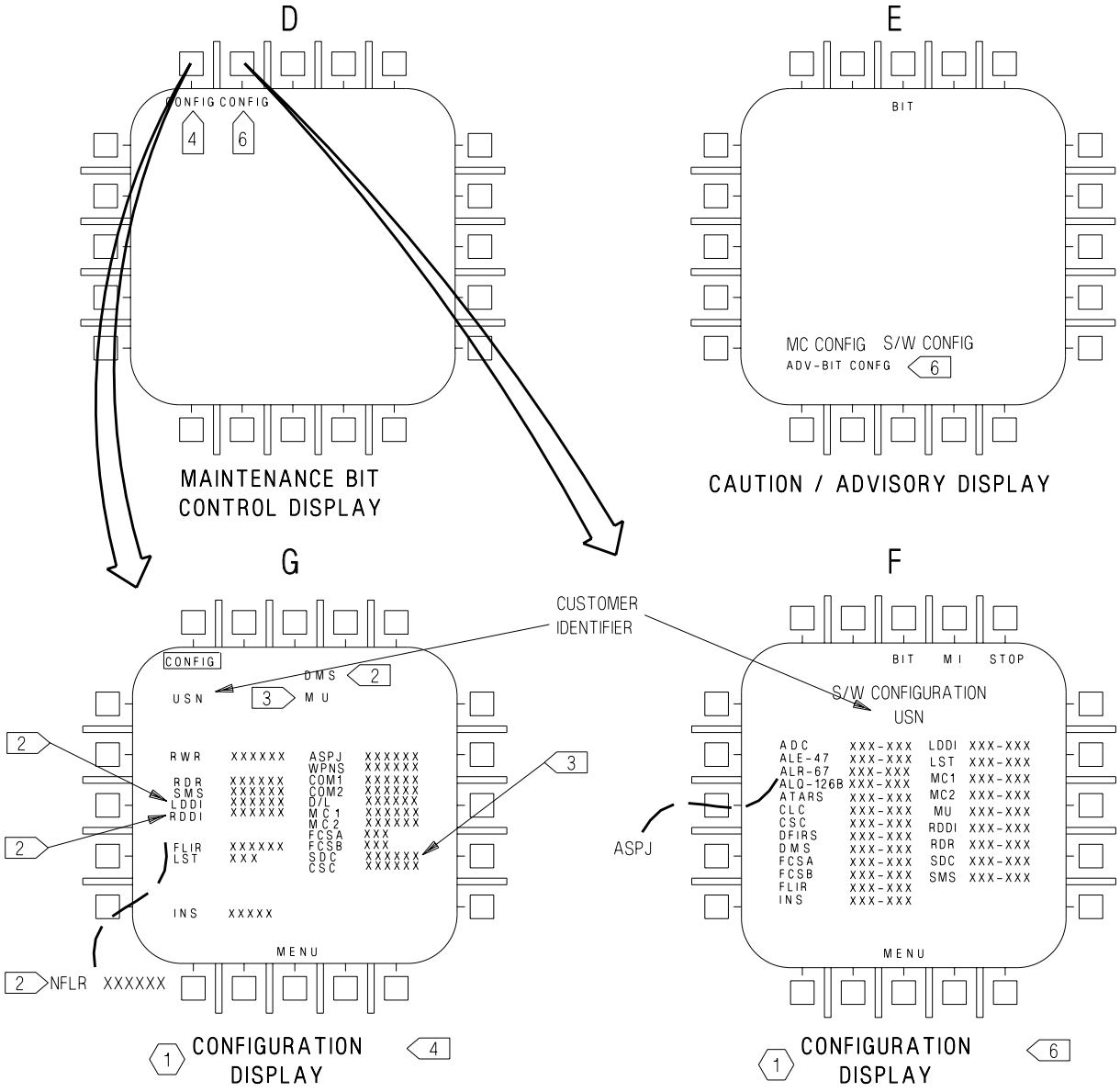
Table 7. Unit Address 28 (MC1) MI Addresses

REF Code	SOFTWARE CONFIGURATION (CONFIG/IDENT)					
	10A+ Address	12A Address	15C Address	17C Address		
I91S15	042617	052040	00021412	00021562		



18AC-SCM-00-(3-1)28-CATI

Figure 1. Test Displays (Sheet 1)



MAJOR CHANGE

ADA541-3-2-035

Figure 1. Test Displays (Sheet 2)

LEGEND

⟨1⟩ REFER TO WP003 00 FOR DISPLAYED PROGRAM LOAD IDENTIFICATION NUMBERS (XXX).

2 163985 AND UP.

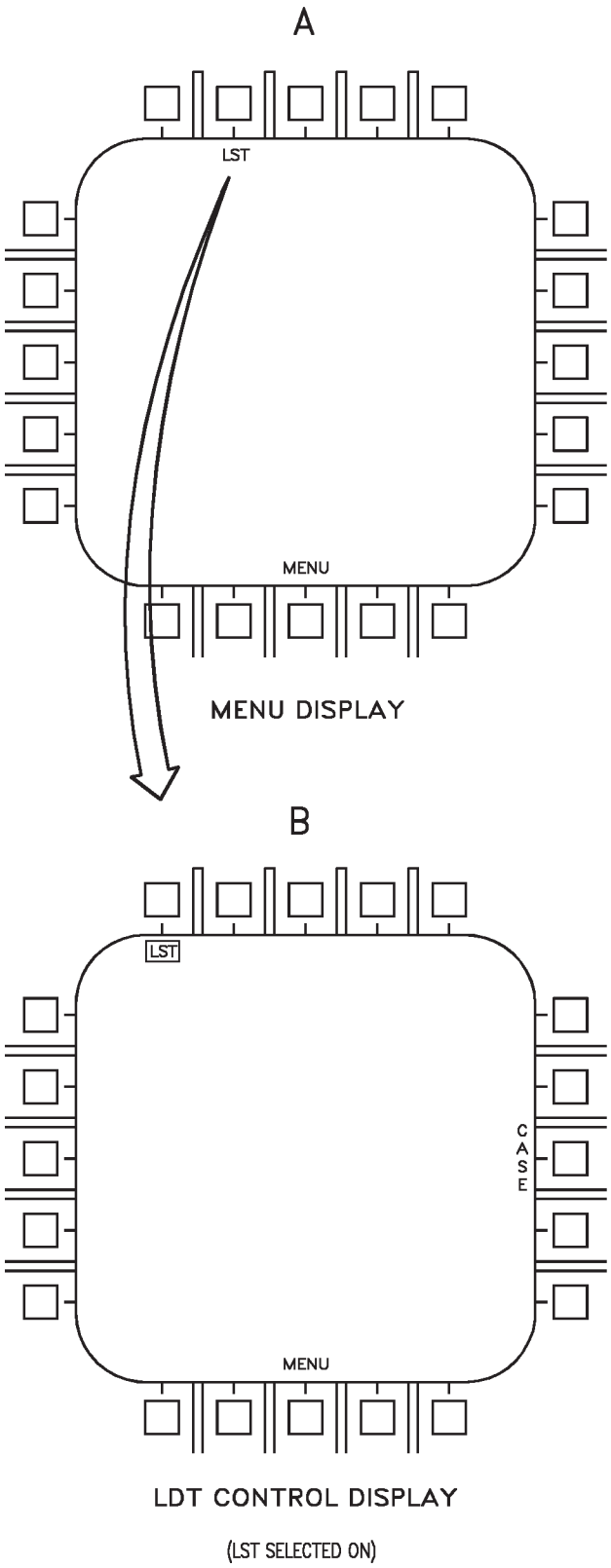
3 F/A-18C AND F/A-18D.

4 DIGITAL DATA COMPUTER CONFIG/IDENT 89A AND UP; OR 89C (A1-F18AC-SCM-000).

5 DIGITAL DATA COMPUTER CONFIG/IDENT 89C AND UP (A1-F18AC-SCM-000).

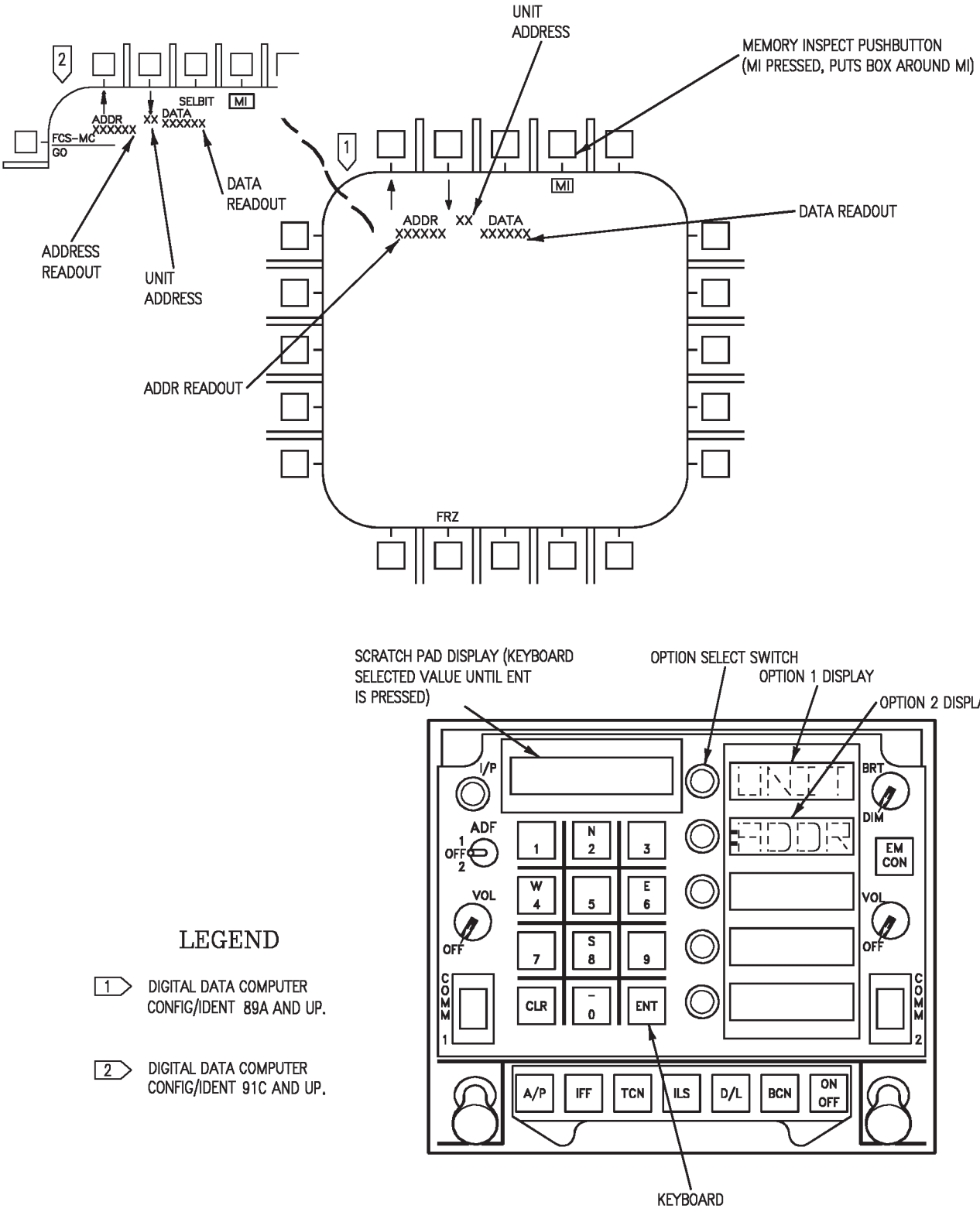
6 DIGITAL DATA COMPUTER CONFIG/IDENT 91C AND UP (A1-F18AC-SCM-000).

Figure 1. Test Displays (Sheet 3)



18AC-SCM-00-(5-1)28-CATI

Figure 2. LDT Turn On Displays



MEMORY INSPECT DISPLAY

18AC-SCM-00-(6-1)28-CATI

Figure 3. Memory Inspect Displays

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

COMPONENT LOCATOR

Reference Material

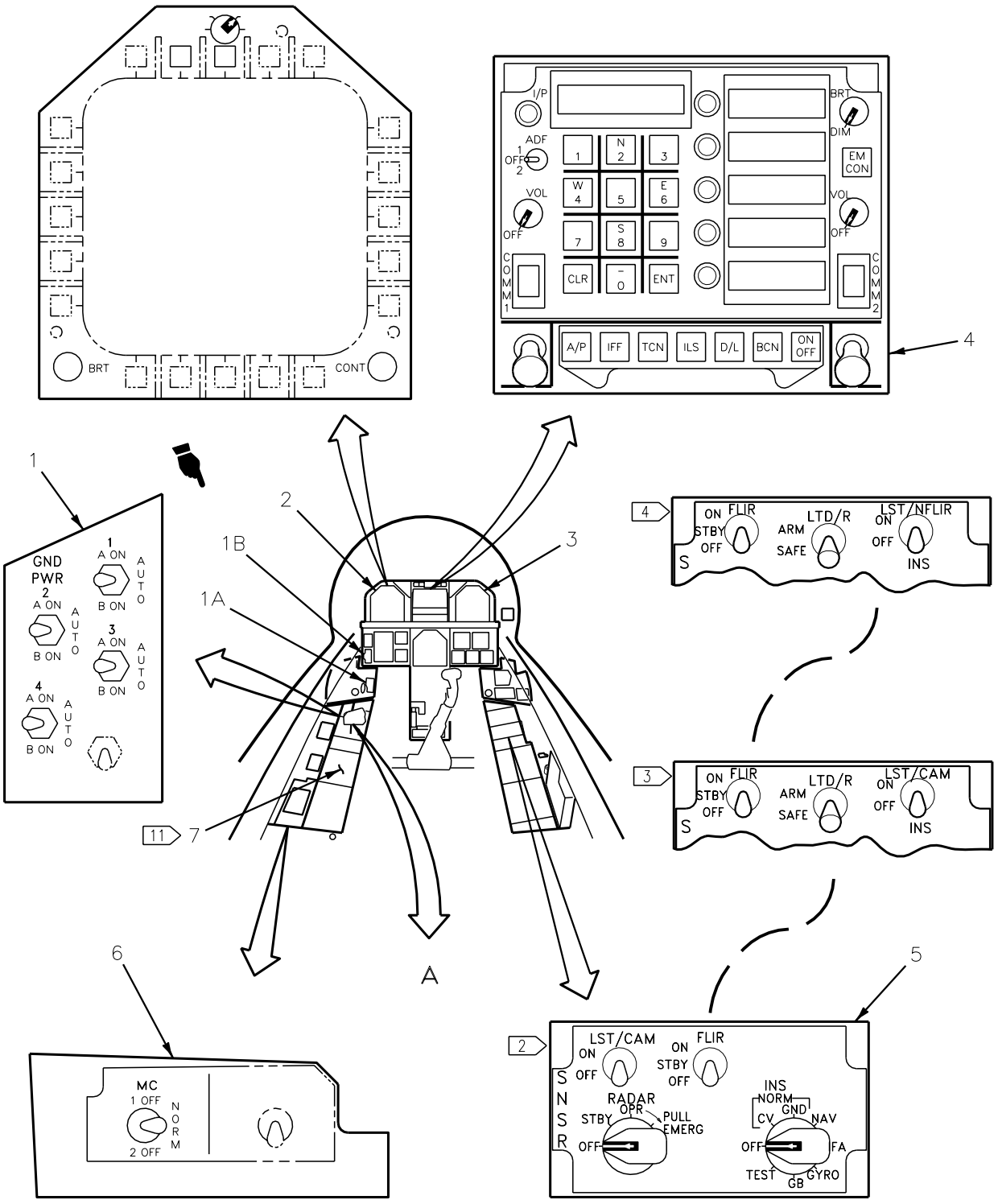
None

Alphabetical Index

Subject	Page No.
Component Locator, Figure 1	2

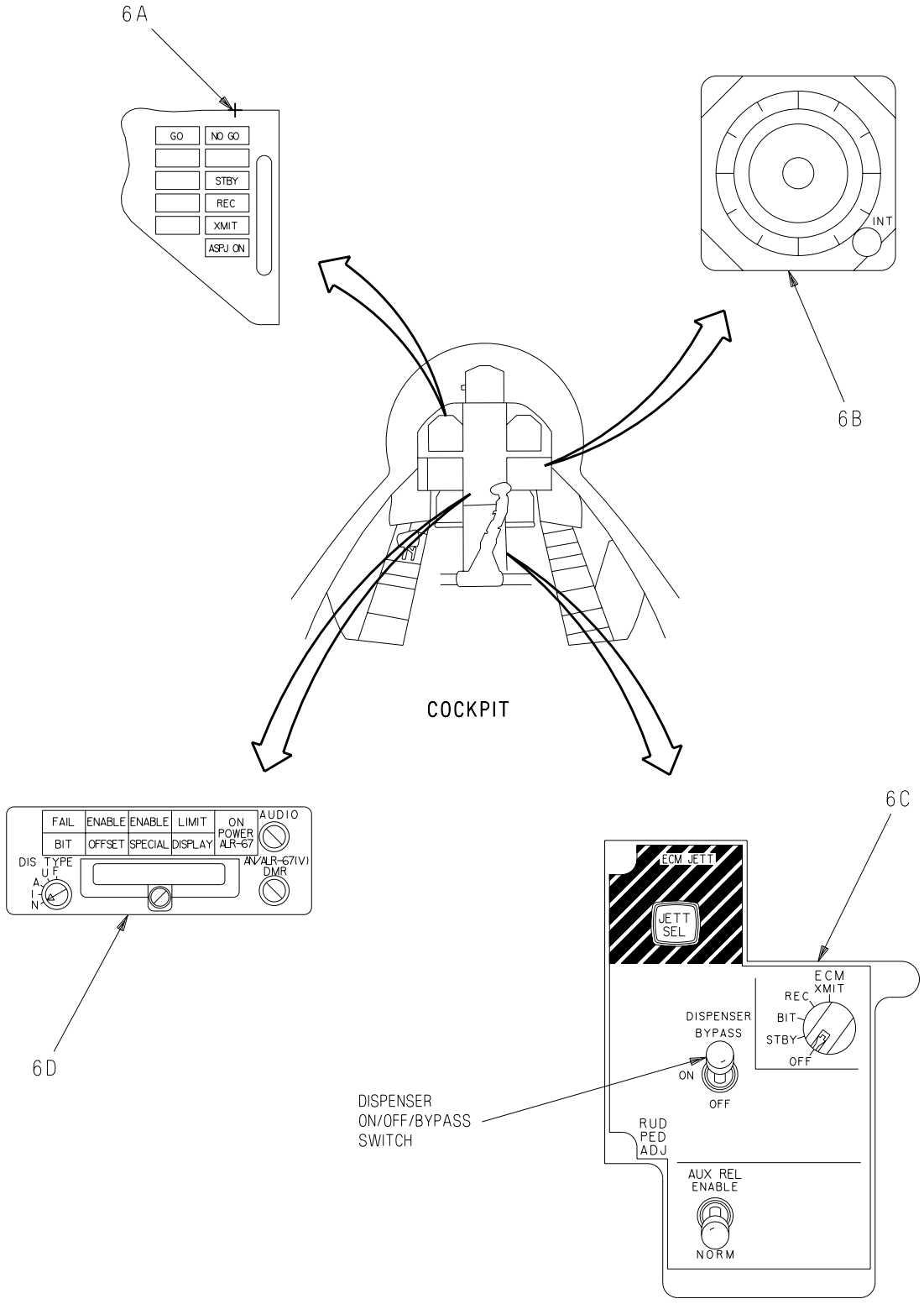
Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 126	-	Deployable Flight Incident Recorder Set (ECP MDA-F/A-18-00321R1C1)	15 Feb 93	-
F/A-18 AFC 175 PT 2	-	Miniaturized Airborne Global Positioning System (GPS) Receiver (MAGR), Incorporation of (ECP MDA-F/A-18-0405)	1 Jan 00	-
F/A-18 AFC 185	-	Have Quick/Sincgars, Incorp of (ECP MDA-F/A-18-00292R1A3 /R2)	15 Feb 94	-
F/A-18 AFC 211	-	AN/APG-65 Replacement With AN/APG-73 (ECP MDA-F/A-18-00508)	1 Sep 95	-
F/A-18 AFC 236	-	AN/APX-111(V) Combined Interrogator/ Transponder (CIT) Identification Friend or Foe (IFF) System, Retrofit of (ECP MDA-F/A-18-0520R1)	1 Jan 00	-
F/A-18 AFC 292	-	U.S. Marine Corps Reserves A+ Avionics Upgrade, Incorporation of (ECP MDA-F/A-18-0583)	15 Oct 00	-



18AC-SCM-00-(2-1)32-CATI

Figure 1. Component Locator (Sheet 1)



MAJOR CHANGE

ADA541-2-1-1-035

Figure 1. Component Locator (Sheet 1A)

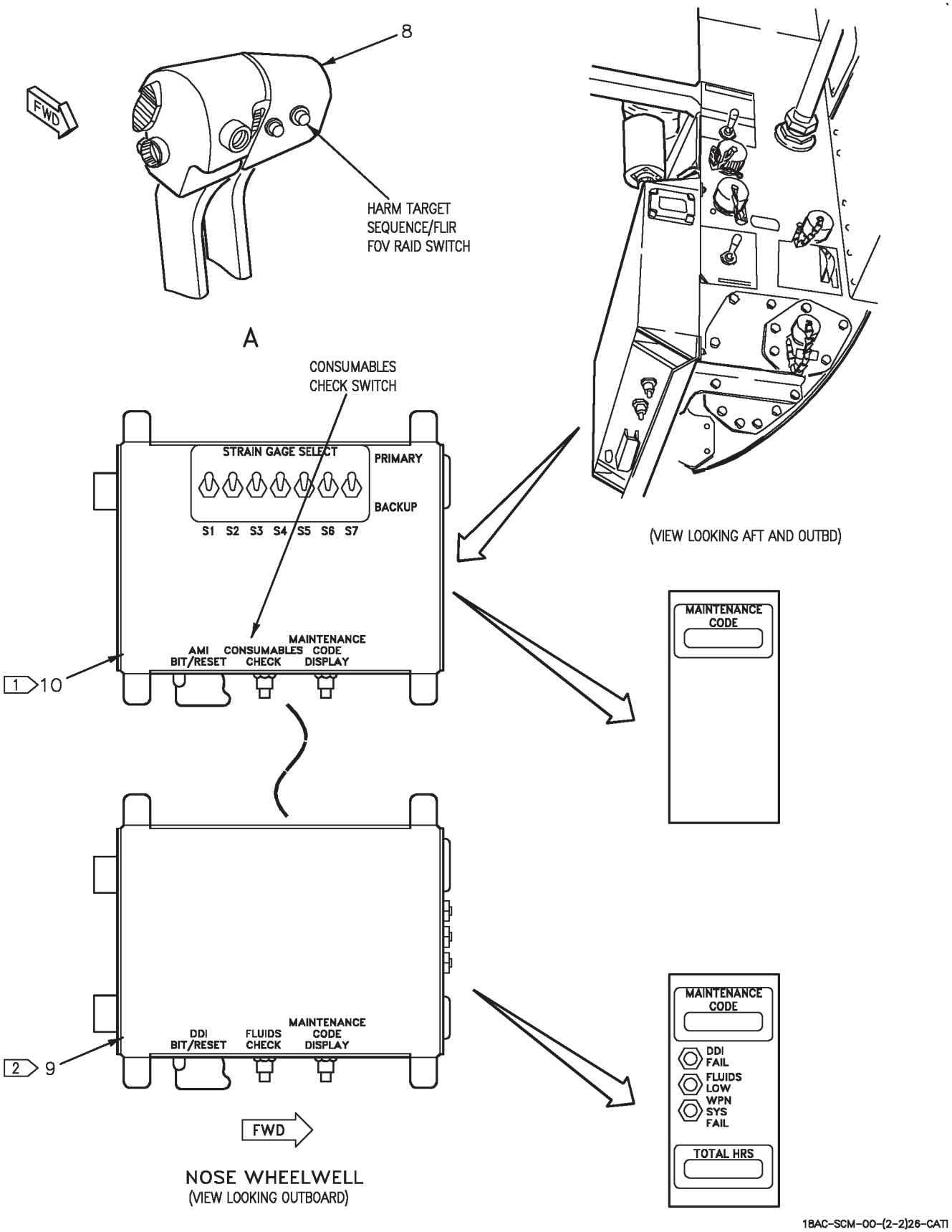
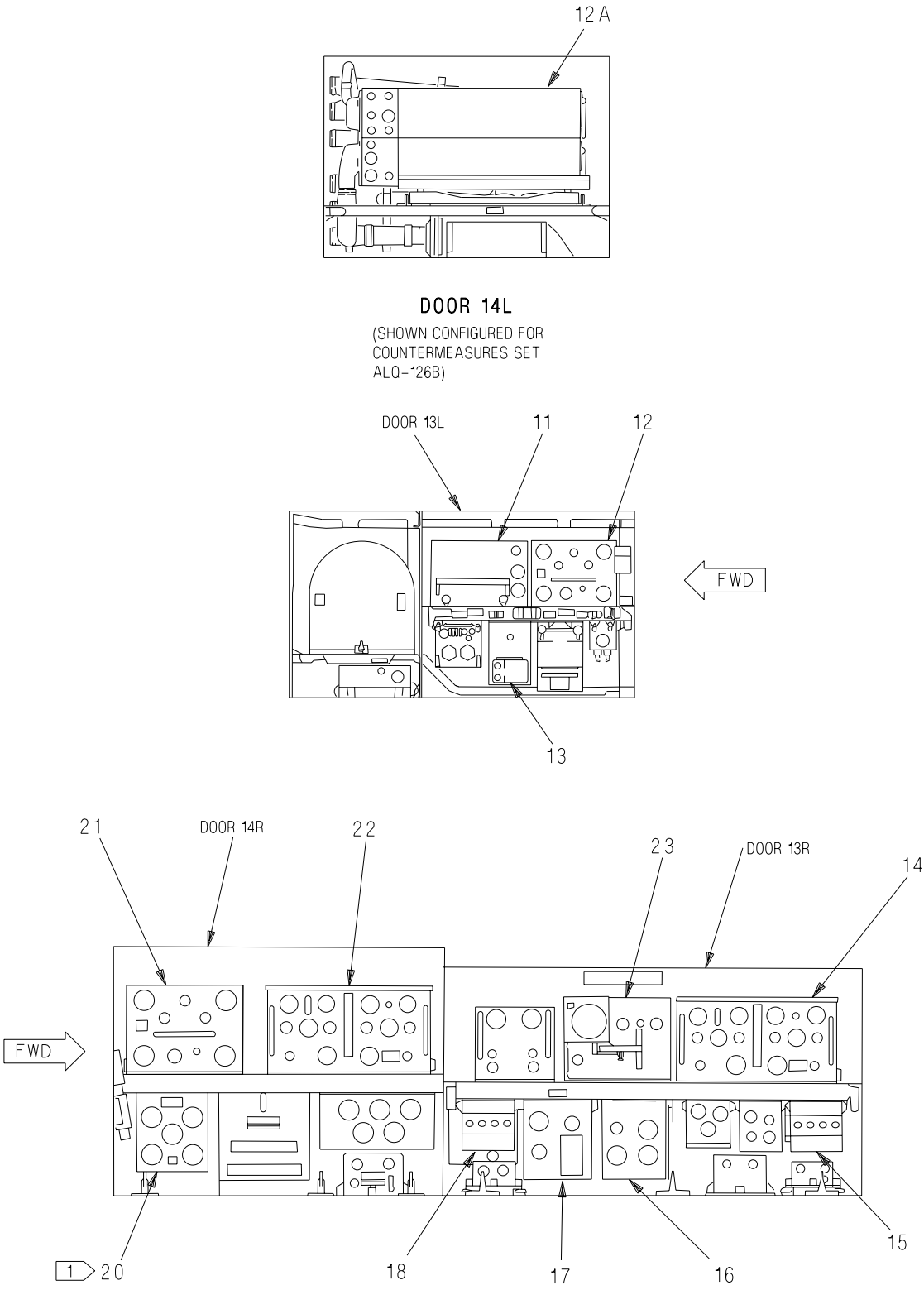


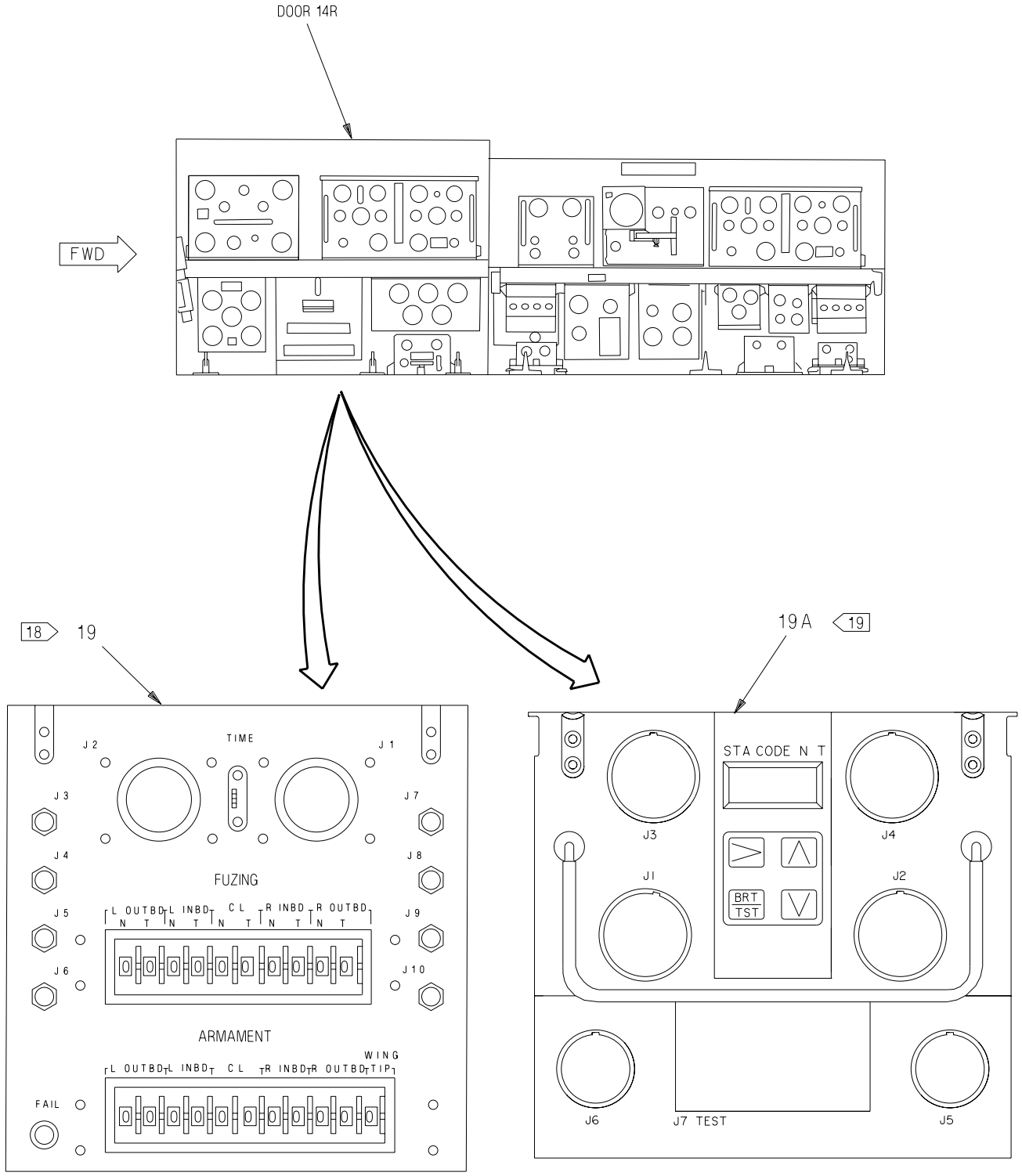
Figure 1. Component Locator (Sheet 2)



MAJOR CHANGE

ADA541-2-3-035

Figure 1. Component Locator (Sheet 3)



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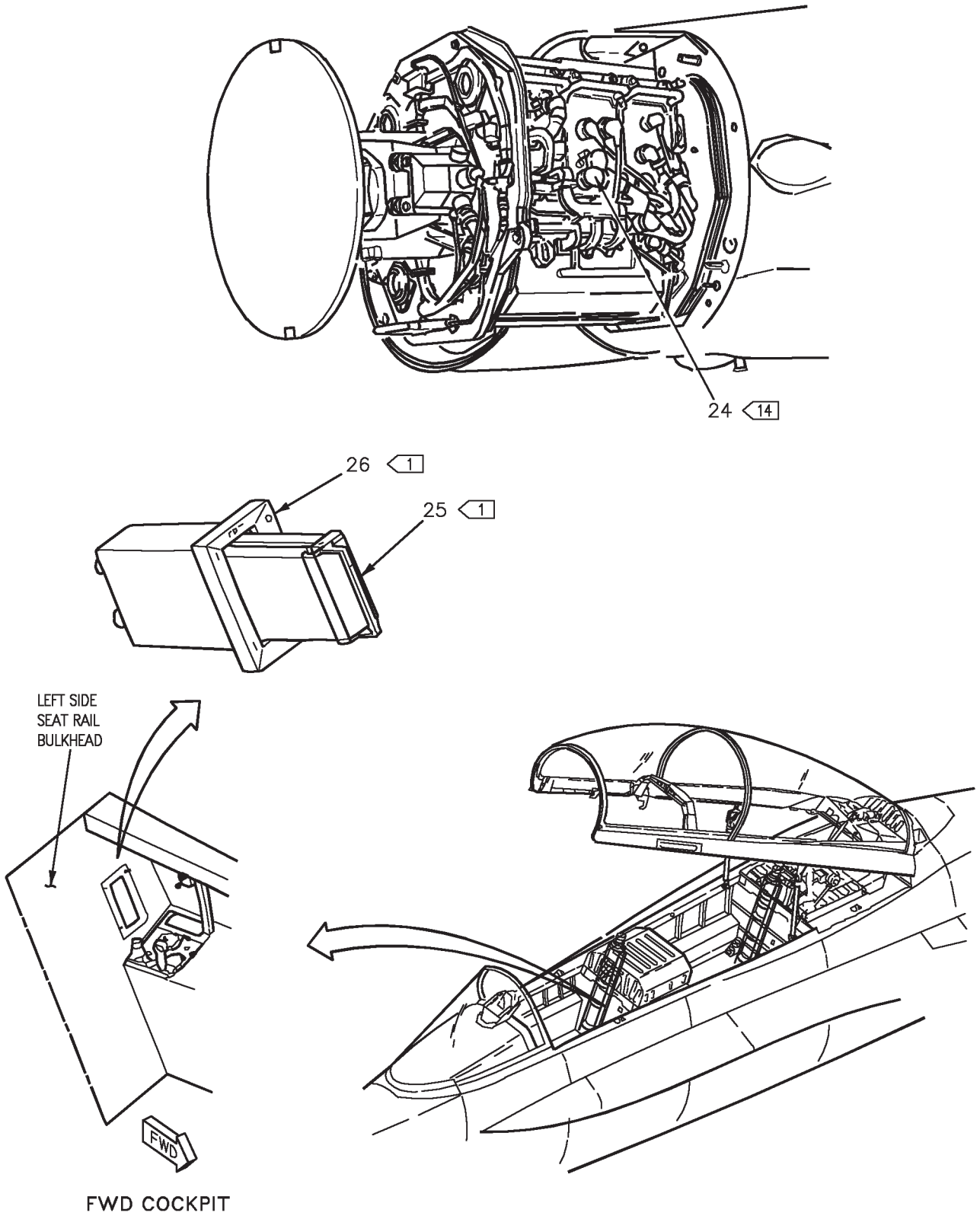
ADA541-2-3-1-035

Figure 1. Component Locator (Sheet 3A)



18AC-SCM-00-(2-4)28-CATI

Figure 1. Component Locator (Sheet 4)

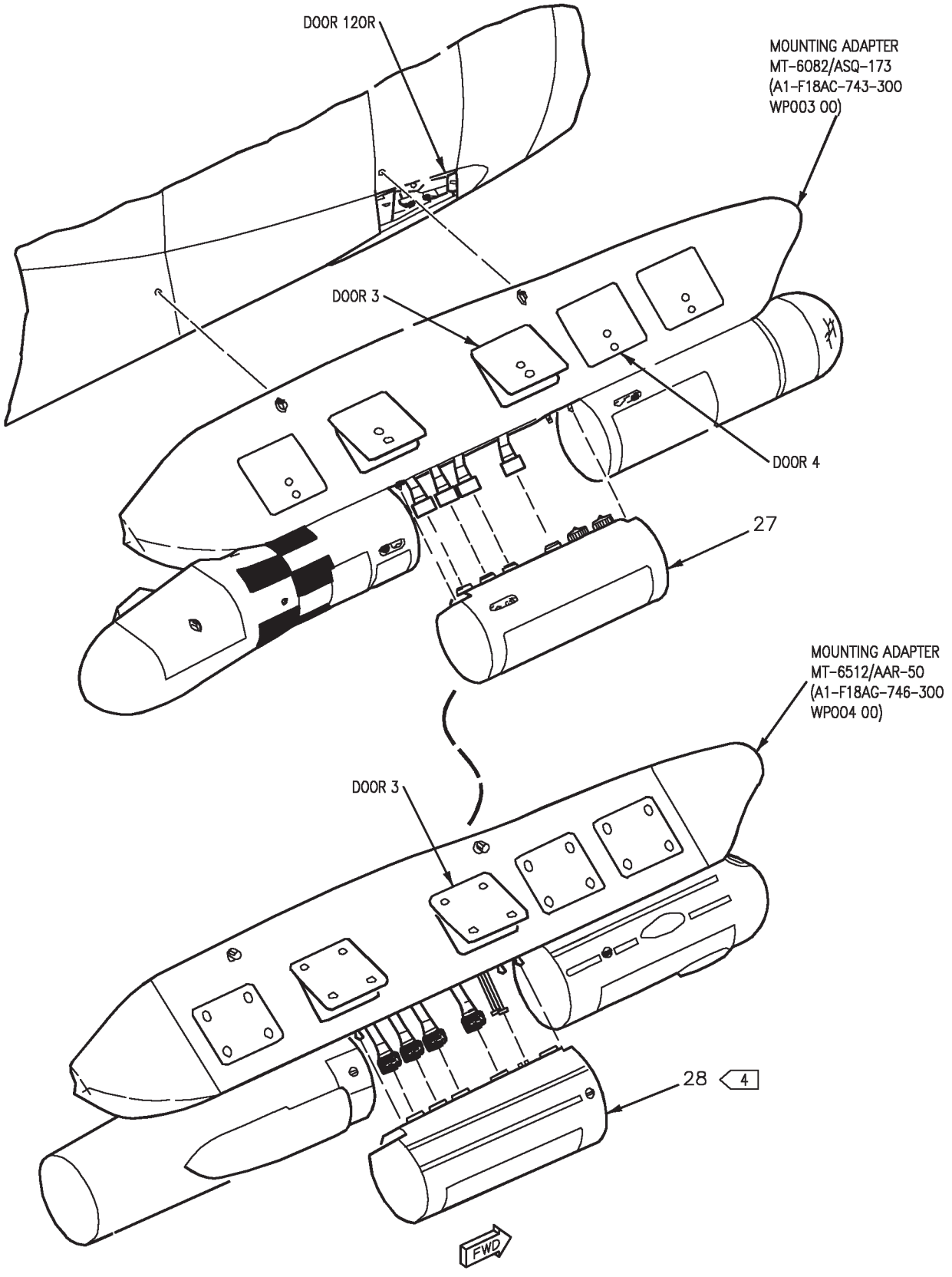
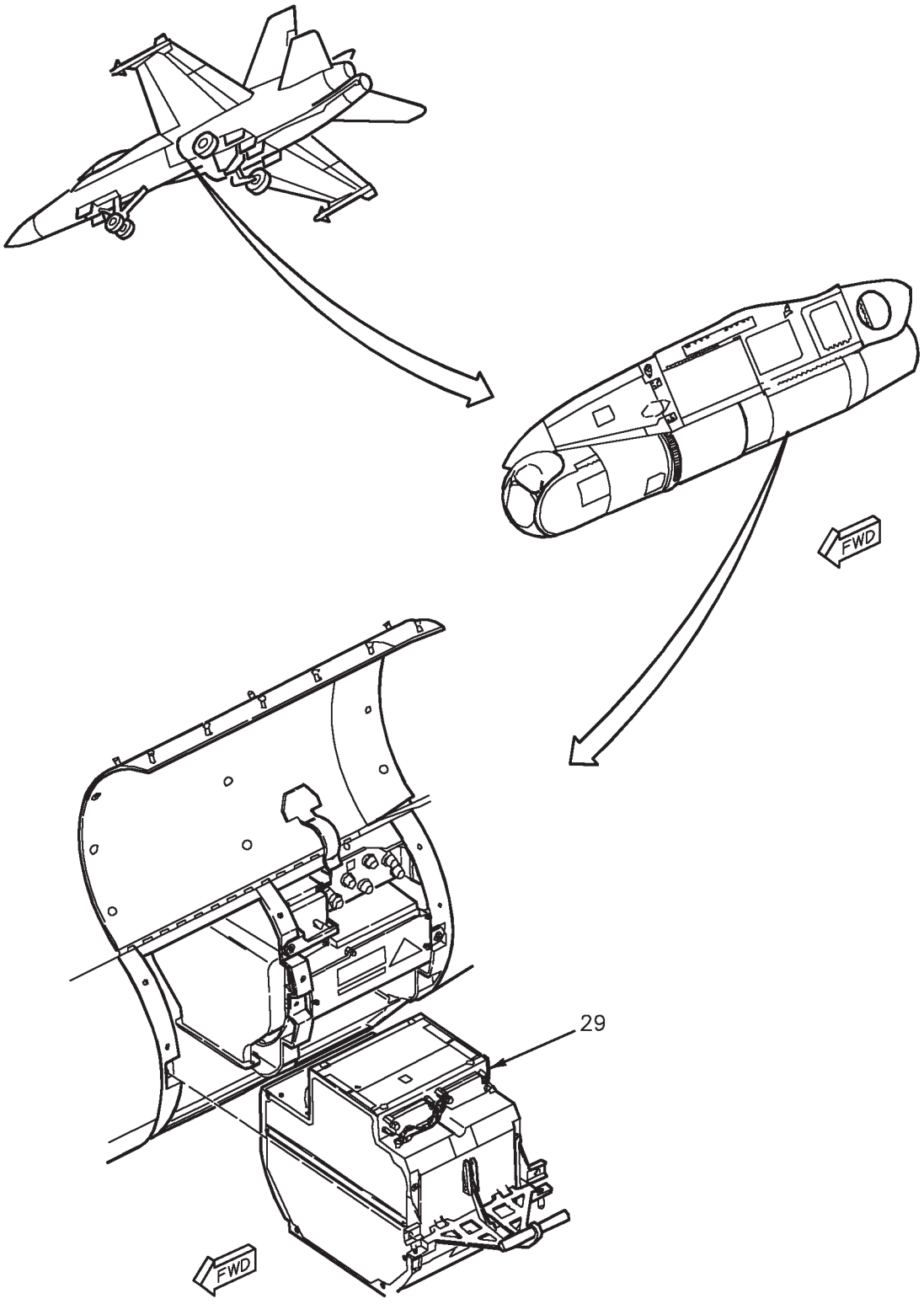
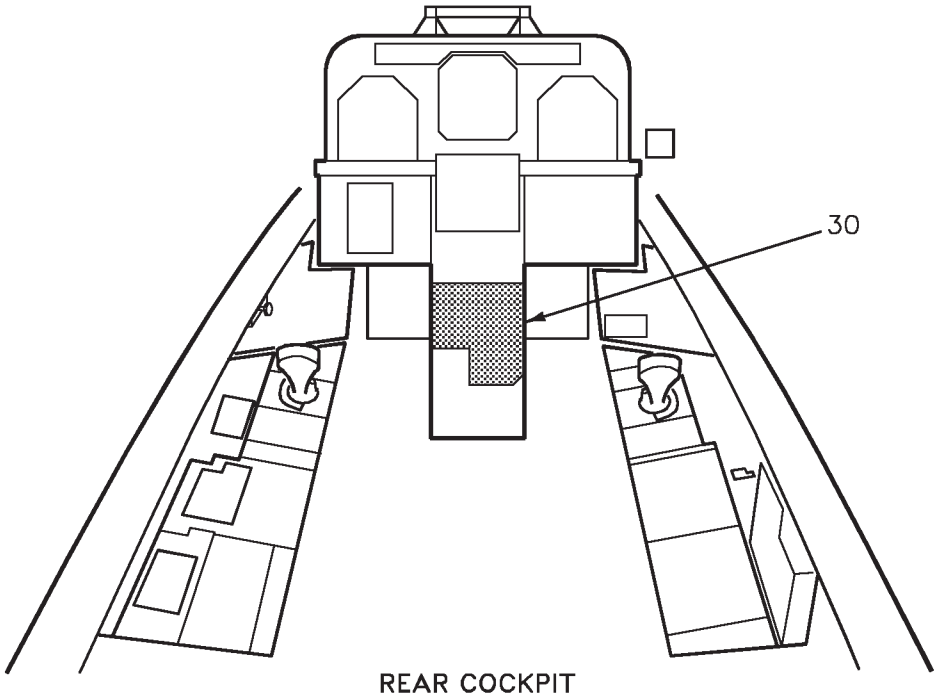


Figure 1. Component Locator (Sheet 5)

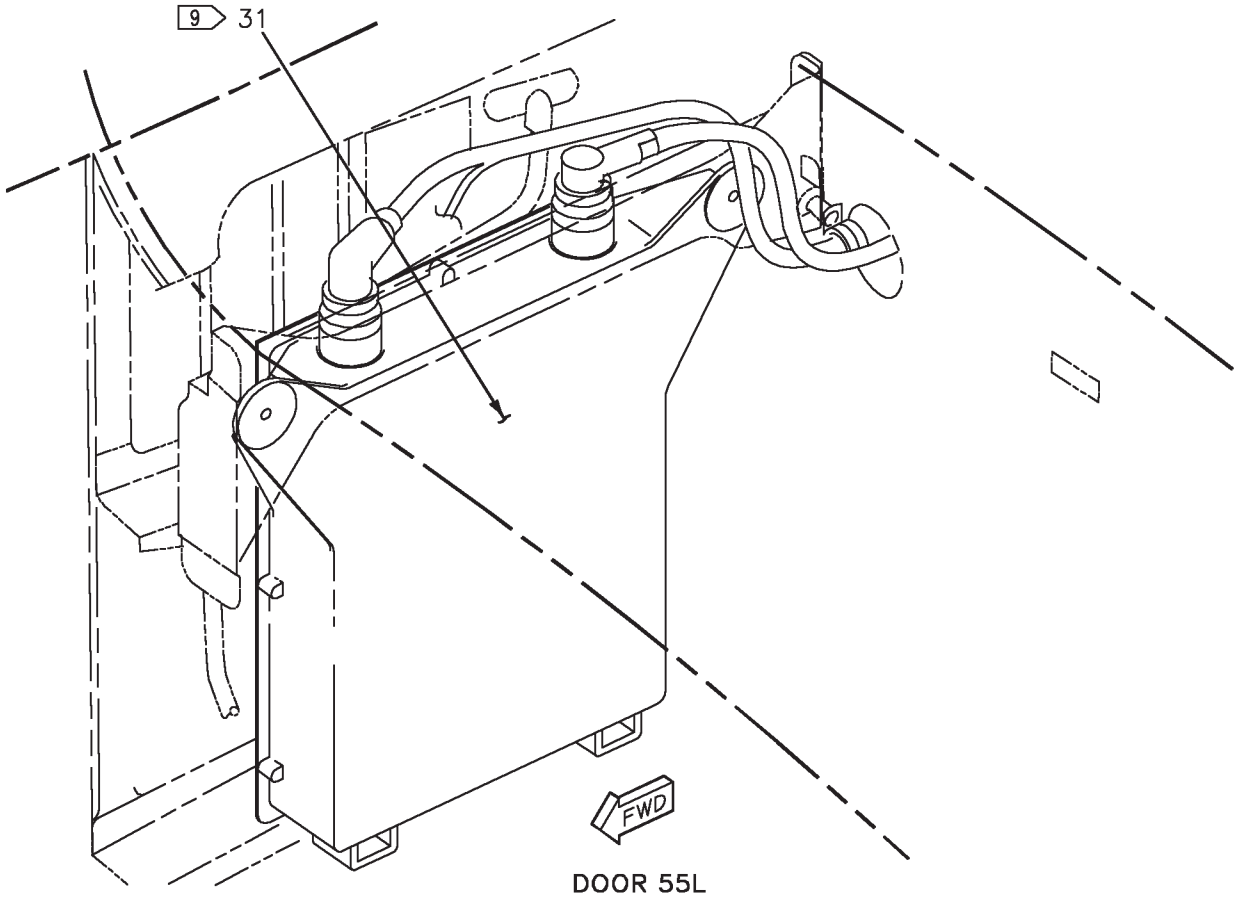


18AC-SCM-00-(2-6)26-CATI

Figure 1. Component Locator (Sheet 6)



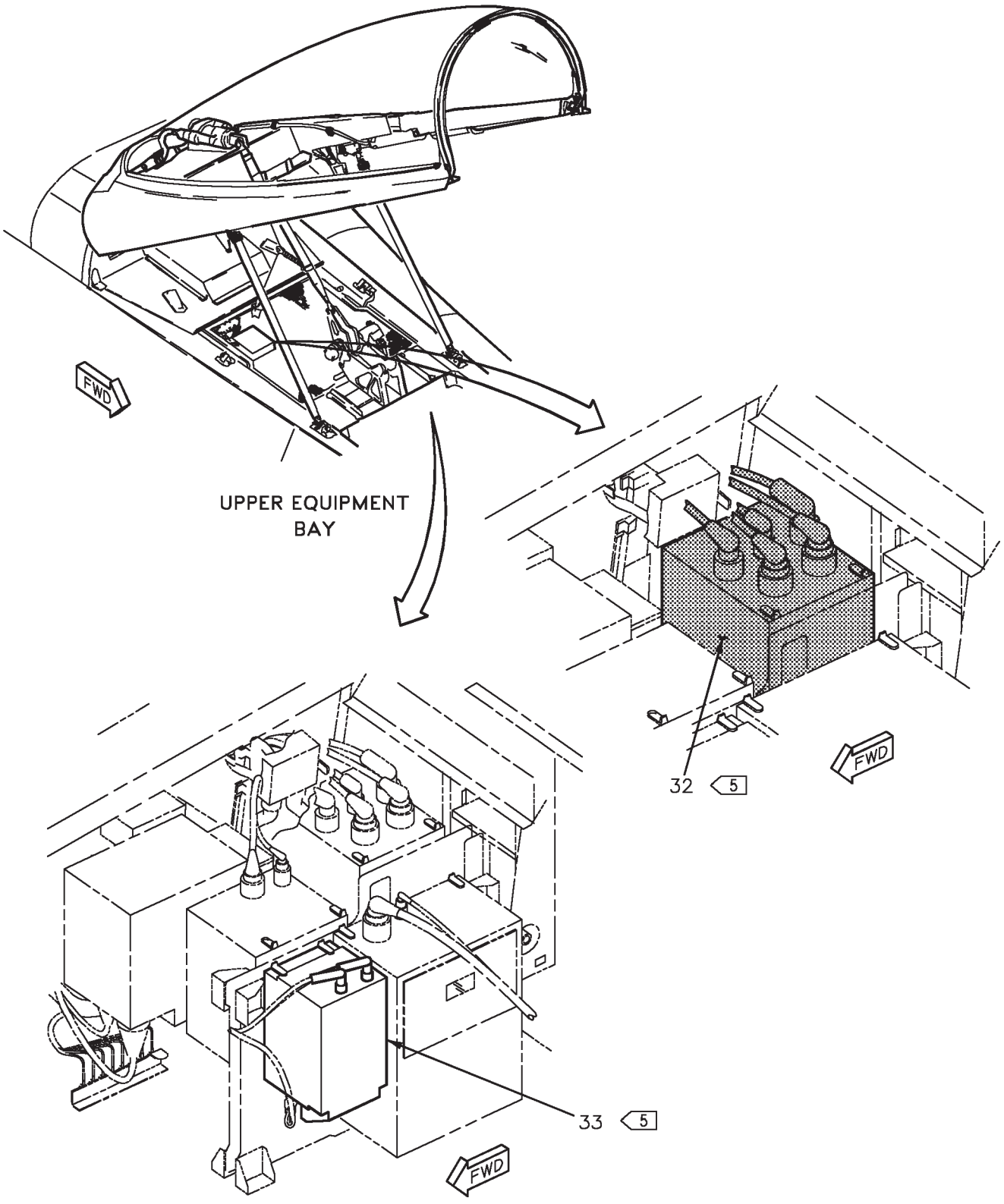
REAR COCKPIT
(163986 AND UP)



DOOR 55L

18AC-SCM-00-(2-7)26-CATI

Figure 1. Component Locator (Sheet 7)



18AC-SCM-00-(2-8)26-CATI

Figure 1. Component Locator (Sheet 8)

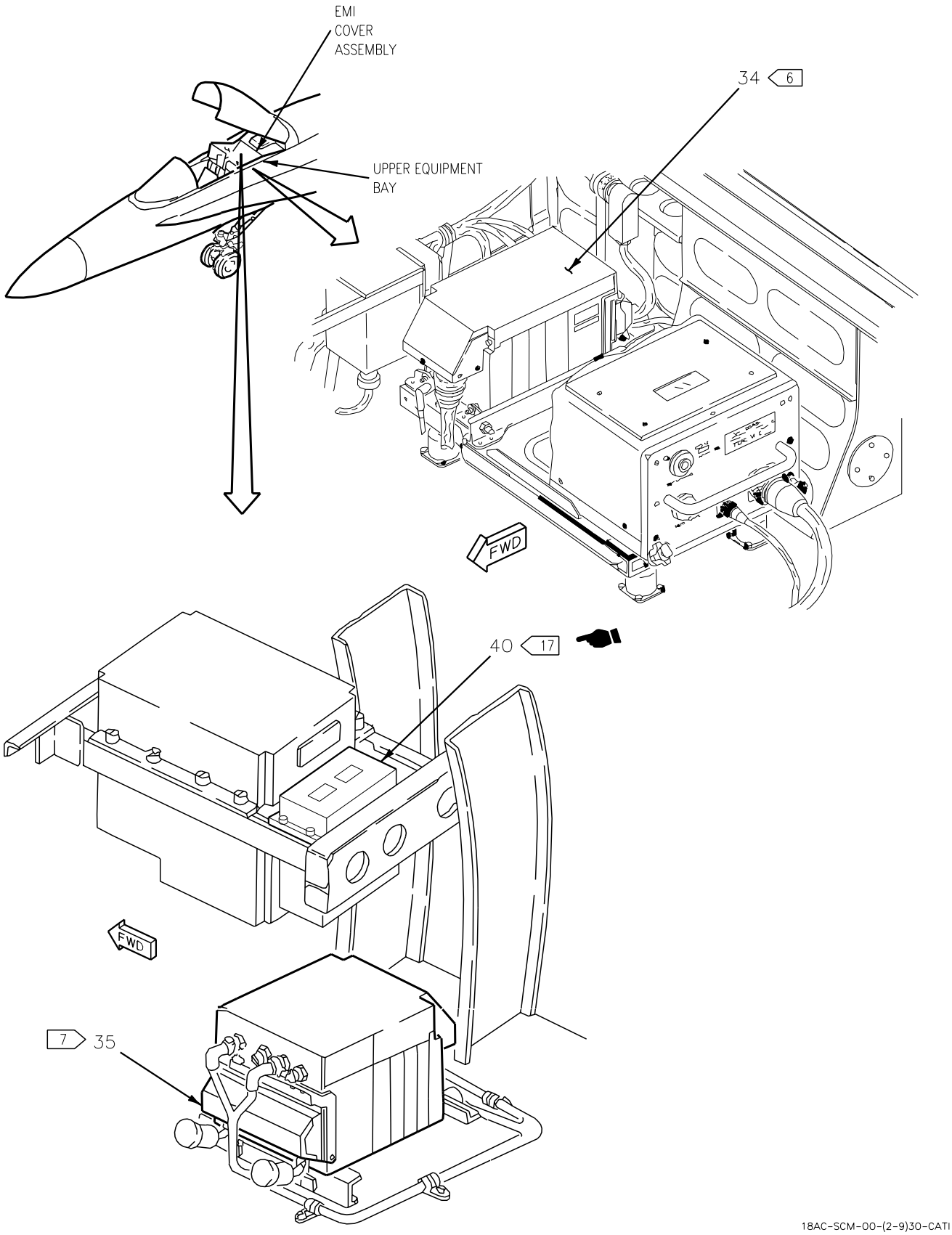
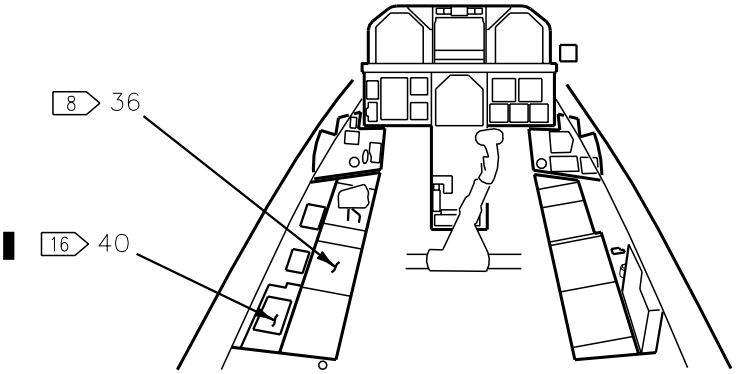
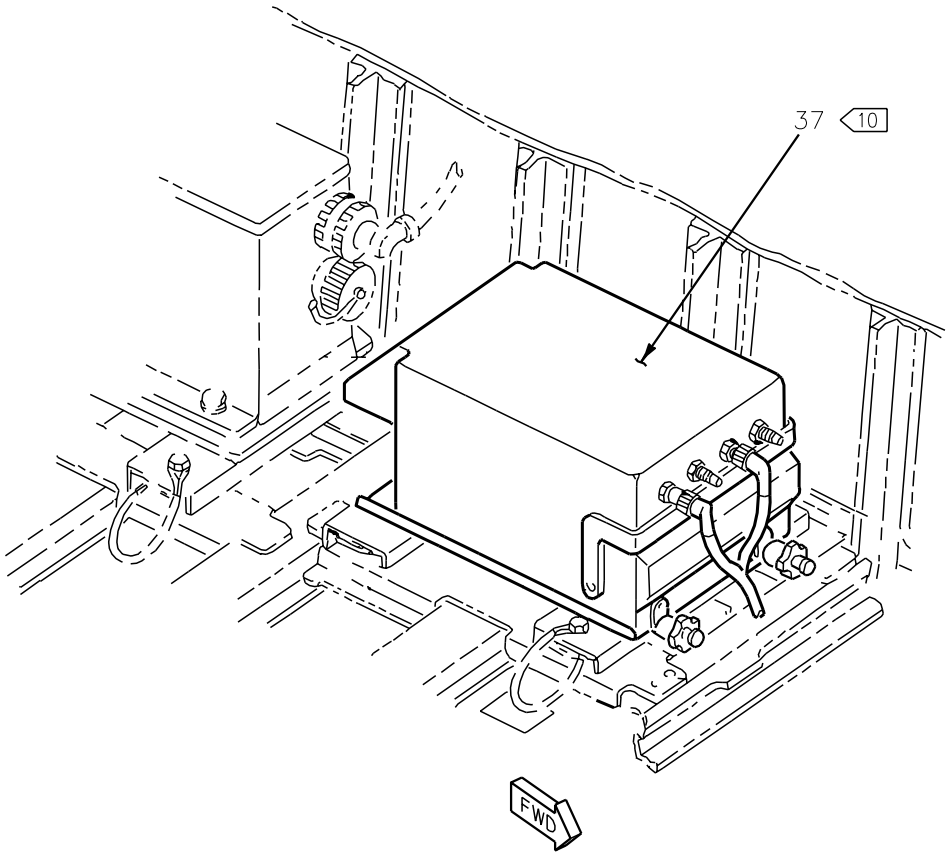


Figure 1. Component Locator (Sheet 9)



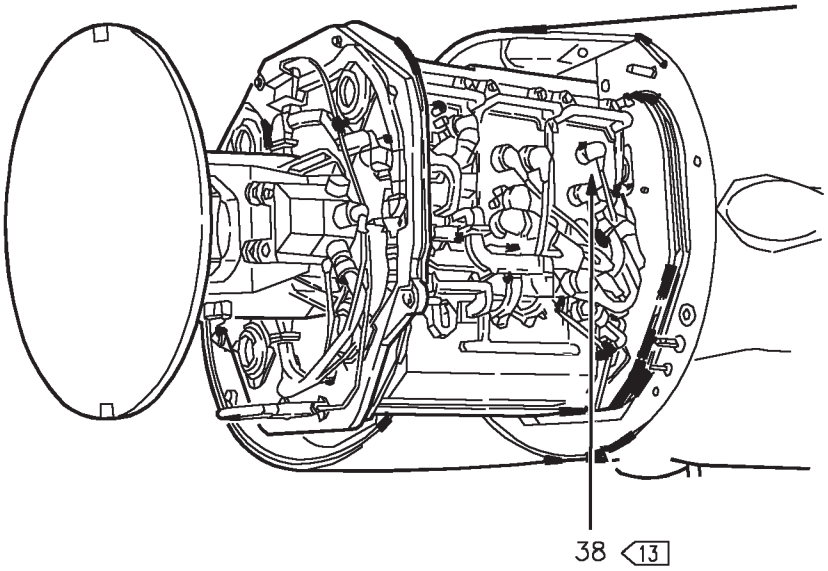
REAR COCKPIT



DOOR 123R

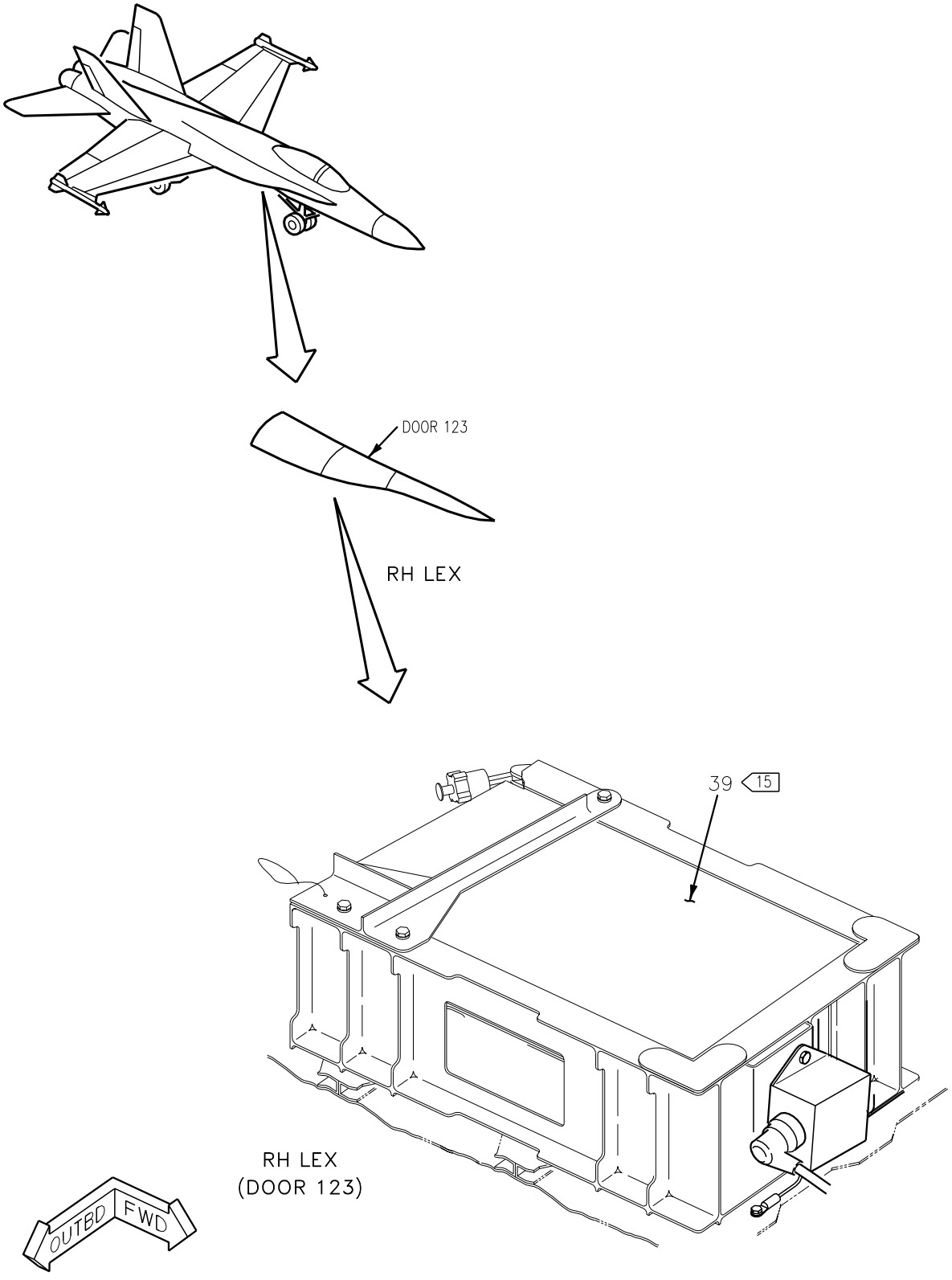
18AC-SCM-00-(2-10)31-CATI

Figure 1. Component Locator (Sheet 10)



18AC-SCM-00-(2-11)28-CATI

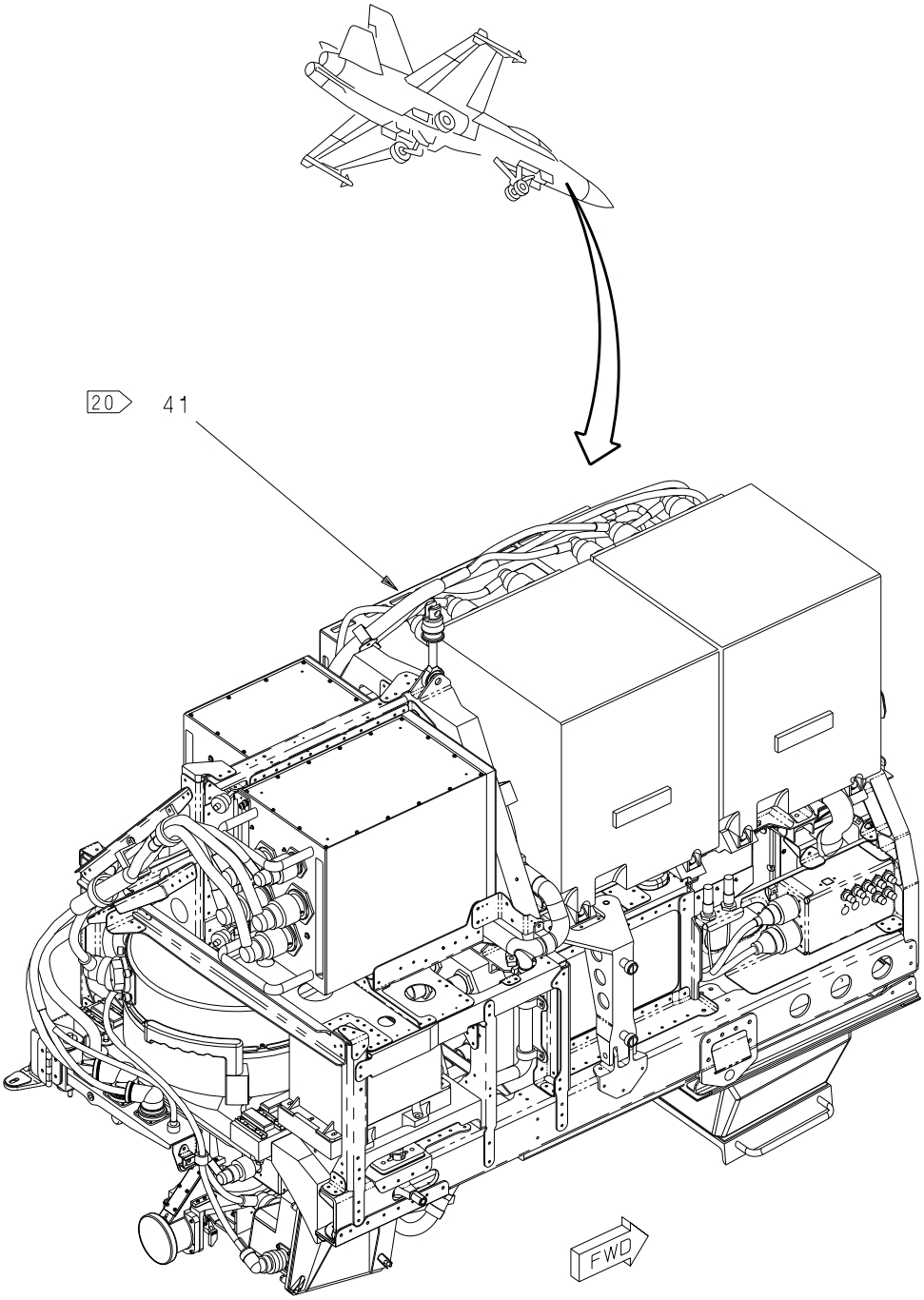
Figure 1. Component Locator (Sheet 11)



MAJOR CHANGE

18AC-SCM-00-(2-12)30-CATI

Figure 1. Component Locator (Sheet 12)

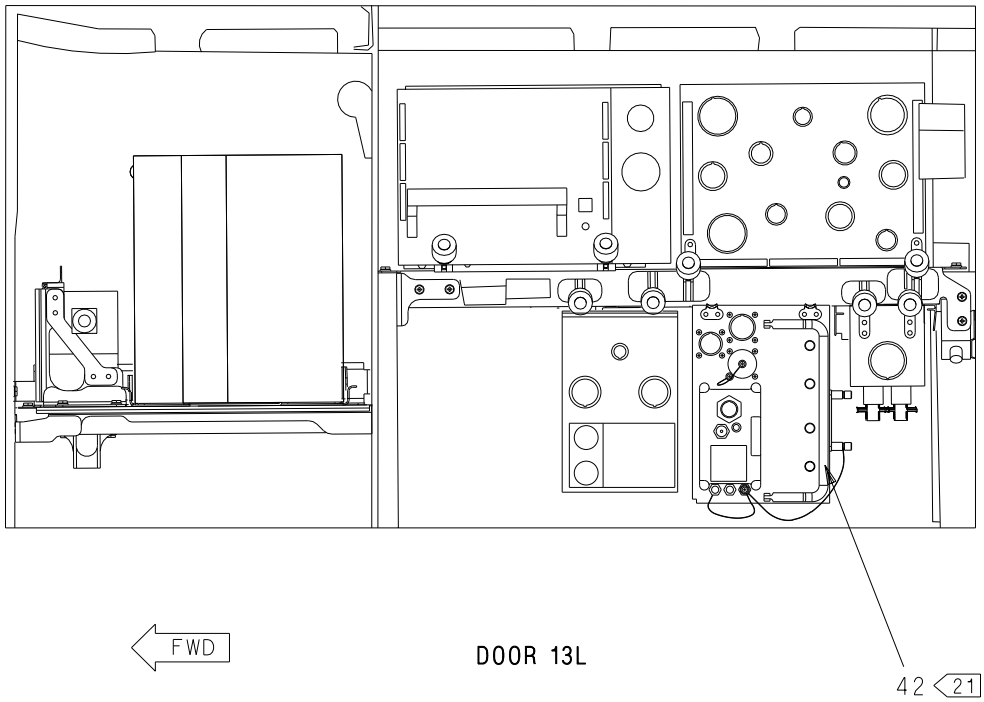


RECCE PALLET

MAJOR CHANGE

ADA541-2-13-036

Figure 1. Component Locator (Sheet 13)



MAJOR CHANGE

ADA541-2-14-039

Figure 1. Component Locator (Sheet 14)

Nomenclature	Index No.	Ref Des
AIR DATA COMPUTER	16	70A-F001
16 ALE-47 PROGRAMMER	40	65A-L003
17 ALE-47 PROGRAMMER	40	65A-K003
ALQ-165 PROCESSOR	12A	64A-E055
18 ARMAMENT COMPUTER CP-1342/AYQ-9(V)	19	61A-F001
19 ARMAMENT COMPUTER CP-2218/AYK-22(V)	19A	61A-F001
COMMAND LAUNCH COMPUTER CP-1001/AWG	23	61A-F010
14 COMPUTER-POWER SUPPLY CP-1325/APG-65	24	60A-A505
CONTROL-CONVERTER	17	82A-F001
CONTROL INDICATOR	6D	62A-J007
CONTROLLER-PROCESSOR	29	61A-P520
COUNTERMEASURES COMPUTER CP-1293/ALR-67(V)	13	62A-E006
9 DATA TRANSFER INTERFACE UNIT J-6008/A	31	85A-S130
20 DIGITAL COMPUTER-CP-2081/ASD-10(V)	41	89A-Y200
4 DIGITAL COMPUTER-CONVERTER CP-1805/AAR-50	28	61A-R561
DIGITAL DATA COMPUTER NO. 1	12	83A-E001
DIGITAL DATA COMPUTER NO. 2	21	83A-F002
10 DIGITAL MAP COMPUTER	30	80A-K025
5 DIGITAL MAP COMPUTER	32	80A-L025
2 DIGITAL DISPLAY INDICATOR ID-2150/ASM-612	9	85A-G003
ECM CONTROL PANEL ASSEMBLY	6C	52A-H087
ELECTRONIC EQUIPMENT CONTROL	4	79A-J006
FLAPS, LANDING GEAR AND STORES PANEL	1B	52A-H084
FORWARD AZIMUTH INDICATOR	6B	62A-J008

Figure 1. Component Locator (Sheet 15)

Nomenclature	Index No.	Ref Des
GND PWR CONTROL PANEL ASSEMBLY	1	1A-H004
INERTIAL NAVIGATION UNIT	11	68A-E001
INTERCOMMUNICATION AMPLIFIER-CONTROL	7	76A-H009
INTERCONNECTING BOX J-3656/ASQ-173	27	61AAR510
LEFT DIGITAL DISPLAY INDICATOR	2	80A-H001
LEFT HAND ADVISORY AND THREAT WARNING INDICATOR PANEL	6A	52A-H073
LEFT HAND VERTICAL CONSOLE CONTROL PANEL	1A	52A-H077
LEFT THROTTLE GRIP	8	52A-H049
MC/HYD ISOL CONTROL PANEL ASSEMBLY	6	52A-H081
1 MEMORY UNIT MOUNT MT-6450/ASQ-194	26	85MTK040
1 MEMORY UNIT MU-860/ASQ-194	25	85A-K503
1 NOSE WHEELWELL DDI	10	85A-G003
13 RADAR DATA PROCESSOR CP-2062/APG-73	38	60A-A503
15 RADIO RECEIVER R-2512A/U (GPS)	39	91A-N001
21 RADIO RECEIVER-TRANSMITTER RT-1763 (CIT)	42	78A-E016
RECEIVER-TRANSMITTER ALQ-126B	12A	64A-E001
11 RECEIVER-TRANSMITTER RT-1250( )/ARC NO. 1	15	76A-F001
12 RECEIVER-TRANSMITTER RT-1556( )/ARC NO. 1	15	76A-F041
11 RECEIVER-TRANSMITTER RT-1250( )/ARC NO. 2	18	76A-F002
12 RECEIVER-TRANSMITTER RT-1556( )/ARC NO. 2	18	76A-F042
5 RECEIVER-TRANSMITTER PROCESSOR RT-1379 ( ) ASW	33	77A-K001
6 RECEIVER-TRANSMITTER PROCESSOR RT-1379 ( ) ASW	34	77A-L001
7 RECEIVER-TRANSMITTER PROCESSOR RT-1379 ( ) ASW	35	77A-L001
8 RECEIVER-TRANSMITTER PROCESSOR RT-1379 ( ) ASW	36	77A-K001

Figure 1. Component Locator (Sheet 16)

Nomenclature	Index No.	Ref Des
10 RECEIVER-TRANSMITTER PROCESSOR RT-1379 ( ) ASW	37	77A-N001
RIGHT DIGITAL DISPLAY INDICATOR	3	80A-J002
ROLL-PITCH-YAW COMPUTER CP-1330/ASW-44 (FCCA)	14	84A-F001
ROLL-PITCH-YAW COMPUTER CP-1330/ASW-44 (FCCB)	22	84A-F002
1 SIGNAL DATA COMPUTER	20	85A-F042
SNSR POD CONTROL BOX PANEL ASSEMBLY	5	52A-J080

Figure 1. Component Locator (Sheet 17)

LEGEND

1 F/A-18C AND F/A-18D.

2 F/A-18A AND F/A-18B.

3 163427 THRU 163782.

4 163985 AND UP.

5 F/A-18C 163985 AND UP.

6 F/A-18A 161702 AND UP AND F/A-18C 163427 THRU 163782.

7 F/A-18A 161353 THRU 161528.

8 F/A-18B, F/A-18D 163434 THRU 163778.

9 164725 AND UP; ALSO 164627 THRU 164724 AFTER AFC 126.

10 F/A-18D 163986 AND UP.

11 F/A-18A AND F/A-18B; ALSO 163427 THRU 164897 BEFORE F/A-18 AFC 185.

12 164898 AND UP; ALSO 163427 THRU 164897 AFTER F/A-18 AFC 185.

13 164898 AND UP; ALSO 164627 THRU 164897 AFTER F/A-18 AFC 211 AND F/A-18A 162394 THRU 163175 AFTER F/A-18 AFC 292.

14 163427 THRU 164626; ALSO 164627 THRU 164897 BEFORE F/A-18 AFC 211.

15 164945 AND UP; ALSO 163429 THRU 164912 AFTER F/A-18 AFC 175 PT 2.

16 F/A-18D 165409 AND UP.

17 F/A-18C 165171 AND UP.

18 160775 THRU 165206.

19 165207 AND UP.

20 F/A-18D 164649 AND UP, RECCE INSTALLED.

21 165222 AND UP; ALSO 163985 THRU 165221 AFTER F/A-18 AFC 236 AND F/A-18A 162394 THRU 163175 AFTER F/A-18 AFC 292.

Figure 1. Component Locator (Sheet 18)

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

LOAD/VERIFICATION PROCEDURES USING LOADER-VERIFIER TEST SET

This WP supersedes WP006 00, dated 1 July 1996.

Title	WP Number
Avionics Load/Verification Procedures Using AN/ASM-607(V)5 Loader-Verifier Test Set F/A-18A AND F/A-18B	006 01
Avionics Load/Verification Procedures Using AN/ASM-607(V)5 Loader-Verifier Test Set F/A-18C AND F/A-18D	006 02
Avionics Load/Verification Procedures Using AN/ASM-687 Loader-Verifier Test Set F/A-18C AND F/A-18D	006 03
Avionics Load/Verification Procedures Using AN/USQ-131 Loader-Verifier Set F/A-18C AND F/A-18D	006 04
Avionics Load/Verification Procedures Using AN/USQ-131 Loader-Verifier Set F/A-18A AND F/A-18B	006 05
EW Load/Verification Procedures Using AN/USQ-131 Loader-Verifier Set F/A-18C AND F/A-18D	006 06
EW Load/Verification Procedures Using AN/USQ-131 Loader-Verifier Set F/A-18A AND F/A-18B	006 07

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

AVIONICS LOAD/VERIFICATION PROCEDURES USING AN/ASM-607(V)5 LOADER-VERIFIER TEST SET

EFFECTIVITY: F/A-18A AND F/A-18B

Reference Material

Line Maintenance Procedures	A1-F18AC-LMM-000
Line Maintenance Access Doors	A1-F18AC-LMM-010
Software Configuration Manual	A1-F18AC-SCM-000
Program Load Versions	WP003 00
Program Load CONFIG/IDENT Verification	WP004 00
Component Locator	WP005 00
Test Equipment Hookup Locator	WP007 00
Maintenance Status Display and Recording System	A1-F18AC-580-300
Magnetic Tape Cartridge MX-9972/ASM-612	WP004 00
Multipurpose Display Group	A1-F18AC-745-200
Displays Test F/A-18A	WP004 00
Displays Test F/A-18B	WP005 00

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Record of Applicable Technical Directives

None

1. INTRODUCTION.

2. This work package includes procedures for loading operational flight programs (OFP) into the digital data computer no. 1 and no. 2 (MC1, MC2), the Armament Computer CP-1342/AYQ-9(V) (SMS) and Command Launch Computer CP-1001( )/AWG (CLC) using the Computer Memory Loader-Verifier Test Set AN/ASM-607(V)5 (MLV).

Table 1. MC1 Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Digital Data Computer No. 1 (MC1) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-607(V)5 — Computer Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00. If Digital Data Computer is to be reloaded as a result of MMP code 34 make sure codes have been cleared on the Digital Display Indicator, ID-2150/ASM-612.		
1. PRELIMINARY. a. On MLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W3P1 to CMPTR connector (J2). b. In aircraft nose wheelwell: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W3P2 to MUX test connector (83J-G003).		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
2. PROCEDURE.		
a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
c. On MLV do substeps below:		
(1) Set POWER switch to ON.	MLV displays POWER UP, then LOT 9/DWN-1 10/UP-2.	Replace MLV.
(2) Select F/A-18A and F/A-18B aircraft by pressing 1 on keyboard.	[1] MLV displays MODE F18 XXXXX.	Replace MLV.
(3) Enter select terminal command by pressing S then T on keyboard.	MLV displays 1-MC1 2-MC2 3-SMS.	Replace MLV.
(4) Select MC1 by pressing 1 on keyboard.	[1] MLV displays MODE F18 XXXXX.	Replace MLV.
d. On MC/HYD ISOL control panel assembly, set MC switch to 2 OFF.	MC switch remains in 2 OFF position.	Do table 5 (A1-F18AC-741-200, WP008 00).
e. On MLV do substeps below:		
(1) Enter auto load command by pressing A then L on keyboard.	MLV displays AL IDΔ.	Replace MLV.
NOTE See WP003 00 for correct program part number and to verify program identification (PID) number.		
(2) Enter MC1 PID number from MLV instruction decal by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. [2] MLV displays AL XXX.	Replace MLV.

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Press ENTER.	1. ENTER light goes off.	Replace MLV.
	2. 2 ▸ MLV displays AL XXX. SEARCHING, then CYCLE POWER/ON MC1 +.	1. Replace tape transport unit. 2. If program still does not load, replace MLV.
f. On MC HYD/ISOL control panel assembly, move MC switch from 2 OFF to 1 OFF, then back to 2 OFF.	MC switch remains in 2 OFF position.	Do table 5 (A1-F18AC-741-200, WP008 00).
g. On MLV, press + on keyboard to continue command.	2 ▸ MLV displays AL XXX LOADING, then AL XXX DONE.	1. Replace tape transport unit. 2. If program still does not load, replace MLV.
3. TURN OFF.		
a. On GND PWR control panel assembly, set 1 switch to AUTO.		
b. On MLV, set POWER switch to OFF.		
c. Replace Magnetic Tape Cartridge MX-9972/ASM-612 (A1-F18AC-580-300, WP004 00)		
d. If OFP is to be loaded into MC2, SMS, or CLC do substeps below :		
(1) If MC2, do table 2, steps 2b thru 3c.		
(2) If SMS, do table 3, steps 2b thru 3c.		
(3) If CLC, do table 4, steps 1a thru 4c.		
e. Do table 1, WP004 00, to verify correct program identification.		
f. Remove electrical power (A1-F18AC-LMM-000).		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
g. On MLV do substeps below: (1) Disconnect W3P1 from CMPTR connector (J2). (2) Disconnect W1P2 from POWER connector (J1). h. In aircraft nose wheelwell: (1) Disconnect W3P2 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089). i. Do displays test listed below: (1) For F/A-18A, see A1-F18AC-745-200, WP004 00. (2) For F/A-18B, see A1-F18AC-745-200, WP005 00. j. Enter, as required, magnetic variation into mission computer memory (A1-F18AC-LMM-000, WP043 00).		

LEGEND

[1] XXXXX is release date of the program in tape transport unit.
[2] XXX is entered PID number.

Table 2. MC2 Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Digital Data Computer No. 2 (MC2) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-607(V)5 — Computer Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00. If Digital Data Computer is to be reloaded as a result of MMP code 37 make sure codes have been cleared on the Digital Display Indicator, ID-2150/ASM-612.		
1. PRELIMINARY. a. On MLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W3P1 to CMPTR connector (J2). b. In aircraft nose wheelwell: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W3P2 to MUX test connector (83J-G003).		

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
2. PROCEDURE.		
a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, troubleshoot (A1-F18AC-FIM-000, WP012 00).
c. On MLV do substeps below:		
(1) Set POWER switch to ON.	MLV displays POWER UP, then LOT 9/DWN-1 10/UP-2.	Replace MLV.
(2) Select F/A-18A and F/A-18B aircraft by pressing 1 on keyboard.	1 ▶ MLV displays MODE F18 XXXXX.	Replace MLV.
(3) Enter select terminal command by pressing S then T on keyboard.	MLV displays 1-MC1 2-MC2 3-SMS.	Replace MLV.
(4) Select MC2 by pressing 2 on keyboard.	1 ▶ MLV displays MODE F18 XXXXX.	Replace MLV.
d. On MC HYD/ISOL control panel assembly, hold MC switch in 1 OFF position.		
e. On MLV do substeps below:		
(1) Enter auto load command by pressing A then L on keyboard.	MLV displays AL ID Δ.	Replace MLV.
NOTE See WP003 00 for correct program part number and to verify program identification (PID) number.		
(2) Enter MC2 PID number from MLV instruction decal by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. 2 ▶ MLV displays AL IDΔ XXX.	Replace MLV.

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Press ENTER.	1. ENTER light goes off.	Replace MLV.
	2. 2 ▸ MLV displays AL XXX SEARCHING, then CYCLE POWER/ON MC2 +.	1. Replace tape transport unit. 2. If program still does not load, replace MLV.
f. On MC HYD/ISOL control panel assembly, move MC switch from 1 OFF to 2 OFF, then back to 1 OFF.	MC switch remains in 1 OFF position.	Do Table 5, (A1-F18AC-741-200, WP008 00).
g. On MLV, press + on keyboard to continue command.	2 ▸ MLV displays AL XXX LOADING, then AL XXX DONE.	1. Replace tape transport unit. 2. If program still does not load, replace MLV.
3. TURN OFF.		
a. On GND PWR control panel assembly, set 1 switch to AUTO.		
b. On MLV, set POWER switch to OFF.		
c. If OFP is to be loaded into MC1, SMS, or CLC, do substeps below:		
(1) If MC1, do table 1, steps 2b thru 3d.		
(2) If SMS, do table 3, steps 2b thru 3c.		
(3) If CLC, do table 4, steps 1a thru 4c.		
d. Do table 1, WP004 00, to verify correct program identification.		
e. Remove electrical power (A1-F18AC-LMM-000).		
f. On MLV do substeps below:		
(1) Disconnect W3P1 from CMPTR connector (J2).		

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Disconnect W1P2 from POWER connector (J1). g. In aircraft nose wheelwell: (1) Disconnect W3P2 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

1 XXXXX is release date of the program on tape transport unit.
2 XXX is entered PID number.

Table 3. SMS Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Armament Computer CP-1342/AYQ-9(V) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-607(V)5 — Computer Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY. a. On MLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W3P1 to CMPTR connector (J2). b. In aircraft nose wheelwell: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W3P2 to MUX test connector (83J-G003).		

Table 3. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
2. PROCEDURE.		
a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, troubleshoot (A1-F18AC-FIM-000, WP012 00).
c. On MLV do substeps below:		
(1) Set POWER switch to ON.	MLV displays POWER UP, then LOT 9/DWN-1 10/UP-2.	Replace MLV.
(2) Select F/A-18A and F/A-18B aircraft by pressing 1 on keyboard.	[1] MLV displays MODE F18 XXXXX.	Replace MLV.
(3) Enter select terminal command by pressing S then T on keyboard.	MLV displays 1-MC1 2-MC2 3-SMS.	Replace MLV.
(4) Select SMS by pressing 3 on keyboard.	[1] MLV displays MODE F18 XXXXX.	Replace MLV.
(5) Enter auto load command by pressing A then L on keyboard.	MLV displays AL ID Δ.	Replace MLV.
NOTE See WP003 00 for correct program part number and to verify program identification (PID) number.		
(6) Enter SMS PID number from MLV instruction decal by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. [2] MLV displays AL XXX.	Replace MLV.
(7) Press ENTER.	1. ENTER light goes off.	Replace MLV.

Table 3. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
	2. [2] MLV displays AL XXX SEARCHING, then AL XXX LOADING, then AL XXX DONE.	1. Replace tape transport unit. 2. If program still does not load, replace MLV.
3. TURN OFF. a. On GND PWR control panel assembly, set 3 switch to AUTO. b. On MLV, set POWER switch to OFF. c. If OFP is to be loaded into MC1 , MC2, or CLC, do substeps below: (1) If MC1, do table 1, steps 2b thru 3d. (2) If MC2, do table 2, steps 2b thru 3c. (3) If CLC, do table 4, steps 1a thru 4c. d. Do table 1, WP004 00, to verify correct program identification. e. Remove electrical power (A1-F18AC-LMM-000). f. On MLV do substeps below: (1) Disconnect W3P1 from CMPTR connector (J2). (2) Disconnect W1P2 from POWER connector (J1). g. In aircraft nose wheelwell: (1) Disconnect W3P2 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle 1J-G089).		

Table 3. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
LEGEND 1 ▶ XXXXX is release data of program on tape transport unit. 2 ▶ XXX is entered PID number.		

Table 4. CLC Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Command Launch Computer CP-1001( )/AWG **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-607(V)5 — Computer Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. STORES SAFETY INSPECTION (A1-F18AC-LWS-000).		

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
WARNING To prevent injury or death of personnel, all live weapons and explosive cartridges must be removed from aircraft and gun must be safetied before doing this test.		
a. Make sure electrical power is off (A1-F18AC-LMM-000).		
b. Make sure all weapons are removed from aircraft.		
c. Make sure all explosive cartridges are removed from cartridge chambers for all Aircraft Bomb Ejection Racks BRU-32 ( ) installed on aircraft.		
d. Make sure all explosive cartridges are removed from cartridge chambers on Aircraft Bomb Ejector Racks BRU-33 ( ) if installed on aircraft.		
e. Make sure all explosive cartridges are removed from cartridge chambers on Aircraft Guided Missile Launcher LAU-116/A AIM-7 fuselage stations if installed on aircraft.		
f. Make sure all Aircraft Guided Missile Launcher LAU-116/A hooks are closed and SAFETY RELEASE knob is rotated clockwise.	SAFETY RELEASE INDICATOR shows GREEN - HOOKS LOCKED.	1.With hooks closed, rotate SAFETY RELEASE knob clockwise. 2. If knob will not rotate, replace Aircraft Guided Missile Launcher LAU-116/A (A1-F18AC-740-300, WP026 00).
g. Make sure all explosives are removed from breeches on Multiple Ejector Racks (MER) if installed on aircraft.		

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
h. Make sure gun electrical signal safety switch is set to safe (extended) position, aft of door 6. i. Make sure gun hold - back mechanism handle is set to cleared; gun holdback handle indicator (extended). j. Close hooks on Bomb Ejector Racks BRU-32( ) for station used to ID HARM and set ground safety handle to LOCKED.		
CAUTION Never set MLV power switch to OFF while any of the following is displayed: SEARCHING TAPE, LOADING CLC, READING FILE RECORD, VERIFYING.		
2. PRELIMINARY. a. On MLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W3P1 to CMPTR connector (J2). b. In aircraft nose wheelwell: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). 3. PROCEDURE. a. Open door 14R (A1-F18AC-LMM-010). b. On Armament/Computer CP-1342/AYQ-9(V):		

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Set ARMAMENT switches to 64 for station used to ID HARM in step 1j. (2) For remaining stations set switches to 00, except stations with tanks installed set switches to 01. c. Apply electrical power (A1-F18AC-LMM-000).		
d. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, troubleshoot (A1-F18AC-FIM-000, WP012 00).
e. Connect ground intercommunications (A1-F18AC-LMM-000). f. On SNSR pod control box panel assembly, make sure RADAR switch is OFF.		
NOTE If a malfunction occurs during this test, make sure circuit breakers shown in (A1-F18AC-740-220, WP008 00) are closed.		
g. On MC/HYD ISOL control panel assembly, set MC switch to NORM.		
h. After 80 to 180 seconds have elapsed, select A/G master mode button.	A/G master mode button lights.	Make sure enough time has elapsed for SMP self test to complete. Do step 3c through step 3h.
i. On JETT Station Select panel assembly do substeps below: (1) Select station used to ID HARM.	Selected station lights.	

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Move JETT select switch from Safe to Stores.	In avionics bay Door 13R, the CLC cooling fan comes on.	Replace CLC (A1-F18AC-740-300, WP010 00).
CAUTION Never set MLV power switch to OFF while any of the following is displayed: SEARCHING TAPE, LOADING CLC, READING FILE RECORD, VERIFYING.		
j. On MLV do substeps below:		
(1) Set power switch to ON.	1 MLV displays POWER UP, LOADING LIBRARIES, then MODE CLC XXXXXX.	Replace MLV.
NOTE For an operational CLC both a HARM operational program (PGM) and Electronic Intelligence (ELINT) files must be loaded.		
(2) If PGM is to be loaded into CLC do step 3j3. If ELINT is to be loaded into CLC do step 3j9.		
(3) Enter load program command by pressing L then P on keyboard.	1. ENTER light comes on. 2. MLV displays LOAD PGM.	Replace MLV.
(4) Press ENTER.	1. ENTER light goes off. 2. MLV displays ENTER PGM ID Δ.	Replace MLV.
NOTE See WP003 00 for correct program part number and to verify program identification (PID) number.		
(5) Enter PGM PID number from MLV instruction decal by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. 2 MLV displays ENTER PGM IDΔ XXX.	Replace MLV.
(6) Press ENTER.	1. ENTER light goes off.	Replace MLV.

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
	2. MLV displays SEARCHING TAPE, LOADING CLC, READING FILE RECORD, VERIFYING, then LOAD PGM DONE.	1. IF MLV Display BUS ERROR 010101010101 do table 5, this WP. 2. Replace tape transport unit.
	3. ENTER light comes on.	3. If program still does not load, replace MLV.
(7) Press ENTER.	1. ENTER light goes off.	Replace MLV.
	2. [1] MLV displays MODE CLC XXXXXX.	
(8) If ELINT is to be loaded into CLC do step 3j9. If not go to step 4.		
(9) Enter load ELINT command by pressing L then E on keyboard.	1. ENTER light comes on. 2. MLV displays LOAD ELINT.	Replace MLV.
(10) Press ENTER.	1. ENTER light goes off. 2. MLV displays ENTER ELINT ID Δ.	Replace MLV.
NOTE See WP003 00 for correct program part number and to verify program identification (PID) number.		
(11) Enter ELINT file PID number from MLV instruction decal by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. [2] MLV displays ENTER ELINT IDΔ XXX.	Replace MLV.
(12) Press ENTER.	1. ENTER light goes off.	Replace MLV.
	2. MLV displays SEARCHING TAPE, LOADING CLC, READING FILE RECORD, VERIFYING, then LOAD ELINT DONE.	1. Replace tape transport unit. 2. If program still does not load, replace MLV.
	3. ENTER light comes on.	
(13) Press ENTER.	1. ENTER light goes off.	Replace MLV.

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
	2. [1] MLV displays MODE CLC XXXXXX.	
k. Verify PGM file by doing substeps below on MLV:		
(1) Press V then P on the keyboard.	1. ENTER light comes on. 2. MLV display VERIFY PGM.	Replace MLV.
(2) Press ENTER.	1. ENTER light goes off. 2. MLV displays ENTER PGM IDΔ.	Replace MLV.
(3) Enter PGM file PID to be verified.	1. ENTER light comes on. 2. [2] MLV displays ENTER PGM IDΔ XXX.	Replace MLV.
(4) Press ENTER.	1. ENTER light goes off. 2. MLV displays SEARCHING TAPE, READING FILE RECORD, VERIFYING, VERIFY PGM DONE. 3. ENTER light comes on.	Replace MLV.
(5) Press ENTER.	1. ENTER light goes off. 2. [1] MLV displays MODE CLC XXXXXX.	Replace MLV.
l. Verify ELINT file by doing substeps below on MLV:		
(1) Press V then E on the keyboard.	1. ENTER light comes on. 2. MLV displays VERIFY ELINT.	Replace MLV.
(2) Press ENTER.	1. ENTER light goes off. 2. MLV displays ENTER ELINT IDΔ.	Replace MLV.

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Enter ELINT file PID to be verified.	1. ENTER light comes on. 2. [2] ▶ MLV displays ENTER ELINT IDΔ XXX.	Replace MLV.
(4) Press ENTER.	1. ENTER light goes off. 2. MLV displays SEARCHING TAPE, READING FILE, RECORD, VERIFYING, VERIFY ELINT DONE. 3. ENTER light comes on.	Replace MLV.
(5) Press ENTER.	1. ENTER light goes off. 2. [1] ▶ MLV displays MODE CLC XXXXXX.	Replace MLV.
4. TURN OFF. a. On GND PWR control panel assembly, set 3 switch to AUTO.		
CAUTION Never set MLV power switch to OFF while any of the following is displayed: SEARCHING TAPE, LOADING CLC, READING FILE RECORD, VERIFYING.		
b. On MLV, set POWER switch to OFF.		
c. If OFP is to be loaded into MC1, MC2, or SMS, do steps 4d and 4e, then do substeps below: (1) If MC1, do table 1, steps 2b thru 3d. (2) If MC2, do table 2, steps 2b thru 3c. (3) If SMS, do table 3, steps 2b thru 3c.		

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
d. Remove electrical power (A1-F18AC-LMM-000). e. Do table 4, WP004 00, to verify correct program identification. f. On MLV do substeps below: (1) Disconnect W3P1 from CMPTR connector (J2). (2) Disconnect W1P2 from POWER connector (J1). g. In aircraft nose wheelwell: (1) Disconnect W3P2 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

1 XXXXXX is release date of the program on tape transport unit.
2 XXX is entered PID number.

Table 5. MLV Displays BUS ERROR Loading CLC

Procedure	Normal Indication	Remedy for Abnormal Indication
CAUTION Never set MLV power switch to OFF while any of the following is displayed: SEARCHING TAPE, LOADING CLC, READING FILE RECORD, VERIFYING.		
1. On MLV press ENTER. 2. On MLV, set POWER switch to OFF. 3. On GND PWR control panel assembly, set 3 switch to AUTO. 4. Wait at least 30 seconds.		
5. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switches remain on (latched).	1. If switches unlatch in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switches will not remain on, troubleshoot (A1-F18AC-FIM-000, WP012 00).
6. Do Table 4 step 3j.		

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

AVIONICS LOAD/VERIFICATION PROCEDURES USING AN/ASM-607(V)5 LOADER-VERIFIER TEST SET

EFFECTIVITY: F/A-18C AND F/A-18D

Reference Material

Flight Incident Recorder and Monitoring System	A1-F18AE-580-300
Memory Unit MU-806/ASQ-194	WP005 00
Line Maintenance Procedures	A1-F18AC-LMM-000
Line Maintenance Access Doors	A1-F18AC-LMM-010
Multipurpose Display Group	A1-F18AC-745-200
Displays Test F/A-18C	WP004 00
Displays Test F/A-18D	WP005 00
Multipurpose Display Group	A1-F18AG-745-200
Displays Test F/A-18C	WP004 00
Displays Test F/A-18D	WP005 00
Software Configuration Manual	A1-F18AC-SCM-000
Program Load Versions	WP003 00
Program Load CONFIG/IDENT Verification	WP004 00
Component Locator	WP005 00
Test Equipment Hookup Locator	WP007 00

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Record of Applicable Technical Directives

None

1. INTRODUCTION.

2. This work package includes procedures for loading operational flight programs (OFP) into the digital data computer no. 1 and no. 2 (MC1, MC2), the Armament Computer CP-1342/AYQ-9(V) (SMS), the Signal Data Computer CP-1726/ASQ-194 (SDC), and the Command Launch Computer CP-1001( )/ AWG (CLC) using the Computer Memory Loader-Verifier Test Set AN/ASM-607(V)5 (MLV).

Table 1. MC1 Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Digital Data Computer No. 1 (MC1) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-607(V)5 — Computer Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00. If Digital Data Computer is to be reloaded as a result of MMP code 34 make sure codes have been cleared on the Aircraft Maintenance Indicator, ID-2388/ASQ-194.		
1. PRELIMINARY. a. On MLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W3P1 to CMPTR connector (J2).		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W3P2 to MUX test connector (83J-G003). c. Remove Memory Unit MU-806/ASQ-194 (A1-F18AE-580-300, WP005 00) and remove data stored. 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, troubleshoot (A1-F18AC-FIM-000, WP012 00).
c. On MLV do substeps below:		
(1) Set POWER switch to ON.	MLV displays POWER UP, then LOT 9/DWN-1 10/UP-2.	Replace MLV.
(2) Select F/A-18C and F/A-18D aircraft by pressing 2 on keyboard.	1 ▸ MLV displays SEARCHING, then MODE F18 XXXXX.	Replace MLV.
(3) Enter select terminal command by pressing S then T on keyboard.	MLV displays MC-1 OR 2 SMS-3 SDC-4.	Replace MLV.
(4) Select MC1 by pressing 1 on keyboard.	1 ▸ MLV displays MODE F18 XXXXX.	Replace MLV.
d. On MC/HYD ISOL control panel assembly, set MC switch to 2 OFF.	MC switch remains in 2 OFF position.	Do table 5 (A1-F18AE-741-200, WP008 00).
e. On MLV do substeps below:		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Enter auto load command by pressing A then L on keyboard.	MLV displays AL ID∆.	Replace MLV.
NOTE See WP003 00 for correct program part number and to verify program identification (PID) number.		
(2) Enter MC1 PID number from MLV instruction decal by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. 2 MLV displays AL XXX.	Replace MLV.
NOTE Make sure program identifier entered and displayed on the MLV is MC1 program.		
(3) Press ENTER.	1. ENTER light goes off. 2. 2 MLV displays AL XXX SEARCHING, then CYCLE POWER/ON MC1 +.	Replace MLV. 1. Replace tape transport unit. 2. If program still does not load, replace MLV.
f. On MC HYD/ISOL control panel assembly, move MC switch from 2 OFF to 1 OFF, then back to 2 OFF.	MC switch remains in 2 OFF position.	Do table 5 (A1-F18AE-741-200, WP008 00).
g. On MLV, press + on keyboard to continue command.	2 MLV displays AL XXX LOADING, then AL XXX DONE.	1. Replace tape transport unit. 2. If program still does not load, replace MLV.
3. TURN OFF. a. On GND PWR control panel assembly, set 1 switch to AUTO. b. On MLV, set POWER switch to OFF. c. Replace Memory Unit MU-806/ASQ-194 (A1-F18AE-580-300, WP005 00).		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
d. If OFP is to be loaded into MC2, SMS Boot, SMS, SDC, or CLC, do substeps below : (1) If MC2, do table 2, steps 2b thru 3c. (2) If SMS Boot, do table 3, steps 2b thru 3c. (3) If SMS, do table 4, steps 2b thru 3c. (4) If SDC, do table 5, steps 2b thru 3c. (5) If CLC, do table 6, steps 1a thru 4c.		
NOTE Electrical power must be removed prior to MC1 CONFIG/IDENT verification or system operation to make sure of proper power up sequencing in mission computer.		
e. Remove electrical power (A1-F18AC-LMM-000).		
f. Do applicable table, WP004 00, to verify correct program identification.		
g. On MLV do substeps below: (1) Disconnect W3P1 from CMPTR connector (J2). (2) Disconnect W1P2 from POWER connector (J1).		
h. In aircraft nose wheelwell: (1) Disconnect W3P2 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
i. Do displays test below: ON F/A-18C 163427 THRU 163782, A1-F18AC-745-200, WP004 00. ON F/A-18D 163434 THRU 163778, A1-F18AC-745-200, WP005 00. ON F/A-18C 163985 AND UP, A1-F18AG-745-200, WP004 00. ON F/A-18D 163986 AND UP, A1-F18AG-745-200, WP005 00. j. Enter, as required, magnetic variation into mission computer memory (A1-F18AC-LMM-000, WP043 00).		

LEGEND

[1] XXXXX is release date of the program in tape transport unit.
[2] XXX is entered PID number.

Table 2. MC2 Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Digital Data Computer No. 2 (MC2) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-607(V)5 — Computer Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00. If Digital Data Computer is to be reloaded as a result of MMP code 37 make sure codes have been cleared on the Aircraft Maintenance Indicator, ID-2388/ASQ-194.		
1. PRELIMINARY. a. On MLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W3P1 to CMPTR connector (J2). b. In aircraft nose wheelwell: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W3P2 to MUX test connector (83J-G003).		

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
2. PROCEDURE.		
a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, troubleshoot (A1-F18AC-FIM-000, WP012 00).
c. On MLV do substeps below:		
(1) Set POWER switch to ON.	MLV displays POWER UP, then LOT 9/DWN-1 10/UP-2.	Replace MLV.
(2) Select F/A-18C and F/A-18D aircraft by pressing 2 on keyboard.	1 ▸ MLV displays MODE F18 XXXXX.	Replace MLV.
(3) Enter select terminal command by pressing S and then T on keyboard.	MLV displays MC-1 or 2, SMS-3, SDC-4.	Replace MLV
(4) Select MC2 by pressing 2 on keyboard.	1 ▸ MLV displays MODE F18 XXXXX.	
d. On MC HYD/ISOL control panel assembly, hold MC switch in 1 OFF position.		
NOTE Make sure program identifier entered and displayed on the MLV is MC2 program.		
e. On MLV do substeps below:		
(1) Enter auto load command by pressing A then L on keyboard.	MLV displays AL ID Δ.	Replace MLV.
(2) Enter MC2 PID number from MLV instruction decal by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. 2 ▸ MLV displays AL XXX.	Replace MLV.

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Press ENTER.	1. ENTER light goes off.	Replace MLV.
	2. 2 ▸ MLV displays AL XXX SEARCHING, then CYCLE POWER/ON MC2 +.	1. Replace tape transport unit. 2. If program still does not load, replace MLV.
f. On MC HYD/ISOL control panel assembly, move MC switch from 1 OFF to 2 OFF, then back to 1 OFF.	MC switch remains in 1 OFF position.	Do Table 5, (A1-F18AE-741-200, WP008 00).
g. On MLV, press + on keyboard to continue command.	2 ▸ MLV displays AL XXX LOADING, then AL XXX DONE.	1. Replace tape transport unit. 2. If program still does not load, replace MLV.
3. TURN OFF.		
a. On GND PWR control panel assembly, set 1 switch to AUTO.		
b. On MLV, set POWER switch to OFF.		
c. If OFP is to be loaded into MC1, SMS Boot, SMS, SDC, or CLC, do substeps below:		
(1) If MC1, do table 1, steps 2b thru 3d.		
(2) If SMS Boot, do table 3, steps 2b thru 3c.		
(3) If SMS, do table 4, steps 2b thru 3c.		
(4) If SDC, do table 5 steps 2b thru 3c.		
(5) If CLC, do table 6, steps 1a thru 4c.		

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE Electrical power must be removed prior to MC2 CONFIG/IDENT verification or system operation to make sure of proper power up sequencing in mission computers.		
d. Remove electrical power (A1-F18AC-LMM-000).		
e. Do applicable table, WP004 00, to verify correct program identification.		
f. On MLV do substeps below:		
(1) Disconnect W3P1 from CMPTR connector (J2).		
(2) Disconnect W1P2 from POWER connector (J1).		
g. In aircraft nose wheelwell:		
(1) Disconnect W3P2 from MUX test connector (83J-G003).		
(2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

[1] XXXXX is release date of the program on tape transport unit.
[2] XXX is entered PID number.

Table 3. SMS Boot Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Armament Computer CP-1342/AYQ-9(V) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-607(V)5 — Computer Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY. a. On MLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W3P1 to CMPTR connector (J2). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W3P2 to MUX test connector (83J-G003).		

Table 3. SMS Boot Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
2. PROCEDURE.		
a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, troubleshoot (A1-F18AC-FIM-000, WP012 00).
c. On MLV do substeps below:		
(1) Set POWER switch to ON.	MLV displays POWER UP, then LOT 9/DWN-1 10/UP-2.	Replace MLV.
(2) Select F/A-18C and F/A-18D aircraft by pressing 2 on keyboard.	1 MLV displays SEARCHING, then MODE F18 XXXXX.	Replace MLV.
(3) Enter select terminal command by pressing S then T on keyboard.	MLV displays MC-1 OR 2 SMS-3 SDC-4.	Replace MLV.
(4) Select SMS by pressing 3 on keyboard.	1 MLV displays MODE F18 XXXXX.	Replace MLV.
(5) Enter auto load command by pressing A then L on keyboard.	MLV displays AL ID Δ.	Replace MLV.
NOTE See WP003 00 for correct program part number and to verify program identification (PID) number.		
(6) Enter SMS Boot PID number from MLV instruction decal by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. 2 MLV displays AL XXX.	Replace MLV.
(7) Press ENTER.	1. ENTER light goes off.	Replace MLV.

Table 3. SMS Boot Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
	2. [2] MLV displays AL XXX SEARCHING, then BOOT? FLIR SW Y/N +.	1. Replace tape transport unit. 2. If program still does not load, replace MLV.
NOTE HARM target sequence/FLIR FOV/RAID must be hold until the SMS boot loading is complete.		
d. Do substeps below: (1) On throttle grip, press and hold HARM target sequence/ FLIR FOV/RAID switch.		
(2) On MLV, press + on keyboard.	[2] MLV displays AL XXX LOADING, then AL XXX DONE.	1. Replace tape transport unit. 2. If program still does not load, replace MLV.
e. On throttle grip, release HARM target sequence/ FLIR FOV/RAID switch.		
f. On GND PWR control panel assembly, set 3 switch to AUTO then set and hold to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, troubleshoot (A1-F18AC-FIM-000, WP012 00).
3. TURN OFF. a. On GND PWR control panel assembly, set 3 switch to AUTO. b. On MLV, set POWER switch to OFF. c. If OFP is to be loaded into MC1, MC2, SMS, SDC, or CLC, do substeps below : (1) If MC1, do table 1, steps 2b thru 3d.		

Table 3. SMS Boot Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) If MC2, do table 2, steps 2b thru 3c. (3) If SMS, do table 4, steps 2b thru 3c. (4) If SDC, do table 5, steps 2b thru 3c. (5) If CLC, do table 7, steps 1a thru 4c. d. Remove electrical power (A1-F18AC-LMM-000). e. Do applicable table, WP004 00, to verify correct program identification. f. On MLV do substeps below: (1) Disconnect W3P1 from CMPTR connector (J2). (2) Disconnect W1P2 from POWER connector (J1). g. In aircraft nose wheelwell do substeps below: (1) Disconnect W3P2 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

[1] XXXXX is release date of program on tape transport unit.
[2] XXX is entered PID number.

Table 4. SMS Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Armament Computer CP-1342/AYQ-9(V) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-607(V)5 — Computer Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY a. On MLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W3P1 to CMPTR connector (J2). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W3P2 to MUX test connector (83J-G003).		

Table 4. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
2. PROCEDURE.		
a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, troubleshoot (A1-F18AC-FIM-000, WP012 00).
c. On MLV do substeps below:		
(1) Set POWER switch to ON.	MLV displays POWER UP, then LOT 9/DWN-1 10/UP-2.	Replace MLV.
(2) Select F/A-18C and F/A-18D aircraft by pressing 2 on keyboard.	1 ▸ MLV displays SEARCHING, then MODE F18 XXXXX.	Replace MLV.
(3) Enter select terminal command by pressing S then T on keyboard.	MLV displays MC-1 OR 2 SMS-3 SDC-4.	Replace MLV.
(4) Select SMS by pressing 3 on keyboard.	1 ▸ MLV displays MODE F18 XXXXX.	Replace MLV.
(5) Enter auto load command by pressing A then L on keyboard.	MLV displays AL ID Δ.	Replace MLV.
NOTE See WP003 00 for correct program part number and to verify program identification (PID) number.		
(6) Enter SMS PID number from MLV instruction decal by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. 2 ▸ MLV displays AL XXX.	Replace MLV.
(7) Press ENTER.	1. ENTER light goes off.	Replace MLV.

Table 4. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
	2. 2 ▸ MLV displays AL XXX SEARCHING, then BOOT? FLIR SW Y/N +.	1. Replace tape transport unit. 2. If program still does not load, replace MLV.
(8) Press + on keyboard.	2 ▸ MLV displays AL XXX LOADING, then AL XXX DONE.	1. Replace tape transport unit. 2. If program still does not load, replace MLV.
d. On GND PWR control panel assembly, set 3 switch to AUTO then set and hold to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, troubleshoot (A1-F18AC-FIM-000, WP012 00).
3. TURN OFF.		
a. On GND PWR control panel assembly, set 3 switch to AUTO.		
b. On MLV, set POWER switch to OFF.		
c. If OFP is to be loaded into MC1, MC2, SMS Boot, SDC, or CLC, do substeps below:		
(1) If MC1, do table 1, steps 2b thru 3d.		
(2) If MC2, do table 2, steps 2b thru 3c.		
(3) If SMS Boot, do table 3, steps 2b thru 3c.		
(4) If SDC, do table 5, steps 2b thru 3c.		
(5) If CLC, do table 6, steps 1a thru 4c.		
d. Remove electrical power (A1-F18AC-LMM-000).		

Table 4. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
e. Do applicable table, WP004 00, to verify correct program identification. f. On MLV do substeps below: (1) Disconnect W3P1 from CMPTR connector (J2). (2) Disconnect W1P2 from POWER connector (J1). g. In aircraft nose wheelwell do substeps below: (1) Disconnect W3P2 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

[1] XXXXX is release date of program on tape transport unit.
[2] XXX is entered PID number.

Table 5. SDC Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Signal Data Computer CP-1726/ASQ-194 (SDC) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-607(V)5 — Computer Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY. a. On MLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W3P1 to CMPTR connector (J2). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W3P2 to MUX test connector (83J-G003).		

Table 5. SDC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On MLV do substeps below:		
(1) Set POWER switch to ON.	MLV displays POWER UP, then LOT 9/DWN-1 10/UP-2.	Replace MLV.
(2) Select F/A-18C and F/A-18D aircraft by pressing 2 on keyboard.	1 ▶ MLV displays SEARCHING, then MODE F18 XXXXX.	Replace MLV.
(3) Enter select terminal command by pressing S then T on keyboard.	MLV displays MC-1 OR 2 SMS-3 SDC-4.	Replace MLV.
(4) Select SDC by pressing 4 on keyboard.	1 ▶ MLV displays MODE F18 XXXXX.	Replace MLV.
(5) Enter auto load command by pressing A then L on keyboard.	MLV displays AL ID Δ.	Replace MLV.
NOTE See WP003 00 for correct program part number and to verify program identification (PID) number.		
(6) Enter SDC PID number from MLV instruction decal by pressing applicable numbers on keyboard. c. If the Signal Data Computer to be loaded does not contain any software load do substeps below: (1) Open door 10L (A1-F18AC-LMM-010). (2) On No. 8 Circuit Breaker/Relay Panel Assembly pull SDC circuit breaker at locator D2.	1. ENTER light comes on. 2. 2 ▶ MLV displays AL XXX.	Replace MLV.

Table 5. SDC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE CONSUMABLES CHECK switch must be pressed and held until circuit breaker is reset.		
(3) On Aircraft Maintenance Indicator ID-2388/ASQ-194 (nose wheelwell), press and hold CONSUMABLES CHECK switch. (4) On No. 8 Circuit Breaker/Relay Panel Assembly reset SDC circuit breaker at locator D2. (5) On Aircraft Maintenance Indicator ID-2388/ASQ-194 (nose wheelwell), press and release CONSUMABLES CHECK switch. (6) Go to step 2f. d. On Aircraft Maintenance-Indicator ID-2388/ASQ-194 (nose wheelwell), press and release CONSUMABLES CHECK switch.		
e. On MLV, press ENTER.	1. ENTER light goes off. 2. 2 ▶ MLV displays AL XXX SEARCHING, AL XXX LOADING, then AL XXX DONE.	Replace MLV. 1. Replace tape transport unit. 2. If program still does not load, replace MLV.
3. TURN OFF. a. On GND PWR control panel assembly, set 1 switch to AUTO. b. On MLV, set POWER switch to OFF. c. If OFP is to be loaded into MC1, MC2, SMS Boot, SMS, or CLC, do substeps below:		

Table 5. SDC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) If MC1, do table 1, steps 2b thru 3d. (2) If MC2, do table 2, steps 2b thru 3c. (3) If SMS Boot, do table 3, steps 2b thru 3c. (4) If SMS, do table 4, steps 2b thru 3c. (5) If CLC, do table 6, steps 1a thru 4c. d. Remove electrical power (A1-F18AC-LMM-000). e. Do applicable table, WP004 00, to verify correct program identification. f. On MLV do substeps below: (1) Disconnect W3P1 from CMPTR connector (J2). (2) Disconnect W1P2 from POWER connector (J1). g. In aircraft nose wheelwell do substeps below: (1) Disconnect W3P2 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

[1] XXXXX is release date of the program on tape transport unit.
[2] XXX is entered PID number.

Table 6. CLC Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Command Launch Computer CP-1001( )/AWG **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-607(V)5 — Computer Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. STORES SAFETY INSPECTION (A1-F18AE-LWS-000).		
WARNING To prevent injury or death of personnel, all live weapons and explosive cartridges must be removed from aircraft and gun must be safetied before doing this test.		
a. Make sure electrical power is off (A1-F18AC-LMM-000). b. Make sure all weapons are removed from aircraft.		

Table 6. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. Make sure all explosive cartridges are removed from cartridge chambers for all Aircraft Bomb Ejection Racks BRU-32 ( ) installed on aircraft.		
d. Make sure all explosive cartridges are removed from cartridge chambers on Aircraft Bomb Ejector Racks BRU-33 ( ) if installed on aircraft.		
e. Make sure all explosive cartridges are removed from cartridge chambers on Aircraft Guided Missile Launcher LAU-116/A AIM-7 fuselage stations if installed on aircraft.		
f. Make sure all Aircraft Guided Missile Launcher LAU-116/A hooks are closed and SAFETY RELEASE knob is rotated clockwise.	SAFETY RELEASE INDICATOR shows GREEN - HOOKS LOCKED.	1.With hooks closed, rotate SAFETY RELEASE knob clockwise. 2. If knob will not rotate, replace Aircraft Guided Missile Launcher LAU-116/A (A1-F18AC-740-300, WP026 00).
g. Make sure all explosives are removed from breeches on Multiple Ejector Racks (MER) if installed on aircraft.		
h. Make sure gun electrical signal safety switch is set to safe (extended) position, aft of door 6.		
i. Make sure gun hold - back mechanism handle is set to cleared; gun holdback handle indicator (extended).		
j. Close hooks on Bomb Ejector Racks BRU-32( ) for station used to ID HARM and set ground safety handle to LOCKED.		

Table 6. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
CAUTION Never set MLV power switch to OFF while any of the following is displayed: SEARCHING TAPE, LOADING CLC, READING FILE RECORD, VERIFYING.		
2. PRELIMINARY. a. On MLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W3P1 to CMPTR connector (J2). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). 3. PROCEDURE. a. Open door 14R (A1-F18AC-LMM-010). b. On Armament/Computer CP-1342/AYQ-9(V) do substeps below: (1) Set ARMAMENT switches to 64 for station used to ID HARM in step 1j. (2) For remaining stations set switches to 00, except stations with tank installed set switches to 01. c. Apply electrical power (A1-F18AC-LMM-000).		

Table 6. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
d. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. Connect ground intercommunications (A1-F18AC-LMM-000).		
f. On SNSR pod control box panel assembly, make sure RADAR switch is OFF.		
NOTE If a malfunction occurs during this test, make sure circuit breakers are closed; ON 163427 THRU 165206 (A1-F18AE-740-200, WP011 00) or ON 165207 AND UP (A1-F18AH-740-200, WP006 00).		
g. On MC/HYD ISOL control panel assembly, set MC switch to NORM.		
h. After 80 to 180 seconds, select A/G master mode button.	A/G master mode button lights.	Make sure enough time has elapsed for the SMS to complete self test, do steps 3c through 3h.
i. On flaps, landing gear and stores panel assembly, select station used to ID HARM.	Selected station light comes on.	ON 163427 THRU 165206 do table 1 (A1-F18AE-740-200, WP017 00). ON 165207 AND UP do table 1 (A1-F18AH-740-200, WP015 00).
j. On left hand vertical console control panel, move JETT select switch from SAFE to STORES.	In avionics bay door 13R, the CLC cooling fan comes on.	ON 163427 THRU 165206 replace CLC (A1-F18AE-740-300, WP011 00). ON 165207 AND UP replace CLC (A1-F18AH-740-300, WP008 00).

Table 6. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
CAUTION Never set MLV power switch to OFF while any of the following is displayed: SEARCHING TAPE, LOADING CLC, READING FILE RECORD, VERIFYING.		
k. On MLV do substeps below:		
(1) Set power switch to ON.	1 MLV displays POWER UP, LOADING LIBRARIES, then MODE CLC XXXXXX.	Replace MLV.
NOTE For an operational CLC both a HARM operational program (PGM) and Electronic Intelligence (ELINT) files must be loaded. PGM file must be loaded before ELINT file.		
(2) If PGM is to be loaded into CLC do step 3j3. If ELINT is to be loaded into CLC do step 3j9.		
(3) Enter load program command by pressing L then P on keyboard.	1. ENTER light comes on. 2. MLV displays LOAD PGM.	Replace MLV.
(4) Press ENTER.	1. ENTER light goes off. 2. MLV displays ENTER PGM ID Δ.	Replace MLV.
NOTE See WP003 00 for correct program part number and to verify program identification (PID) number.		
(5) Enter PGM PID number from MLV instruction decal by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. 2 MLV displays ENTER PGM IDΔ XXX.	Replace MLV.
(6) Press ENTER.	1. ENTER light goes off.	Replace MLV.

Table 6. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
	2. MLV displays SEARCHING TAPE, LOADING CLC, READING FILE RECORD, VERIFYING, then LOAD PGM DONE.	1. IF MLV Display BUS ERROR 010101010101 do table 7, this WP. 2. Replace tape transport unit. 3. If program still does not load, replace MLV.
	3. ENTER light comes on.	
(7) Press ENTER.	1. ENTER light goes off.	Replace MLV.
	2. [1] MLV displays MODE CLC XXXXXX.	
(8) If ELINT is to be loaded into CLC do step 3j9. If not go to step 4.		
(9) Enter load ELINT command by pressing L then E on keyboard.	1. ENTER light comes on.	Replace MLV.
	2. MLV displays LOAD ELINT.	
(10) Press ENTER.	1. ENTER light goes off.	Replace MLV.
	2. MLV displays ENTER ELINT ID Δ.	

NOTE

See WP003 00 for correct program part number and to verify program identification (PID) number.

(11) Enter ELINT file PID number from MLV instruction decal by pressing applicable numbers on keyboard.	1. ENTER light comes on.	Replace MLV.
	2. [2] MLV displays ENTER ELINT IDΔ XXX.	
(12) Press ENTER.	1. ENTER light goes off.	Replace MLV.
	2. MLV displays SEARCHING TAPE, LOADING CLC, READING FILE RECORD, VERIFYING, then LOAD ELINT DONE.	1. Replace tape transport unit. 2. If program still does not load, replace MLV.
	3. ENTER light comes on.	

Table 6. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(13) Press ENTER.	1. ENTER light goes off. 2. [1] MLV displays MODE CLC XXXXXX.	Replace MLV.
l. Verify PGM file by doing substeps below on MLV:		
(1) Press V then P on the keyboard.	1. ENTER light comes on. 2. MLV display VERIFY PGM.	Replace MLV.
(2) Press ENTER.	1. ENTER light goes off. 2. MLV displays ENTER PGM IDΔ.	Replace MLV.
(3) Enter PGM file PID to be verified.	1. ENTER light comes on. 2. [2] MLV displays ENTER PGM IDΔ XXX.	Replace MLV.
(4) Press ENTER.	1. ENTER light goes off. 2. MLV displays SEARCHING TAPE, READING FILE RECORD, VERIFYING, VERIFY PGM DONE. 3. ENTER light comes on.	Replace MLV.
(5) Press ENTER.	1. ENTER light goes off. 2. [1] MLV displays MODE CLC XXXXXX.	Replace MLV.
m. Verify ELINT file by doing substeps below on MLV:		
(1) Press V then E on the keyboard.	1. ENTER light comes on. 2. MLV displays VERIFY ELINT.	Replace MLV.
(2) Press ENTER.	1. ENTER light goes off. 2. MLV displays ENTER ELINT IDΔ.	Replace MLV.

Table 6. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Enter ELINT file PID to be verified.	1. ENTER light comes on. 2. [2] MLV displays ENTER ELINT IDΔ XXX.	Replace MLV.
(4) Press ENTER.	1. ENTER light goes off. 2. MLV displays SEARCHING TAPE, READING FILE, RECORD, VERIFYING, VERIFY ELINT DONE. 3. ENTER light comes on.	Replace MLV.
(5) Press ENTER.	1. ENTER light goes off. 2. [1] MLV displays MODE CLC XXXXXX.	Replace MLV.
4. TURN OFF. a. On GND PWR control panel assembly, set 3 switch to AUTO.		
CAUTION Never set MLV power switch to OFF while any of the following is displayed: SEARCHING TAPE, LOADING CLC, READING FILE RECORD, VERIFYING.		
b. On MLV, set POWER switch to OFF.		
c. If OFP is to be loaded into MC1, MC2, SMS Boot, SMS, or SDC, do substeps below:		
(1) If MC1, do table 1, steps 2b thru 3d.		
(2) If MC2, do table 2, steps 2b thru 3c.		
(3) If SMS Boot, do table 3, steps 2b thru 3c.		

Table 6. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(4) If SMS, do table 4, steps 2b thru 3c. (5) If SDC, do table 5, steps 2b thru 3c. d. Remove electrical power (A1-F18AC-LMM-000). e. Do table 4, WP004 00, to verify correct program identification. f. On MLV do substeps below: (1) Disconnect W3P1 from CMPTR connector (J2). (2) Disconnect W1P2 from POWER connector (J1). g. In aircraft nose wheelwell: (1) Disconnect W3P2 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

[1] XXXXXX is release date of the program on tape transport unit.
[2] XXX is entered PID number.

Table 7. MLV Displays BUS ERROR

Procedure	Normal Indication	Remedy for Abnormal Indication
CAUTION Never set MLV power switch to OFF while any of the following is displayed: SEARCHING TAPE, LOADING CLC, READING FILE RECORD, VERIFYING.		
1. On MLV press ENTER. 2. On MLV, set POWER switch to OFF. 3. On GND PWR control panel assembly, set 3 switch to AUTO. 4. Wait at least 30 seconds.		
5. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switches remain on (latched).	1. If switches unlatch in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switches will not remain on, troubleshoot (A1-F18AC-FIM-000, WP012 00).
6. Do Table 6 step 3j.		

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

AVIONICS LOAD/VERIFICATION PROCEDURES USING AN/ASM-687 LOADER-VERIFIER TEST SET

EFFECTIVITY: F/A-18C AND F/A-18D

Reference Material

Flight Incident Recorder and Monitoring System	A1-F18AE-580-300
Memory Unit MU-860/ASQ-194	WP005 00
Line Maintenance Procedures	A1-F18AC-LMM-000
Multipurpose Display Group	A1-F18AC-745-200
Displays Test F/A-18C	WP004 00
Displays Test F/A-18D	WP005 00
Multipurpose Display Group	A1-F18AG-745-200
Displays Test F/A-18C	WP004 00
Displays Test F/A-18D	WP005 00
Software Configuration Manual	A1-F18AC-SCM-000
Program Load Versions	WP003 00
Program Load CONFIG/IDENT Verification	WP004 00
Component Locator	WP005 00
Test Equipment Hookup Locator	WP007 00

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Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 126	-	Deployable Flight Incident Recorder Set (ECP MDA-F/A-18-00321R1C1)	15 Feb 93	-
F/A-18 AFC 211	-	AN/APG-65, Replacement With AN/APG-73 (ECP MDA-F/A-18-00508)	1 Jul 96	-
F/A-18 AFC 292	-	U.S. Marine Corps Reserves A+ Avionics Upgrade, Incorporation of (ECP MDA-F/A-18-0583)	15 Oct 00	-

1. INTRODUCTION.

2. This work package includes procedures for doing the Advanced Memory Loader-Verifier Test Set AN/ASM-687 (AMLV) self test and for using the AMLV to load operational flight programs (OFP) into the following weapon replaceable assemblies (WRA):

a. Digital Data Computer No. 1 and No. 2 (MC1, MC2)

b. Armament Computer (SMS)

c. Signal Data Computer CP-1726/ASQ-194 (SDC)

d. Computer-Power Supply CP-1325/APG-65 (CPS), 163427 THRU 164279; ALSO 164627 THRU 164897 BEFORE F/A-18 AFC-211

e. Radar Data Processor CP-2062/APG-73, 164898 AND UP; ALSO 164627 THRU 164987 AFTER F/A-18 AFC-211 AND F/A-18A 162394 THRU 163175 AFTER F/A-18 AFC 292

f. Air Data Computer CP-1334( )/A (ADC)

g. Control-Converter C-10382/A (CSC)

h. Command Launch Computer CP-1001( ) /AWG (CLC)

i. Digital Map Computer CP-1802/ASQ-196 (DMC)

j. Data Transfer Interface Unit J-6008/A (DFIRS), 164725 AND UP; ALSO 164627 THRU 164724 AFTER F/A-18 AFC 126

Table 1. MC1 Load/Verification Procedure For CP-1699A/AYK-14(V) (XN-6) Computer

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Digital Data Computer No. 1 (MC1)

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/ASM-687	Advanced Memory Loader-Verifier Test Set

Materials Required

None

NOTE

For Component Locator, refer to WP005 00.

For Test Equipment Hookup, refer to WP007 00.

Table 1. MC1 Load/Verification Procedure For CP-1699A/AYK-14(V) (XN-6) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W2P1 to CMPTR connector (J2). (3) Connect W2P2 to AGE/232 connector (J3). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P2 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). c. Remove Memory Unit MU-806/ASQ-194 (A1-F18AE-580-300, WP005 00) and remove data stored. 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

Table 1. MC1 Load/Verification Procedure For CP-1699A/AYK-14(V) (XN-6) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
CAUTION To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.		
c. On AMLV do substeps below:		
(1) Set POWER switch to ON.	1 AMLV displays operating placards with flag rotating in the farthest right cell symbol of AMLV display, then mode F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(2) Enter autoload command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	AMLV displays TTC slot and the program identification (PID) number of the first OFP on the tape transport followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine MC1 OFP PID number and press R on the keyboard.	1 AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.

Table 1. MC1 Load/Verification Procedure For CP-1699A/AYK-14(V) (XN-6) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(5) Enter autoload command by pressing A and then L.	1. AMLV displays AL Δ . 2. Enter light comes on.	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification number.		
(6) Enter MC1 OFP PID number by pressing applicable numbers on the keyboard.	1. ENTER light goes off. 2. [2] AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
(7) Press ENTER.	1. ENTER light goes off. 2. AMLV displays OPENING FILE, WAITING, then POWER UP MC1. 3. ENTER light comes on.	Do AMLV self test procedure, table 7.
d. On MC/HYD ISOL control panel assembly, hold MC switch to 2 OFF position.		
e. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
f. On MC/HYD ISOL control panel assembly release 2 OFF switch.		
g. On AMLV do substeps below:		
(1) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.

Table 1. MC1 Load/Verification Procedure For CP-1699A/AYK-14(V) (XN-6) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
	2. 2 ▸ AMLV displays operating placards with rotating flag in farthest right cell symbol, then XXXXXX DONE message.	1. Replace tape transport cartridge. 2. If program still does not load replace AMLV.
	3. ENTER light comes on.	Do AMLV self test procedure, table 7.
(2) Press ENTER.	1. ENTER light goes off. 2. 1 ▸ AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
3. TURN OFF. a. If system OFP loading is complete, on AMLV, set POWER switch to OFF.		
NOTE Placing the MC/HYD ISOL switch to the center position before 1 switch on GND PWR control panel assembly is set to AUTO may corrupt MC2 OFP.		
b. On GND PWR control panel assembly, set 1 switch to AUTO. c. Replace Memory Unit MU-806/ASQ-194 (A1-F18AE-580-300, WP005 00). d. Do applicable table, WP004 00, to verify correct program identification.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
e. Remove electrical power (A1-F18AC-LMM-000).		

Table 1. MC1 Load/Verification Procedure For CP-1699A/AYK-14(V) (XN-6) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
f. On AMLV do substeps below: (1) Disconnect W2P2 from AGE/232 connector (J3). (2) Disconnect W2P1 from CMPTR connector (J2). (3) Disconnect W1P2 from POWER connector (J1). g. In aircraft nose wheelwell do substeps below: (1) Disconnect W2P3 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089). h. Do displays test listed below: ON F/A-18C 161353 THRU 163782, A1-F18AC-745-200, WP004 00. ON F/A-18D 161353 THRU 163782, A1-F18AC-745-200, WP005 00. ON F/A-18C 163985 AND UP , A1-F18AG-745-200, WP004 00. ON F/A-18D 163985 AND UP , A1-F18AG-745-200, WP005 00.		

Table 1. MC1 Load/Verification Procedure For CP-1699A/AYK-14(V) (XN-6) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
i. Enter, as required, stored data variation into mission computer memory (A1-F18AC-LMM-000, WP043 00).		

LEGEND

[1] MMDDYY is release date of the program in tape transport cartridge.

[2] XXXXXX is entered PID number.

Table 2. MC2 Load/Verification Procedure For CP-1699A/AYK-14(V) (XN-6) Computer

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Digital Data Computer No. 2 (MC2) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-687 — Advanced Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W2P1 to CMPTR connector (J2).		

Table 2. MC2 Load/Verification Procedure For CP-1699A/AYK-14(V) (XN-6) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Connect W2P2 to AGE/232 connector (J3). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
CAUTION To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.		
c. On AMLV do substeps below: (1) Set POWER switch to ON.	1 ▶ AMLV displays operating placards with flag rotating in the farthest right cell symbol of AMLV display, then mode F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.

Table 2. MC2 Load/Verification Procedure For CP-1699A/AYK-14(V) (XN-6) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Enter autoload command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	AMLV displays TTC slot and the program identification (PID) number of the first OFP on the tape transport followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine MC2 OFP PID number and press R on the keyboard.	1 AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(5) Enter autoload command by pressing A and then L.	1. AMLV displays AL Δ . 2. Enter light comes on.	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification number.		
(6) Enter MC2 PID number by pressing applicable numbers on the keyboard.	1. ENTER light goes off. 2. 2 AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
(7) Press ENTER.	1. ENTER light goes off. 2. AMLV displays OPENING FILE, WAITING, then POWER UP MC1. 3. ENTER light comes on.	Do AMLV self test procedure, table 7.
d. On MC/HYD ISOL control panel assembly, hold MC switch to 1 OFF position.		

Table 2. MC2 Load/Verification Procedure For CP-1699A/AYK-14(V) (XN-6) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
e. On GND PWR control panel assembly set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
f. On MC/HYD ISOL control panel assembly release 1 OFF switch.		
g. On AMLV do substeps below:		
(1) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. 2 AMLV displays operating placards with rotating flag in farthest right cell symbol, then XXXXXX DONE message.	1. Replace tape transport cartridge. 2. If program still does not load replace AMLV.
	3. ENTER light comes on.	Do AMLV self test procedure, table 7.
(2) Press ENTER.	1. ENTER light goes off. 2. 1 AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
3. TURN OFF.		
a. If system OFP loading is complete, on AMLV, set POWER switch to OFF.		

NOTE

Placing the MC/HYD ISOL switch to the center position before 1 switch on GND PWR control panel assembly is set to AUTO may corrupt MC1 OFP.

b. On GND PWR control panel assembly, set 1 switch to AUTO.		

Table 2. MC2 Load/Verification Procedure For CP-1699A/AYK-14(V) (XN-6) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. Replace Memory Unit MU-806/ASQ-194 (A1-F18AE-580-300, WP005 00).		
d. Do applicable table, WP004 00, to verify correct program identification.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
e. Remove electrical power (A1-F18AC-LMM-000).		
f. On AMLV do substeps below:		
(1) Disconnect W2P2 from AGE/232 connector (J3).		
(2) Disconnect W2P1 from CMPTR connector (J2).		
(3) Disconnect W1P2 from POWER connector (J1).		
g. In aircraft nose wheelwell do substeps below:		
(1) Disconnect W2P3 from MUX test connector (83J-G003).		
(2) Disconnect W1P1 from utility power receptacle (1J-G089).		
h. Do displays test listed below:		

Table 2. MC2 Load/Verification Procedure For CP-1699A/AYK-14(V) (XN-6) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
ON F/A-18C 161353 THRU 163782, A1-F18AC-745-200, WP004 00. ON F/A-18D 161353 THRU 163782, A1-F18AC-745-200, WP005 00. ON F/A-18C 163985 AND UP , A1-F18AG-745-200, WP004 00. ON F/A-18D 163985 AND UP , A1-F18AG-745-200, WP005 00. i. Enter, as required, stored data variation into mission computer memory (A1-F18AC-LMM-000, WP043 00).		

LEGEND

[1] MMDDYY is release date of the program in tape transport cartridge.

[2] XXXXXX is entered PID number.

Table 3. SMS Boot Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Armament Computer **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-687 — Advanced Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W2P1 to CMPTR connector (J2).		

Table 3. SMS Boot Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Connect W2P2 to AGE/232 connector (J3). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
CAUTION To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.		
c. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).

Table 3. SMS Boot Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
d. On AMLV do substeps below:		
(1) Set POWER switch to ON.	1 AMLV displays operating messages with flag rotating in far right cell symbol of AMLV display, then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(2) Enter file open command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	AMLV displays TTC slot and program identification (PID) number of the first OFP on the tape transport cartridge followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine SMS Boot PID number and press R on the keyboard.	1 AMLV displays operating message(s) and then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(5) Enter autoload command by pressing A and then L on the keyboard.	1. AMLV displays AL ID Δ. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(6) Enter SMS Boot PID numbers by pressing applicable numbers on the keyboard.	1. ENTER light goes off. 2. 2 AMLV displays AL XXXXXX. 3. ENTER light comes on.	
(7) Press ENTER.	1. ENTER light goes off. 2. AMLV displays operating messages and then BOOT FOV- Sw Y/N+.?	Do AMLV self test procedure, table 7.

Table 3. SMS Boot Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE HARM target sequence/FLIR FOV/RAID must be held until the SMS boot loading is complete.		
e. On throttle grip, press and hold HARM target sequence/ FLIR FOV/RAID switch.		
f. On AMLV, press + on keyboard.	1. 2 AMLV displays operating messages with rotating flag in farthest right cell symbol, and a XXXXXX DONE message. 2. ENTER light comes on.	1. Replace tape transport cartridge. 2. If program still does not load, replace AMLV.
g. On throttle grip, release HARM target sequence/ FLIR FOV/RAID switch.		
h. If OFP is to be loaded into the SMS do substeps below:		
(1) On GND PWR control panel assembly set the 3 switch to AUTO and then back to B ON and hold for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
(2) Do table 4, steps 2b thru 3c.		
3. TURN OFF.		
a. On AMLV, set POWER switch to OFF.		
b. Remove electrical power (A1-F18AC-LMM-000).		

Table 3. SMS Boot Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. Do applicable table, WP004 00, to verify correct program identification. d. On AMLV do substeps below: (1) Disconnect W2P2 from AGE/232 connector (J3). (2) Disconnect W2P1 from CMPTR connector (J2). (3) Disconnect W1P2 from POWER connector (J1). e. In aircraft nose wheelwell do substeps below: (1) Disconnect W3P2 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

[1] MMDDYY is release date of program on tape transport cartridge.

[2] XXXXXX is entered PID number.

Table 4. SMS Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Armament Computer CP-1342/AYQ-9(V) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-687 — Advanced Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W2P1 to CMPTR connector (J2). (3) Connect W2P2 to AGE/232 connector (J3).		

Table 4. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W3P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
	CAUTION To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.	
c. On AMLV do substeps below:		
(1) Set POWER switch to ON.	1 AMLV displays operating messages with flag rotating in far right cell symbol of AMLV display, then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(2) Enter file open command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.

Table 4. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Press ENTER.	AMLV displays TTC slot and program identification (PID) number of the first OFP on the tape transport cartridge followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine SMS PID number and press R on the keyboard.	1 AMLV displays operating messages, then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(5) Enter auto load command by pressing A and then L on the keyboard.	1. AMLV displays AL ID Δ. 2. ENTER light goes off.	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification (PID) number.		
(6) Enter SMS PID number by pressing applicable numbers on the keyboard.	1. ENTER light comes on. 2. 2 AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
d. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On AMLV do substeps below:		
(1) Press ENTER.	1. ENTER light goes off. 2. AMLV displays OPENING FILE, then BOOT FOV- SW Y/N +.	Do AMLV self test procedure, table 7.

Table 4. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Press + on keyboard.	1. 2 AMLV displays operating messages with rotating flag in farthest right cell symbol, the a XXXXXX DONE message. 2. ENTER light comes on.	1. Replace tape transport cartridge. 2. If program still does not load, replace AMLV.
(3) Press ENTER.	1. ENTER light goes off. 2. 1 AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
3. TURN OFF. a. On GND PWR control panel assembly, set 3 switch to AUTO. b. On AMLV, set POWER switch to OFF. c. Do applicable table, WP004 00, to verify correct program identification. d. Remove electrical power (A1-F18AC-LMM-000). e. On AMLV do substeps below: (1) Disconnect W2P2 from AGE/232 connector (J3). (2) Disconnect W2P1 from CMPTR connector (J2). (3) Disconnect W1P2 from POWER connector (J1).		

Table 4. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
f. In aircraft nose wheelwell do substeps below: (1) Disconnect W2P3 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

1 ▸ MMDDYY is release date of program on tape transport cartridge.

2 ▸ XXXXXX is entered PID number.

Table 5. SDC Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Signal Data Computer CP-1726/ASQ-194 (SDC) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-687 — Advanced Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W2P1 to CMPTR connector (J2).		

Table 5. SDC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Connect W2P2 to AGE/232 connector (J3). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000).		
CAUTION To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.		
b. On GND PWR control panel assembly, set EXT PWR switch to RESET and hold for 3 seconds. c. On AMLV do substeps below:		
(1) Set POWER switch to ON.	1 AMLV displays operating messages with flag rotating in far right cell symbol of AMLV display, then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(2) Enter file open command by pressing F then O on keyboard.	AMLV displays FO.	Do AMLV self test procedure, table 7.

Table 5. SDC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Press ENTER.	AMLV displays TTC slot and program identification (PID) number of the first OFP on the tape transport cartridge followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine SDC PID number and press R on the keyboard.	1 ▸ AMLV displays operating messages, then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(5) Enter auto load command by pressing A and then L on the keyboard.	AMLV displays AL ID Δ.	Do AMLV self test procedure, table 7.
(6) Enter SDC PID number from AMLV instruction decal by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. 2 ▸ AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
NOTE On Aircraft Maintenance Indicator ID-2388/ASQ-194 (nose wheelwell) press and release CONSUMABLES CHECK switch when directed by AMLV display.		
(7) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. 2 ▸ AMLV displays operating messages and direction to press consumables check switch with a rotating flag in farthest right cell symbol, then a XXXXXX DONE message.	1. Replace tape transport unit. 2. If program still does not load, replace AMLV.
	3. ENTER light comes on.	

Table 5. SDC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(8) Press ENTER.	1. ENTER light goes off. 2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
NOTE If the SDC will not load OFP and indicates an absence of memory (no OFP) the FIRAMS/SDP circuit breaker 85CBC004 position D2 on RLY CB PNL NO. 8 (door 10L), must be cycled while consumable check switch is pressed and held. Repeat steps 2.c.5. thru 2.c.8.		
3. TURN OFF. a. On AMLV, set POWER switch to OFF. b. Enter date/time of day data (A1-F18AC-LMM-000, WP018 02). c. Do applicable table, WP004 00, to verify correct program identification. d. Remove electrical power (A1-F18AC-LMM-000). e. On AMLV do substeps below: (1) Disconnect W2P2 from AGE/232 connector (J3). (2) Disconnect W2P1 from CMPTR connector (J2). (3) Disconnect W1P2 from POWER connector (J1). f. In aircraft nose wheelwell do substeps below: (1) Disconnect W2P3 from MUX test connector (83J-G003).		

Table 5. SDC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

1 MMDDYY is release date of the program on tape transport cartridge.

2 XXXXXX is entered PID number.

Table 6. RADAR Load/Verification Procedure WITH COMPUTER POWER SUPPLY CP-1325/APG-65 PART NUMBER 3525681-150 AND UP

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Computer- Power Supply CP-1325/APG-65 (CPS)

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/ASM-687	Advanced Memory Loader-Verifier Test Set

Materials Required

None

NOTE

For Component Locator, refer to WP005 00.

For Test Equipment Hookup, refer to WP007 00.

ON COMPUTER POWER SUPPLY CP-1325/APG-65 PART NUMBER 3525681-150 THRU 3525681-155 , this procedure may not be used to do the initial load of CONFIG/IDENT 89X software. The initial load must be done at intermediate shop level.
ON COMPUTER POWER SUPPLY CP-1325/APG-65 PART NUMBER 3525681-155 AND UP , this procedure is used to do the load of CONFIG/IDENT 91C AND UP software.

1. PRELIMINARY.

CAUTION

To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.

Table 6. RADAR Load/Verification Procedure WITH COMPUTER POWER SUPPLY CP-1325/APG-65 PART NUMBER 3525681-150 AND UP (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
a. On AMLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W2P1 to CMPTR connector (J2). (3) Connect W2P2 to AGE/232 connector (J3). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W3P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.

Table 6. RADAR Load/Verification Procedure WITH COMPUTER POWER SUPPLY CP-1325/APG-65 PART NUMBER 3525681-150 AND UP (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The GND PWR control panel assembly switches 1, 3, and 4 must be set to AUTO when loading the Radar OFP.		
c. On GND PWR control panel assembly, set and hold 2 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
d. On SNSR pod control box panel assembly, set RADAR switch to STBY.		
e. On AMLV do substeps below:		
(1) Set POWER switch to ON.	1 AMLV displays operating messages with flag rotating in far right cell symbol of AMLV display, then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(2) Enter autoload command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	AMLV displays TTC slot and program identification (PID) number of the first OFP on the tape transport cartridge followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine RDR PID number and press R on the keyboard.	1 AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.

Table 6. RADAR Load/Verification Procedure WITH COMPUTER POWER SUPPLY CP-1325/APG-65 PART NUMBER 3525681-150 AND UP (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(5) Enter auto load command by pressing A then L on the keyboard.	AMLV displays AL ID Δ.	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification (PID) number.		
(6) Enter RADAR PID number by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. [2] AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
(7) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [2] AMLV displays operating messages with a rotating flag in farthest right cell symbol then a XXXXXX DONE message. 3. ENTER light comes on.	1. Replace tape transport cartridge. 2. If program still does not load, replace AMLV.
(8) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	
f. On SNSR pod control box panel assembly, set RADAR switch to OFF.		
3. TURN OFF.		
a. On AMLV, set POWER switch to OFF.		
b. On GND PWR control panel assembly, set 2 switch to AUTO.		
c. Do applicable table, WP004 00, to verify correct program identification.		

Table 6. RADAR Load/Verification Procedure WITH COMPUTER POWER SUPPLY CP-1325/APG-65 PART NUMBER 3525681-150 AND UP (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
d. Remove electrical power (A1-F18AC-LMM-000). e. On AMLV do substeps below: (1) Disconnect W2P2 from AGE/232 connector (J2). (2) Disconnect W2P1 from CMPTR connector (J2). (3) Disconnect W1P2 from POWER connector (J1). f. In aircraft nose wheelwell do substeps below: (1) Disconnect W2P3 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

[1] MMDDYY is release date of the program in tape transport cartridge.

[2] XXXXXX is entered PID number.

Table 7. AMLV Self Test Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components None **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-687 — Advanced Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W2P1 to CMPTR connector (J2).		

Table 7. AMLV Self Test Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Connect W2P2 to AGE/ 232 connector (J3). (4) Connect W2P3 to AMLV junction box connector (J1) located in the AMLV case lid. b. In aircraft nose wheelwell, connect W1P1 to utility power receptacle (1J-G089). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
CAUTION To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.		
c. On AMLV do substeps below:		
(1) Set power switch to ON.	1 AMLV displays BOOTUP-AMLV, READING DIRECTORY, AMLV BIT TEST, REWINDING TTC with flag rotating in far right cell symbol of AMLV display, then MODE F/A-18 MMDDYY.	Replace AMLV.

Table 7. AMLV Self Test Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Enter self test command by pressing S then T on keyboard.	1. Enter light comes on. 2. AMLV displays AMLV SELF TEST.	Replace AMLV.
(3) Press ENTER.	1. Enter light goes off. 2. AMLV displays TESTING MCM INTRC, TESTING, REWINDING TTC then SELF TEST PASS (flag rotates in far right cell symbol of AMLV display during test). 3. ENTER light comes on.	Replace AMLV.
(4) Press ENTER.	1. ENTER light goes off. 2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	Replace AMLV.
(5) Enter lamp test command by pressing L then T on keyboard.	1. ENTER light comes on. 2. AMLV displays LMP/ KEYBOARD TEST.	Replace AMLV.
(6) Press ENTER.	AMLV displays first 5 positions with all segments lighted.	Replace AMLV.
(7) Observe AMLV display and press ENTER.	AMLV TEST display moves right for each press until last segments is lighted.	Replace AMLV.
(8) Observe AMLV display and press ENTER.	AMLV TEST display shows E in first position and moves right one position for each press until at right position.	Replace AMLV.
(9) Press ENTER.	[1] AMLV displays MODE F/A-18 MMDDYY.	Replace AMLV.
3. TURN OFF. a. On AMLV, set power switch to OFF. b. Remove electrical power (A1-F18AC-LMM-000).		

Table 7. AMLV Self Test Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On AMLV do substeps below: (1) Disconnect W2P3 from AMLV junction box connector (J1) located in the AMLV case lid. (2) Disconnect W2P2 from AGE/232 connector (J3). (3) Disconnect W2P1 from CMPTR connector (J2). (4) Disconnect W1P2 from POWER connector (J1). d. On aircraft nose wheelwell, disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

[1] XXXXXX is release date of the program in tape transport cartridge.

Table 8. ADC Load/Verification Procedure WITH AIR DATA COMPUTER CP-1334( )/A PART NUMBER 4031000-920 AND UP

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Air Data Computer CP-1334( )/A (ADC) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-687 — Advanced Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W2P1 to CMPTR connector (J2).		

Table 8. ADC Load/Verification Procedure WITH AIR DATA COMPUTER CP-1334( )/A PART NUMBER 4031000-920 AND UP (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Connect W2P2 to AGE/ 232 connector (J3). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

CAUTION

To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.

NOTE

Both Digital Data Computer No. 1 and No. 2 (MC1, MC2) must be OFF (GND PWR control panel assembly switch 1 must be set to AUTO) when loading the ADC OFP.

Table 8. ADC Load/Verification Procedure WITH AIR DATA COMPUTER CP-1334( )/A PART NUMBER 4031000-920 AND UP (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On AMLV do substeps below:		
(1) Set POWER switch to ON.	1 ▸ AMLV displays operating messages with flag rotating in far right cell symbol of AMLV display, then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(2) Enter autoload command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	AMLV displays TTC slot and program identification (PID) number of the first OFP on the tape transport cartridge followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine ADC OFP PID number and press R on the keyboard.	1 ▸ AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(5) Enter auto load command by pressing A then L on the keyboard.	AMLV displays AL ID Δ.	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification (PID) number.		
(6) Enter ADC OFP PID number by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. 2 ▸ AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
d. On GND PWR control panel assembly, set and hold 2 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000).

Table 8. ADC Load/Verification Procedure WITH AIR DATA COMPUTER CP-1334( )/A PART NUMBER 4031000-920 AND UP (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
		2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On AMLV do substeps below:		
(1) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [2] AMLV displays operating messages with a rotating flag in farthest right cell symbol then a XXXXXX DONE message.	1. Replace tape transport cartridge. 2. If program still does not load, replace AMLV.
	3. ENTER light comes on.	
(2) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	
f. Verify program load by doing substeps below on AMLV:		
(1) Enter auto verify command by pressing A then V on the keyboard.	AMLV displays AV ID Δ.	Do AMLV self test procedure, table 7.
(2) Enter ADC PID number from AMLV instruction decal by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. [2] AMLV displays AV Δ XXXXXX.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [2] AMLV displays operating messages with a rotating flag in farthest right cell symbol then a XXXXXX DONE message.	1. Replace tape transport cartridge and do table 8. 2. If program still does not verify, replace AMLV.
(4) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.

Table 8. ADC Load/Verification Procedure WITH AIR DATA COMPUTER CP-1334( )/A PART NUMBER 4031000-920 AND UP (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
	2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	
3. TURN OFF. a. On AMLV, set POWER switch to OFF. b. On GND PWR control panel assembly, set 2 switch to AUTO. c. Do applicable table, WP004 00, to verify correct program identification. d. Remove electrical power (A1-F18AC-LMM-000). e. On AMLV do substeps below: (1) Disconnect W2P2 from AGE/232 connector (J2). (2) Disconnect W2P1 from CMPTR connector (J2). (3) Disconnect W1P2 from POWER connector (J1). f. In aircraft nose wheelwell do substeps below: (1) Disconnect W2P3 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

[1] MMDDYY is release date of the program in tape transport cartridge.

[2] XXXXXX is entered PID number.

Table 9. CSC Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Control-Converter C-10382/A (CSC) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-687 — Advanced Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV do substeps below: (1) Connect W1P2 to POWER connector (J1). (2) Connect W2P1 to CMPTR connector (J2).		

Table 9. CSC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Connect W2P2 to AGE/ 232 connector (J3). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

CAUTION

To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.

NOTE

Both Digital Data Computer No. 1 and No. 2 (MC1, MC2) must be OFF (GND PWR control panel assembly switch 1 must be set to AUTO) when loading the ADC OFP.

Table 9. CSC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On AMLV do substeps below:		
(1) Set POWER switch to ON.	1 AMLV displays operating messages with flag rotating in far right cell symbol of AMLV display, then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(2) Enter file open command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	AMLV displays TTC slot and program identification (PID) number of the first OFP on the tape transport cartridge followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine CSC OFP PID number and press R on the keyboard.	1 AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(5) Enter auto load command by pressing A then L on the keyboard.	AMLV displays AL ID Δ.	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification (PID) number.		
(6) Enter CSC OFP PID number by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. 2 AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
d. On GND PWR control panel assembly, set and hold 2 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000).

Table 9. CSC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
		2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On AMLV do substeps below:		
(1) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [2] AMLV displays operating messages with a rotating flag in farthest right cell symbol then a XXXXXX DONE message.	1. Replace tape transport cartridge. 2. If program still does not load, replace AMLV.
	3. ENTER light comes on.	
(2) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	
f. Verify program load by doing substeps below on AMLV:		
(1) Enter auto verify command by pressing A then V on the keyboard.	AMLV displays AV ID Δ.	Do AMLV self test procedure, table 7.
(2) Enter CSC OFP PID number by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. [2] AMLV displays AV Δ XXXXXX.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [2] AMLV displays operating messages with a rotating flag in farthest right cell symbol then a XXXXXX DONE message.	1. Replace tape transport cartridge and do table 8. 2. If program still does not verify, replace AMLV.
(4) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.

Table 9. CSC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
	2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	
3. TURN OFF. a. On AMLV, set POWER switch to OFF. b. On GND PWR control panel assembly, set 2 switch to AUTO. c. Do applicable table, WP004 00, to verify correct program identification. d. Remove electrical power (A1-F18AC-LMM-000). e. On AMLV do substeps below: (1) Disconnect W2P2 from AGE/232 connector (J2). (2) Disconnect W2P1 from CMPTR connector (J2). (3) Disconnect W1P2 from POWER connector (J1). f. In aircraft nose wheelwell do substeps below: (1) Disconnect W2P3 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

[1] MMDDYY is release date of the program in tape transport cartridge.

[2] XXXXXX is entered PID number.

Table 10. CLC Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Command Launch Computer CP-1001( )/AWG **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/ASM-687 — Advanced Memory Loader-Verifier Test Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. STORES SAFETY INSPECTION (A1-F18AE-LWS-000).		
WARNING To prevent injury or death of personnel, all live weapons and explosive cartridges must be removed from aircraft and gun must be safetied before doing this test.		
a. Make sure electrical power is off (A1-F18AC-LMM-000). b. Make sure all weapons are removed from aircraft.		

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. Make sure all explosive cartridges are removed from cartridge chambers for all Aircraft Bomb Ejection Racks BRU-32 ( ) installed on aircraft.		
d. Make sure all explosive cartridges are removed from cartridge chambers on Aircraft Bomb Ejector Racks BRU-33 ( ) if installed on aircraft.		
e. Make sure all explosive cartridges are removed from cartridge chambers on Aircraft Guided Missile Launcher LAU-116/A AIM-7 fuselage stations if installed on aircraft.		
f. Make sure all Aircraft Guided Missile Launcher LAU-116/A hooks are closed and SAFETY RELEASE knob is rotated clockwise.	SAFETY RELEASE INDICATOR shows GREEN - HOOKS LOCKED.	1.With hooks closed, rotate SAFETY RELEASE knob clockwise. 2. If knob will not rotate, replace Aircraft Guided Missile Launcher LAU-116/A (A1-F18AC-740-300, WP026 00).
g. Make sure all explosives are removed from breeches on Multiple Ejector Racks (MER) if installed on aircraft.		
h. Make sure gun electrical signal safety switch is set to safe (extended) position, aft of door 6.		
i. Make sure gun hold-back mechanism handle is set to cleared; gun hold-back handle indicator (extended).		
j. Close hooks on Bomb Ejector Racks BRU-32( ) for station used to ID HARM and set ground safety handle to LOCKED.		

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
2. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV do substeps below: (1) Connect W1P2 to AMLV POWER connector (J1). (2) Connect W2P1 to AMLV CMPTR connector (J2). (3) Connect W2P2 to AGE/232 connector (J3).		
b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003).		
3. PROCEDURE. a. Open door 14R (A1-F18AC-LMM-010).		
b. On Armament/Computer CP-1342/AYQ-9(V) do substeps below: (1) Set ARMAMENT switches to 64 for station used to ID HARM in step 1.j. (2) For remaining stations set switches to 00, except stations with tank installed set switches to 01.		

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. Apply electrical power (A1-F18AC-LMM-000).		
d. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. Connect ground intercommunications (A1-F18AC-LMM-000).		
f. On SNSR pod control box panel assembly, make sure RADAR switch is OFF.		
NOTE If a malfunction occurs during this test, make sure circuit breakers are closed; ON 163427 THRU 165206 (A1-F18AE-740-200, WP011 00) or ON 165207 AND UP (A1-F18AH-740-200, WP006 00).		
g. On MC/HYD ISOL control panel assembly, set MC switch to NORM.		
h. After 80 to 180 seconds, select A/G master mode button.	A/G master mode button lights.	Make sure enough time has elapsed for the SMP to complete self test, do steps 3c through 3h.
i. On flaps, landing gear and stores panel assembly, select station used to ID HARM.	Selected station light comes on.	ON 163427 THRU 165206 do table 1 (A1-F18AE-740-200, WP017 00). ON 165207 AND UP do table 1 (A1-F18AH-740-200, WP015 00).
j. On left hand vertical console control panel, move JETT select switch from SAFE to STORES.	In avionics bay door 13R, the CLC cooling fan comes on.	ON 163427 THRU 165206 replace CLC (A1-F18AE-740-300, WP011 00). ON 165207 AND UP replace CLC (A1-F18AH-740-300, WP008 00).

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
CAUTION To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.		
k. On AMLV, do substeps below:		
(1) Set power switch to ON.	1 AMLV displays BOOT UP - AMLV, READING DIRECTORY, AMLV BIT TEST, REWINDING TTC, then MODE F/A-18 MMDDYY.	Replace AMLV.
NOTE For an operational CLC both a HARM operational program (PGM) and Electronic Intelligence (ELINT) files must be loaded. PGM file must be loaded before ELINT file.		
(2) If PGM is to be loaded into CLC, do step 3.k.(3). If ELINT is to be loaded into CLC, do step 3.k.(9).		
(3) Enter load program command by pressing L then P on keyboard.	1. ENTER light comes on. 2. AMLV displays LOAD PGM.	Replace AMLV.
(4) Press ENTER.	1. ENTER light goes off. 2. AMLV displays ENTER PGM ID Δ.	Replace AMLV.
NOTE See WP003 00 for correct program part number and to verify program identification (PID) number.		

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(5) Enter PGM PID number from AMLV instruction decal by pressing applicable numbers on keyboard.	ENTER light comes on.	Replace AMLV.
(6) Press ENTER.	1. ENTER light goes off. 2. [2] AMLV displays OPENING FILE, LOADING XXXXXX indicating that file XXXXXX is being loaded into CLC, REPOSITIONING, VERIFYING, CLOSING FILE, then LOAD XXXXXX DONE. 3. ENTER light comes on.	Replace AMLV. 1. IF AMLV Display BUS ERROR 7021 do table 11, this WP. 2. Replace tape transport unit. 3. If program still does not load, replace AMLV.
(7) Press ENTER.	1. ENTER light goes off. 2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	Replace AMLV.
(8) If ELINT is to be loaded into CLC do step 3j9. If not go to step 4.		
NOTE PGM file must be loaded before ELINT file.		
(9) Enter load ELINT command by pressing L then E on keyboard.	1. ENTER light comes on. 2. AMLV displays LOAD ELINT.	Replace AMLV.
(10) Press ENTER.	1. ENTER light goes off. 2. AMLV displays ELINT ID Δ.	Replace AMLV.
NOTE See WP003 00 for correct program part number and to verify program identification (PID) number.		
(11) Enter ELINT file PID number from AMLV instruction decal by pressing applicable numbers on keyboard.	ENTER light comes on.	Replace AMLV.

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(12) Press ENTER.	1. ENTER light goes off.	Replace AMLV.
	2. [2] AMLV displays OPENING FILE, LOADING XXXXXX, indicating that file XXXXXX is being loaded in the CLC, REPOSITIONING, VERIFYING, CLOSING FILE, LOAD XXXXXX DONE.	1. Replace tape transport unit. 2. If program still does not load, replace AMLV.
	3. ENTER light comes on.	
(13) Press ENTER.	1. ENTER light goes off.	Replace AMLV.
	2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	
NOTE Files are automatically verified while being loaded. A file may be verified if it is suspected to be corrupted.		
l. If files are to be verified do the steps below on AMLV. If not go to step 4.		
(1) Press V then P on the keyboard.	1. ENTER light comes on.	Replace AMLV.
	2. AMLV display VERIFY PGM.	
(2) Press ENTER.	1. ENTER light goes off.	Replace AMLV.
	2. AMLV displays ENTER PGM IDΔ.	
(3) Enter PGM file PID to be verified.	1. ENTER light comes on.	Replace AMLV.
	2. [2] AMLV displays ENTER PGM IDΔ XXXXXX.	

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(4) Press ENTER.	1. ENTER light goes off. 2. [2] AMLV displays OPENING FILE, VERIFYING CLC, CLOSING FILE, VERIFYING XXXXXX DONE. 3. ENTER light comes on.	Replace AMLV.
(5) Press ENTER.	1. ENTER light goes off. 2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	Replace AMLV.
m. Verify ELINT file by doing substeps below on AMLV:		
(1) Press V then E on the keyboard.	1. ENTER light comes on. 2. AMLV displays VERIFY ELINT.	Replace AMLV.
(2) Press ENTER.	1. ENTER light goes off. 2. AMLV displays ENTER ELINT IDΔ.	Replace AMLV.
(3) Enter ELINT file PID to be verified.	1. ENTER light comes on. 2. [2] AMLV displays ENTER ELINT IDΔ XXXXXX.	Replace AMLV.
(4) Press ENTER.	1. ENTER light goes off. 2. [2] AMLV displays OPENING FILE, VERIFYING CLC, CLOSING FILE,VERIFY XXXXXX DONE. 3. ENTER light comes on.	Replace AMLV.
(5) Press ENTER.	1. ENTER light goes off. 2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	Replace AMLV.

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
4. TURN OFF. a. On GND PWR control panel assembly, set 3 switch to AUTO. b. On AMLV, set POWER switch to OFF. c. Do applicable table, WP004 00, to verify correct program identification. d. Remove electrical power (A1-F18AC-LMM-000). e. On AMLV do substeps below: (1) Disconnect W2P1 from CMPTR connector (J2). (2) Disconnect W1P2 from POWER connector (J1). f. In aircraft nose wheelwell do substeps below: (1) Disconnect W2P3 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

[1] MMDDYY is release date of the program on tape transport unit.
[2] XXXXXX is entered PID number.

Table 11. AMLV Displays BUS ERROR

Procedure	Normal Indication	Remedy for Abnormal Indication
CAUTION To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.		
1. On AMLV do substeps below: a. Press ENTER. b. Set POWER switch to OFF.		
2. On GND PWR control panel assembly: a. Set 3 switch to AUTO and wait at least 30 seconds.		
b. Set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test(A1-F18AC-420-200, WP006 00).
3. Do table 10 step 3j.		

Table 12. MC1 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Digital Data Computer No. 1 (MC1)

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/ASM-687	Advanced Memory Loader-Verifier Test Set

Materials Required

None

NOTE

For Component Locator, refer to WP005 00.

For Test Equipment Hookup, refer to WP007 00.

1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV do substeps below:		
(1) Connect W1P2 to POWER connector (J1).		
(2) Connect W2P1 to CMPTR connector (J2).		

Table 12. MC1 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Connect W2P2 to AGE/232 connector (J3). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). c. Remove Memory Unit MU-806/ASQ-194 (A1-F18AE-580-300, WP005 00) and remove data stored. 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

CAUTION

To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.

Table 12. MC1 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On AMLV do substeps below:		
(1) Set POWER switch to ON.	1 AMLV displays operating placards with flag rotating in the farthest right cell symbol of AMLV display, then mode F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(2) Enter file open command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	AMLV displays TTC slot and the PID number of the first OFP on the tape transport followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine MC1 boot load program identification (PID) number and press R on the keyboard.	1 AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
NOTE The XN-8 boot load program takes less than one second to transfer from the AMLV to the XN-8 computer.		
(5) Enter autoload command by pressing A and then L.	1. AMLV displays AL Δ . 2. Enter light comes on.	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification number.		

Table 12. MC1 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(6) Enter MC1 boot load PID number by pressing applicable numbers on the keyboard.	1. ENTER light goes off. 2. [2] AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
(7) Press ENTER.	1. ENTER light goes off. 2. AMLV displays OPENING FILE, WAITING, then POWER UP MC1. 3. ENTER light comes on.	Do AMLV self test procedure, table 7.
d. On MC/HYD ISOL control panel assembly, hold MC switch to 2 OFF position.		
e. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
f. On MC/HYD ISOL control panel assembly release MC 2 switch.		
g. On AMLV do substeps below:		
(1) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [2] AMLV displays operating placards with rotating flag in farthest right cell symbol, then LOAD XXXXXX DONE message.	1. Replace tape transport cartridge. 2. If program still does not load replace AMLV.
	3. ENTER light comes on.	Do AMLV self test procedure, table 7.

Table 12. MC1 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Press ENTER.	1. ENTER light goes off. 2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(3) Enter file open command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(4) Press ENTER.	AMLV displays TTC slot and the PID number of the first OFP on the tape transport followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(5) Determine MC1 OFP PID number and press R on the keyboard.	[1] AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification number.		
(6) Enter autoload command by pressing A and then L.	1. AMLV displays AL Δ. 2. Enter light comes on.	Do AMLV self test procedure, table 7.
(7) Enter MC1 OFP PID number by pressing applicable numbers on the keyboard.	1. ENTER light goes off. 2. [2] AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
(8) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.

Table 12. MC1 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
	2. [2] AMLV displays operating placards with rotating flag in farthest right cell symbol, then LOAD XXXXXX DONE message.	1. Replace tape transport cartridge. 2. If program still does not load replace AMLV.
	3. ENTER light comes on.	Do AMLV self test procedure, table 7.
(9) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	
3. TURN OFF. a. If system OFP loading is complete, on AMLV, set POWER switch to OFF.		

NOTE

Placing the MC/HYD ISOL switch to the center position before switch GND PWR 1 is set to AUTO may corrupt MC2 OFP.

b. On GND PWR control panel assembly, set 1 switch to AUTO. c. Replace Memory Unit MU-806/ASQ-194 (A1-F18AE-580-300, WP005 00). d. Do applicable table, WP004 00, to verify correct program identification.		

NOTE

Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.

e. Remove electrical power (A1-F18AC-LMM-000).		

Table 12. MC1 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
f. On AMLV do substeps below: (1) Disconnect W2P2 from AGE/232 connector (J3). (2) Disconnect W2P1 from CMPTR connector (J2). (3) Disconnect W1P2 from POWER connector (J1). g. In aircraft nose wheelwell do substeps below: (1) Disconnect W2P3 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089). h. Do displays test listed below: ON F/A-18C 161353 THRU 163782, A1-F18AC-745-200, WP004 00. ON F/A-18D 161353 THRU 163782, A1-F18AC-745-200, WP005 00. ON F/A-18C 163985 AND UP , A1-F18AG-745-200, WP004 00. ON F/A-18D 163985 AND UP , A1-F18AG-745-200, WP005 00. i. Enter, as required, stored data variation into mission computer memory (A1-F18AC-LMM-000, WP043 00).		

Table 12. MC1 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
LEGEND		
1 MMDDYY is release date of the program in tape transport cartridge.		
2 XXXXXX is entered PID number.		

Table 13. MC2 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Digital Data Computer No. 2 (MC2)

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/ASM-687	Advanced Memory Loader-Verifier Test Set

Materials Required

None

NOTE

For Component Locator, refer to WP005 00.

For Test Equipment Hookup, refer to WP007 00.

1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV do substeps below:		
(1) Connect W1P2 to POWER connector (J1).		
(2) Connect W2P1 to CMPTR connector (J2).		

Table 13. MC2 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Connect W2P2 to AGE/232 connector (J3). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

CAUTION

To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.

Table 13. MC2 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On AMLV do substeps below:		
(1) Set POWER switch to ON.	1 AMLV displays operating placards with flag rotating in the farthest right cell symbol of AMLV display, then mode F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(2) Enter file open command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	AMLV displays TTC slot and the PID number of the first OFP on the tape transport followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine MC2 boot load program identification (PID) number and press R on the keyboard.	1 AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
NOTE The XN-8 boot load program takes less than one second to transfer from the AMLV to the XN-8 computer.		
(5) Enter autoload command by pressing A and then L.	1. AMLV displays AL Δ. 2. Enter light comes on.	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification number.		

Table 13. MC2 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(6) Enter MC2 boot load PID number by pressing applicable numbers on the keyboard.	1. ENTER light goes off. 2. [2] AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
(7) Press ENTER.	1. ENTER light goes off. 2. AMLV displays OPENING FILE, WAITING, then POWER UP MC2. 3. ENTER light comes on.	Do AMLV self test procedure, table 7.
d. On MC/HYD ISOL control panel assembly, hold MC switch to 1 OFF position.		
e. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
f. On MC/HYD ISOL control panel assembly release MC switch.		
g. On AMLV do substeps below:		
(1) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [2] AMLV displays operating placards with rotating flag in farthest right cell symbol, then LOAD XXXXXX DONE message.	1. Replace tape transport cartridge. 2. If program still does not load replace AMLV.
	3. ENTER light comes on.	Do AMLV self test procedure, table 7.

Table 13. MC2 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Press ENTER.	1. ENTER light goes off. 2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(3) Enter file open command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(4) Press ENTER.	AMLV displays TTC slot and the PID number of the first OFP on the tape transport followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(5) Determine MC2 OFP PID number and press R on the keyboard.	[1] AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification number.		
(6) Enter autoload command by pressing A and then L.	1. AMLV displays AL Δ . 2. Enter light comes on.	Do AMLV self test procedure, table 7.
(7) Enter MC2 OFP PID number by pressing applicable numbers on the keyboard.	1. ENTER light goes off. 2. [2] AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
(8) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.

Table 13. MC2 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
	2. [2] AMLV displays operating placards with rotating flag in farthest right cell symbol, then LOAD XXXXXX DONE message. 3. ENTER light comes on.	1. Replace tape transport cartridge. 2. If program still does not load replace AMLV. Do AMLV self test procedure, table 7.
(9) Press ENTER.	1. ENTER light goes off. 2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
3. TURN OFF. a. If system OFP loading is complete, on AMLV, set POWER switch to OFF.		
NOTE Placing the MC/HYD ISOL switch to the center position before switch GND PWR 1 is set to AUTO may corrupt MC1 OFP.		
b. On GND PWR control panel assembly, set 1 switch to AUTO. c. Do applicable table, WP004 00, to verify correct program identification.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
d. Remove electrical power (A1-F18AC-LMM-000).		

Table 13. MC2 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
e. On AMLV do substeps below: (1) Disconnect W2P2 from AGE/232 connector (J3). (2) Disconnect W2P1 from CMPTR connector (J2). (3) Disconnect W1P2 from POWER connector (J1). f. In aircraft nose wheelwell do substeps below: (1) Disconnect W2P3 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089). g. Do displays test listed below: ON F/A-18C 161353 THRU 163782, A1-F18AC-745-200, WP004 00. ON F/A-18D 161353 THRU 163782, A1-F18AC-745-200, WP005 00. ON F/A-18C 163985 AND UP , A1-F18AG-745-200, WP004 00. ON F/A-18D 163985 AND UP , A1-F18AG-745-200, WP005 00. h. Enter, as required, stored data variation into mission computer memory (A1-F18AC-LMM-000, WP043 00).		

Table 13. MC2 Load/Verification Procedure For CP-2060( )/AYK-14(V) (XN-8) Computer (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
LEGEND		
1 MMDDYY is release date of the program in tape transport cartridge.		
2 XXXXXX is entered PID number.		

Table 14. DFIRS Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Data Transfer Interface Unit J-6008/A

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/ASM-687	Advanced Memory Loader-Verifier Test Set

Materials Required

None

NOTE

For Component Locator, refer to WP005 00.

For Test Equipment Hookup, refer to WP007 00.

1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV do substeps below:		
(1) Connect W1P2 to POWER connector (J1).		
(2) Connect W2P1 to CMPTR connector (J2).		

Table 14. DFIRS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Connect W2P2 to AGE/232 connector (J3). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). 2. PROCEDURE.		

NOTE

Both Digital Data Computers No. 1 and No. 2 (MC1, MC2) must be OFF when loading an OFP into DFIRS.

a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.

Table 14. DFIRS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On AMLV do substeps below:		
(1) Set POWER switch to ON.	1 AMLV displays operating placards with flag rotating in the farthest right cell symbol of AMLV display, then mode F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(2) Enter file open command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	AMLV displays TTC slot and the program identification (PID) number of the first OFP on the tape transport followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine DFIRS OFP PID number and press R on the keyboard.	1 AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(5) Enter autoload command by pressing A and then L.	1. AMLV displays AL Δ . 2. Enter light comes on.	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification number.		
(6) Enter DFIRS OFP load PID number by pressing applicable numbers on the keyboard.	1. ENTER light goes off. 2. 2 AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
d. On GND PWR control panel assembly, set and hold 3 switch to A ON position for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000).

Table 14. DFIRS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
		2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On AMLV do substeps below:		
(1) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. 2 AMLV displays operating placards with rotating flag in farthest right cell symbol, then LOAD XXXXXX DONE message.	1. Replace tape transport cartridge. 2. If program still does not load replace AMLV.
	3. ENTER light comes on.	Do AMLV self test procedure, table 7.
(2) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. 1 AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	
f. Verify program load by doing substeps below on AMLV:		
(1) Enter auto verify command by pressing A then V on the keyboard.	AMLV displays AV ID Δ.	Do AMLV self test procedure, table 7.
(2) Enter DFIRS PID number from step 2c4.	1. ENTER light comes on.	Do AMLV self test procedure, table 7.
	2. 2 AMLV displays AV Δ XXXXXX.	
(3) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. 1 AMLV displays operating messages with a rotating flag in farthest right cell symbol then a VERIFY XXXXXX DONE message.	1. Replace tape transport cartridge and do table 14. 2. If program still does not verify, replace AMLV.

Table 14. DFIRS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(4) Press ENTER.	1. ENTER light goes off. 2. [1] AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
3. TURN OFF. a. On AMLV, set POWER switch to OFF. b. On GND PWR control panel assembly, set 2 switch to AUTO. c. Do applicable table, WP004 00, to verify correct program identification.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
d. Remove electrical power (A1-F18AC-LMM-000). e. On AMLV do substeps below: (1) Disconnect W2P2 from AGE/232 connector (J2). (2) Disconnect W2P1 from CMPTR connector (J2). (3) Disconnect W1P2 from POWER connector (J1).		

Table 14. DFIRS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
f. In aircraft nose wheelwell do substeps below: (1) Disconnect W2P3 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

1 ▸ MMDDYY is release date of the program in tape transport cartridge.

2 ▸ XXXXXX is entered PID number.

Table 15. DMC Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Digital Map Computer CP-1802/ASQ-194

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/ASM-687	Advanced Memory Loader-Verifier Test Set

Materials Required

None

NOTE

For Component Locator, refer to WP005 00.

For Test Equipment Hookup, refer to WP007 00.

1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV do substeps below:		
(1) Connect W1P2 to POWER connector (J1).		
(2) Connect W2P1 to CMPTR connector (J2).		

Table 15. DMC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Connect W2P2 to AGE/232 connector (J3). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). 2. PROCEDURE.		
NOTE Both Digital Data Computers No. 1 and No. 2 (MC1, MC2) must be OFF when loading an OFP into DMC.		
a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
CAUTION To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.		

Table 15. DMC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On AMLV do substeps below:		
(1) Set POWER switch to ON.	1 AMLV displays operating placards with flag rotating in the farthest right cell symbol of AMLV display, then mode F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(2) Enter file open command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	AMLV displays TTC slot and the program identification (PID) number of the first OFP on the tape transport followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine DMC OFP PID number and press R on the keyboard.	1 AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(5) Enter autoload command by pressing A and then L.	1. AMLV displays AL Δ . 2. Enter light comes on.	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification number.		
(6) Enter DMC OFP load PID number by pressing applicable numbers on the keyboard.	1. ENTER light goes off. 2. 2 AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
d. On GND PWR control panel assembly, set and hold 2 switch to B ON position for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000).

Table 15. DMC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
		2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On AMLV do substeps below:		
(1) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2 [2] AMLV displays operating placards with rotating flag in farthest right cell symbol, then LOAD XXXXXX DONE message.	1. Replace tape transport cartridge. 2. If program still does not load replace AMLV.
	3. ENTER light comes on.	Do AMLV self test procedure, table 7.
(2) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	
f. Verify program load by doing substeps below on AMLV:		
(1) Enter auto verify command by pressing A then V on the keyboard.	AMLV displays AV ID Δ.	Do AMLV self test procedure, table 7.
(2) Enter DMC PID number from step 2c4.	1. ENTER light comes on.	Do AMLV self test procedure, table 7.
	2. [2] AMLV displays AV Δ XXXXXX.	
(3) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [1] AMLV displays operating messages with a rotating flag in farthest right cell symbol then a VERIFY XXXXXX DONE message.	1. Replace tape transport cartridge and do table 14. 2. If program still does not verify, replace AMLV.
(4) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.

Table 15. DMC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
	2. [1] AMLV displays MODE F/A-18 MMDDYY.	
3. TURN OFF. a. On AMLV, set POWER switch to OFF. b. On GND PWR control panel assembly, set 2 switch to AUTO. c. Do applicable table, WP004 00, to verify correct program identification.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
d. Remove electrical power (A1-F18AC-LMM-000). e. On AMLV do substeps below: (1) Disconnect W2P2 from AGE/232 connector (J2). (2) Disconnect W2P1 from CMPTR connector (J2). (3) Disconnect W1P2 from POWER connector (J1). f. In aircraft nose wheelwell do substeps below: (1) Disconnect W2P3 from MUX test connector (83J-G003).		

Table 15. DMC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

1 MMDDYY is release date of the program in tape transport cartridge.

2 XXXXXX is entered PID number.

Table 16. RADAR Load/Verification Procedure WITH RADAR DATA PROCESSOR CP-2062/APG-73

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Radar Data Processor CP-2062/APG-73

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/ASM-687	Advanced Memory Loader-Verifier Test Set

Materials Required

None

NOTE

For Component Locator, refer to WP005 00.

For Test Equipment Hookup, refer to WP007 00.

1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV do substeps below:		
(1) Connect W1P2 to POWER connector (J1).		
(2) Connect W2P1 to CMPTR connector (J2).		

Table 16. RADAR Load/Verification Procedure WITH RADAR DATA PROCESSOR CP-2062/APG-73 (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Connect W2P2 to AGE/232 connector (J3). b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

CAUTION

To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.

NOTE

The GND PWR control panel assembly switches 1, 3, and 4 must be set to AUTO when loading the Radar OFP.

c. On GND PWR control panel assembly, set and hold 2 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000).

Table 16. RADAR Load/Verification Procedure WITH RADAR DATA PROCESSOR CP-2062/APG-73 (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
		2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
d. On SNSR pod control box panel assembly, set RADAR switch to STBY.		
e. On AMLV do substeps below:		
(1) Set POWER switch to ON.	1 AMLV displays operating messages with flag rotating in far right cell symbol of AMLV display, then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(2) Enter file open command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	AMLV displays TTC slot and program identification (PID) number of the first OFP on the tape transport cartridge followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine RDR PID number and press R on the keyboard.	1 AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(5) Enter auto load command by pressing A then L on the keyboard.	AMLV displays AL ID Δ.	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification (PID) number.		

Table 16. RADAR Load/Verification Procedure WITH RADAR DATA PROCESSOR CP-2062/APG-73 (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(6) Enter RADAR PID number by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. [2] AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
(7) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [2] AMLV displays operating messages with a rotating flag in farthest right cell symbol then a XXXXXX DONE message. 3. ENTER light comes on.	1. Replace tape transport cartridge. 2. If program still does not load, replace AMLV.
(8) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	
f. On SNSR pod control box panel assembly, set RADAR switch to OFF.		
3. TURN OFF.		
a. On AMLV, set POWER switch to OFF.		
b. On GND PWR control panel assembly, set 2 switch to AUTO.		
c. Do applicable table, WP004 00, to verify correct program identification.		
d. Remove electrical power (A1-F18AC-LMM-000).		
e. On AMLV do substeps below:		
(1) Disconnect W2P2 from AGE/232 connector (J2).		

Table 16. RADAR Load/Verification Procedure WITH RADAR DATA PROCESSOR CP-2062/APG-73 (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Disconnect W2P1 from CMPTR connector (J2). (3) Disconnect W1P2 from POWER connector (J1). f. In aircraft nose wheelwell do substeps below: (1) Disconnect W2P3 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

1 ▸ MMDDYY is release date of the program in tape transport cartridge.

2 ▸ XXXXXX is entered PID number.

Table 17. FLIR Load/Verification Procedure WITH AN/AAS-38B CONTROLLER PROCESSOR

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Detecting Set AN/AAS-38B

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/ASM-687	Advanced Memory Loader-Verifier Test Set

Materials Required

None

NOTE

For Component Locator, refer to WP005 00.
For Test Equipment Hookup, refer to WP007 00.

1. PRELIMINARY.

CAUTION

To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.

a. On AMLV do substeps below:

(1) Connect W1P2 to POWER connector (J1).

(2) Connect W2P1 to CMPTR connector (J2).

(3) Connect W2P2 to AGE/232 connector (J3).

Table 17. FLIR Load/Verification Procedure WITH AN/AAS-38B CONTROLLER PROCESSOR (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. In aircraft nose wheelwell do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
CAUTION To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.		
c. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
d. Do initiated built-in test steps 1 through 2a (A1-F18AC-744-200, WP004 00).	FLIR pod is in STANDBY.	Do initiated built-in test (A1-F18AC-744-200, WP004 00).

Table 17. FLIR Load/Verification Procedure WITH AN/AAS-38B CONTROLLER PROCESSOR (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
e. On AMLV do substeps below:		
(1) Set POWER switch to ON.	1 AMLV displays operating messages with flag rotating in far right cell symbol of AMLV display, then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(2) Enter autoload command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	AMLV displays TTC slot and program identification (PID) number of the first OFP on the tape transport cartridge followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine FLIR OFP PID number and press R on the keyboard.	1 AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(5) Enter auto load command by pressing A then L on the keyboard.	AMLV displays AL ID> .	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification (PID) number.		
(6) Enter FLIR OFP PID number by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. 2 AMLV displays AL> XXXXXX.	Do AMLV self test procedure, table 7.

Table 17. FLIR Load/Verification Procedure WITH AN/AAS-38B CONTROLLER PROCESSOR (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(7) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [2] AMLV displays LOAD XXXXXX with a rotating flag in farthest right cell symbol then a XXXXXX DONE message.	1. Replace tape transport cartridge. 2. If program still does not load, replace AMLV.
	3. ENTER light comes on.	
(8) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	
f. Verify program load by doing substeps below on AMLV:		
(1) Enter auto verify command by pressing A then V on the keyboard.	AMLV displays AV ID>.	Do AMLV self test procedure, table 7.
(2) Enter FLIR PID number from AMLV instruction decal by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. [2] AMLV displays AV> XXXXXX.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [2] AMLV displays operating messages with a rotating flag in farthest right cell symbol then a XXXXXX DONE message.	1. Replace tape transport cartridge and do table 8. 2. If program still does not verify, replace AMLV.
(4) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	

Table 17. FLIR Load/Verification Procedure WITH AN/AAS-38B CONTROLLER PROCESSOR (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
g. Do initiated built-in test steps 3a and 3b (A1-F18AC-744-200, WP004 00). 3. TURN OFF. a. On AMLV, set POWER switch to OFF. b. On GND PWR control panel assembly, set 3 switch to AUTO. c. Do applicable table, WP004 00, to verify correct program identification. d. Remove electrical power (A1-F18AC-LMM-000). e. On AMLV do substeps below: (1) Disconnect W2P2 from AGE/232 connector (J3). (2) Disconnect W2P1 from CMPTR connector (J2). (3) Disconnect W1P2 from POWER connector (J1). f. In aircraft nose wheelwell do substeps below: (1) Disconnect W3P2 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

[1] MMDDYY is release date of the program in tape transport cartridge.

[2] XXXXXX is entered PID number.

Table 18. CIT Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Combined Interrogator Transponder AN/APX-111(V) (CIT)

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/ASM-687	Advanced Memory Loader-Verifier Test Set

Materials Required

None

NOTE

For Component Locator, refer to WP005 00.

For Test Equipment Hookup, refer to WP007 00.

1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting AMLV cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On AMLV, do substeps below:		
(1) Connect W1P2 to POWER connector (J1).		
(2) Connect W2P1 to CMPTR connector (J2).		

Table 18. CIT Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Connect W2P2 to AGE/ 232 connector (J3). b. In aircraft nose wheelwell, do substeps below: (1) Connect W1P1 to utility power receptacle (1J-G089). (2) Connect W2P3 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

CAUTION

To prevent damage to AMLV and tape transport cartridges (TTC)s do not set EXT PWR switch on GND PWR control panel assembly to OFF, NORM or RESET while AMLV is powered on and TTC is in motion. Do not set the AMLV POWER switch to OFF unless MODE F/A-18 MMDDYY (mmddyy= month-day-year) is displayed on AMLV, indicating that the TTC is not in motion. The TTC is in motion when the AMLV is displaying one of the following: BOOT-UP AMLV, SEARCHING, AMLV BIT TEST, CHANGING TTC'S, WAITING, LOADING, READING DIRECTORY, OPENING FILE, ABORTING, VERIFYING, TESTING, REPOSITIONING, REWINDING, or when the flag in the farthest right cell symbol is rotating.

NOTE

Both Digital Data Computer No. 1 and No. 2 (MC1, MC2) must be OFF (GND PWR control panel assembly switch 1 must be set to AUTO) when loading the ADC OFP.

Table 18. CIT Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On AMLV, do substeps below:		
(1) Set POWER switch to ON.	1 AMLV displays operating messages with flag rotating in far right cell symbol of AMLV display, then MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(2) Enter file open command by pressing F then O on keyboard.	1. AMLV displays FO. 2. ENTER light comes on.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	AMLV displays TTC slot and program identification (PID) number of the first OFP on the tape transport cartridge followed by a description of the associated processor.	Do AMLV self test procedure, table 7.
NOTE The 6 digit number after 1G or 2G (1G=TTC slot 1 and 2G=TTC slot 2) is the identification number of the first OFP residing in the TTC library. The remaining PID/processors can be displayed by pressing the + key to scroll forward. The - key is used to back up after a forward command.		
(4) Determine CIT OFP PID number and press R on the keyboard.	1 AMLV displays MODE F/A-18 MMDDYY.	Do AMLV self test procedure, table 7.
(5) Enter auto load command by pressing A then L on the keyboard.	AMLV displays AL ID Δ.	Do AMLV self test procedure, table 7.
NOTE See WP003 00 to verify program identification (PID) number.		
(6) Enter CIT OFP PID number by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. 2 AMLV displays AL Δ XXXXXX.	Do AMLV self test procedure, table 7.
d. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000).

Table 18. CIT Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
		2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On GND PWR control panel assembly, set and hold 2 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
f. On UFC, turn on IFF system. Allow 30 seconds for IFF to complete power on BIT.	Make sure IFF BIT status is GO on BIT display.	Do table 2, WP004 00.
g. On GND PWR control panel assembly, set 1 switch to AUTO.	Turns off MC's, and IFF system remains on.	
h. On AMLV, do substeps below:		
(1) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [2] AMLV displays operating messages with a rotating flag in farthest right cell symbol then a XXXXXX DONE message. 3. ENTER light comes on.	1. Replace tape transport cartridge. 2. If program still does not load, replace AMLV.
(2) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	
i. Verify program load by doing substeps below on AMLV:		
(1) Enter auto verify command by pressing A then V on the keyboard.	AMLV displays AV ID Δ.	Do AMLV self test procedure, table 7.

Table 18. CIT Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Enter CIT OFP PID number by pressing applicable numbers on keyboard.	1. ENTER light comes on. 2. [2] AMLV displays AV Δ XXXXXX.	Do AMLV self test procedure, table 7.
(3) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [2] AMLV displays operating messages with a rotating flag in farthest right cell symbol then a XXXXXX DONE message.	1. Replace tape transport cartridge and do table 8. 2. If program still does not verify, replace AMLV.
(4) Press ENTER.	1. ENTER light goes off.	Do AMLV self test procedure, table 7.
	2. [1] AMLV displays REWINDING TTC then MODE F/A-18 MMDDYY.	
3. TURN OFF.		
a. On AMLV, set POWER switch to OFF.		
b. On GND PWR control panel assembly, set 2 switch to AUTO.		
c. Do applicable table, WP004 00, to verify correct program identification.		
d. Remove electrical power (A1-F18AC-LMM-000).		
e. On AMLV, do substeps below:		
(1) Disconnect W2P2 from AGE/232 connector (J3).		
(2) Disconnect W2P1 from CMPTR connector (J2).		
(3) Disconnect W1P2 from POWER connector (J1).		

Table 18. CIT Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
f. In aircraft nose wheelwell, do substeps below: (1) Disconnect W2P3 from MUX test connector (83J-G003). (2) Disconnect W1P1 from utility power receptacle (1J-G089).		

LEGEND

1 MMDDYY is release date of the program in tape transport cartridge.

2 XXXXXX is entered PID number.

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

AVIONICS LOAD/VERIFICATION PROCEDURES USING AN/USQ-131 LOADER-VERIFIER SET

EFFECTIVITY: F/A-18C AND F/A-18D

Reference Material

Airborne Weapons/Stores Loading Manual	A1-F18AE-LWS-000
Line Maintenance Procedures	A1-F18AC-LMM-000
Multipurpose Display Group	A1-F18AC-745-200
Displays Test F/A-18C	WP004 00
Displays Test F/A-18D	WP005 00
Multipurpose Display Group	A1-F18AG-745-200
Displays Test F/A-18C	WP004 00
Displays Test F/A-18D	WP005 00
Software Configuration Manual	A1-F18AC-SCM-000
Program Load Versions	WP003 00
Program Load CONFIG/IDENT Verification	WP004 00
Component Locator	WP005 00
Test Equipment Hookup Locator	WP007 00
Memory Loader-Verifier Set AN/USQ-131	NAVAIR 16-30USQ131-1
Extended BIT	WP003 00

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Record of Applicable Technical Directives

Type/ Number	Date	Title and ECP No.	Date Incorp.	Remarks
F/A-18 AFC 126	-	Deployable Flight Incident Recorder Set (ECP MDA-F/A-18-00321R1C1)	15 Feb 93	-

1. INTRODUCTION.

2. This work package includes procedures for loading operational flight programs (OFP) using the Memory Loader-Verifier Set AN/USQ-131 (MLVS).

3. Each programmable WRA has a separate table. The WRAs include:.

a. Digital Data Computer No. 1 and No. 2 (MC1, MC2).

b. Armament Computer CP-1342/AYQ-9(V)/ CP-2218/AYK-22(V) (SMS).

c. Signal Data Computer CP-1726/ASQ-194 (SDC).

d. Radar Data Processor CP-2062/APG-73 (RDP).

e. Air Data Computer CP-1334A/A (ADC).

f. Control-Converter C-10382/A (CSC).

g. Digital Map Computer CP-1802/ASQ-196 (DMC).

h. ON 164725 AND UP; ALSO 164724 THRU 164724 AFTER F/A-18 AFC 126, Data Transfer Interface Unit J-6008/A, (DFIRS).

i. Command Launch Computer CP-1001( )/AWG (CLC).

j. Forward Looking Infrared Receiver AN/AAS-38B (FLIR)

k. Digital Communication System RT-1824(C)/ARC (DCS).

l. Enhanced Interference Blanker Unit MX-11741A (EIBU).

m. Computer Power Supply CP-1325/APG-65 (CPS).

Table 1. MC1 Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Digital Data Computer No. 1 (MC1) **Related Systems Required** Avionics Cooling System Electrical System		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
Support Equipment Required		
Part Number or Type Designation	**Nomenclature**	
AN/USQ-131	Memory Loader-Verifier Set	
Materials Required		
None		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003).		
2. PROCEDURE.		
a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
c. On MLVS, do substeps below:		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
d. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below: Turn On MC1	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
e. On MC/HYD ISOL control panel assembly, set MC switch to 2 OFF position and hold during step f.		
f. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN.		
NOTE Placing the MC/HYD ISOL switch to the center position before 1 switch on GND PWR control panel assembly is set to AUTO may corrupt MC2 OFP.		
a. On GND PWR control panel assembly, set 1 switch to AUTO. b. If system OFP loading is complete, on MLVS, set PWR switch to OFF.		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power-up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
f. Do displays test below: ON F/A-18C 163427 THRU 163782, A1-F18AC-745-200, WP004 00. ON F/A-18D 163434 THRU 163778, A1-F18AC-745-200, WP005 00. ON F/A-18C 163985 AND UP, A1-F18AG-745-200, WP004 00. ON F/A-18D 163986 AND UP, A1-F18AG-745-200, WP005 00. g. Enter stored data variation into mission computer memory as required.		

Table 2. MC2 Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Digital Data Computer No. 2 (MC2) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None		

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below:		
(1) Connect power cable connector P1 to connector J1.		
(2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below:		
(1) Connect power cable connector P2 to utility power receptacle (1J-G089).		
(2) Connect data cable connector P2 to MUX test connector (83J-G003).		
2. PROCEDURE.		
a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
c. On MLVS, do substeps below:		

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE MLVS display is made up of two lines of text. The steps below indicate the text display for each step.		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
d. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below: Turn on MC2	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
e. On MC/HYD ISOL control panel assembly, hold MC switch to 1 OFF position.		
f. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
NOTE The letters (p, a) in the steps below are address numbers which increment as the load proceeds.		
	MLVS screen displays the below In sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN.		
NOTE Placing the MC/HYD ISOL switch to the center position before 1 switch on GND PWR control panel assembly is set to AUTO may corrupt MC1 OFP.		
a. On GND PWR control panel assembly, set 1 switch to AUTO. b. If system OFP loading is complete, on MLVS, set PWR switch to OFF.		

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number.		
d. Remove electrical power (A1-F18AC-LMM-000).		
e. On MLVS, do substeps below:		
(1) Disconnect data cable connector P1 from connector J2.		
(2) Disconnect power cable connector P1 from connector J1.		
f. In aircraft nose wheelwell, do substeps below:		
(1) Disconnect data cable connector P2 from MUX test connector (83J-G003).		
(2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
g. Do displays test below: ON F/A-18C 162427 THRU 163782, A1-F18AC-745-200, WP004 00. ON F/A-18D 163434 THRU 163778, A1-F18AC-745-200, WP005 00. ON F/A-18C 163985 AND UP, A1-F18AG-745-200, WP004 00. ON F/A-18D 163985 AND UP, A1-F18AG-745-200, WP005 00.		

Table 3. SMS Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Armament Computer **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None		

Table 3. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003).		
2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

Table 3. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
d. On MLVS, do substeps below:	MLVS screen has displays listed below.	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		

Table 3. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct SMS boot load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE HARM target sequence/FLIR FOV/RAID switch must be held until the SMS boot loading is complete.		
e. On throttle grip, press and hold HARM target sequence/FLIR FOV/RAID switch		
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
f. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
g. On throttle grip, release HARM target sequence/FLIR FOV/RAID switch.		

Table 3. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
h. On GND PWR control panel assembly, set 3 switch to AUTO and then back to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
i. On MLVS, do substeps below:		
(1) Momentarily press EXEC until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(2) Press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN. a. If system OFP loading is complete, on MLVS set PWR switch to OFF. b. On GND PWR control panel assembly, set 3 switch to AUTO.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number.		

Table 3. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 4. SDC Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Signal Data Computer CP-1726/ASQ-194 (SDC) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None		

Table 4. SDC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P2 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003).		
2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
c. On MLVS, do substeps below:	MLVS screen has displays listed below.	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 4. SDC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 4. SDC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE On Aircraft Maintenance Indicator ID-2388/ASQ-194 (nose wheelwell DDI) press and release CONSUMABLES CHECK switch when directed by MLVS display.		
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
d. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE If the SDC will not load OFP and indicates an absence of memory (no OFP) the FIRAMS/SDP circuit breaker 85CBC004 position D2 on RLY CB PNL NO. 8 (door 10L), must be cycled while consumable check switch is pressed and held. Repeat step 2.d.		
3. SHUTDOWN. a. If system OFP loading is complete, on MLVS set PWR switch to OFF. b. Enter date/time of day data (A1-F18AC-LMM-000, WP018 02).		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		

Table 4. SDC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. Do table 2, WP004 00, to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J2. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 5. ADC Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Air Data Computer CP-1334A/A (ADC) **Related Systems Required** Avionics Cooling System Electrical System		

Table 5. ADC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
Support Equipment Required **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable P1 to connector J1. (2) Connect data cable P2 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003).		
2. PROCEDURE.		

Table 5. ADC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step. Both MC1 and MC2 must be OFF (GND PWR control panel assembly switch 1 must be set to AUTO) when loading the ADC OFP.		
c. On MLVS, do substeps below:		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		

Table 5. ADC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
d. On GND PWR control panel assembly, set and hold 2 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
e. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN. a. If system OFP loading is complete, on MLVS set PWR switch to OFF.		

Table 5. ADC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. On GND PWR control panel assembly, set 2 switch to AUTO.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 6. CSC Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Control-Converter C-10382/A (CSC) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below:		

Table 6. CSC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step. Both MC1 and MC2 must be OFF (GND PWR control panel assembly switch 1 must be set to AUTO) when loading the CSC OFP.		
c. On MLVS, do substeps below: (1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2 . If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	

Table 6. CSC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
d. On GND PWR control panel assembly, set and hold 2 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
e. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 6. CSC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
3. SHUTDOWN. a. If system OFP loading is complete, on MLVS set PWR switch to OFF. b. On GND PWR control panel assembly, set 2 switch to AUTO.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 7. DFIRS Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Data Transfer Interface Unit J-6008/A **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below:		

Table 7. DFIRS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003). 2. PROCEDURE.		
NOTE Both MC1 and MC2 must be OFF, GND PWR control panel assembly switch 1 must be set to AUTO, when loading a DFIRS OFP.		
b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
a. Apply electrical power (A1-F18AC-LMM-000).		
c. On MLVS, set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2 . If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	

Table 7. DFIRS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
d. On GND PWR control panel assembly, set and hold 3 switch to A ON position for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On MLVS, do substeps below: (1) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = file number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(2) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(3) Repeat step e.(2) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		

Table 7. DFIRS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(4) On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN. a. If system OFP loading is complete, on MLVS set PWR switch to OFF. b. On GND PWR control panel assembly, set 3 switch to AUTO.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below:		

Table 7. DFIRS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 8. DMC Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Digital Map Computer CP-1802/ASQ-194 (DMC)

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/USQ-131	Memory Loader-Verifier Set

Materials Required

None

NOTE

For Component Locator, refer to WP005 00.

For Test Equipment Hookup, refer to WP007 00.

Table 8. DMC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003).		
2. PROCEDURE.		
a. Apply electrical power (A1-F18AC-LMM-000).		
NOTE Both MC1 and MC2 must be OFF, GND PWR control panel assembly switch 1 must be set to AUTO, when loading DMC OFP.		
b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

Table 8. DMC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
c. On MLVS, set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
d. On GND PWR control panel assembly, set and hold 2 switch to A ON position for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000) 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On MLVS, do substeps below: (1) Set UP/VRFY/DOWN switch in the UP position.		

Table 8. DMC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(2) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(3) Repeat step e.(2) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
(4) On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN. a. If system OFP loading is complete, on MLVS set PWR switch to OFF. b. On GND PWR control panel assembly, set 3 switch to AUTO.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		

Table 8. DMC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 9. RDP Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Radar Data Processor CP-2062/APG-73 (RDP) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set		

Table 9. RDP Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
Materials Required None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2. b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

Table 9. RDP Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
NOTE The GND PWR control panel assembly switches 1, 3, and 4 must be set to AUTO when loading the radar OFP.		
c. On GND PWR control panel assembly, set and hold 2 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
d. On SNSR pod control box panel assembly, set RADAR switch to STBY.		
e. On MLVS, do substeps below:		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2 . If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	

Table 9. RDP Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step e.(3) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
(5) On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
f. On SNSR pod control box panel assembly, set RADAR switch to OFF.		
3. SHUTDOWN.		
a. On MLVS, set PWR switch to OFF.		

Table 9. RDP Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. On GND PWR control panel assembly, set 2 switch to AUTO.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 10. CLC Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Command Launch Computer CP-1001( )/AWG **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-1311 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. STORES SAFETY INSPECTION (A1-F18AE-LWS-000).		
WARNING To prevent injury or death of personnel, all live weapons and explosive cartridges must be removed from aircraft and gun must be safetied before doing this test.		
a. Make sure electrical power is off (A1-F18AC-LMM-000). b. Make sure all weapons are removed from aircraft (A1-F18AE-LWS-000).		

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. Make sure all explosive cartridges are removed from cartridge chambers for all Aircraft Bomb Ejection Racks BRU-32( ) installed on aircraft (A1-F18AE-LWS-000). d. Make sure all explosive cartridges are removed from cartridge chambers on Aircraft Bomb Ejector Racks BRU-33 ( ) if installed on aircraft (A1-F18AE-LWS-000). e. Make sure all explosive cartridges are removed from cartridge chambers on Aircraft Guided Missile Launcher LAU-116/A AIM-7 fuselage stations if installed on aircraft (A1-F18AE-LWS-000). f. Make sure all Aircraft Guided Missile Launcher LAU-116/A hooks are closed and SAFETY RELEASE knob is rotated clockwise. g Make sure all explosives are removed from breeches on Multiple Ejector Racks (MER) if installed on aircraft (A1-F18AE-LWS-000). h. Make sure gun electrical signal safety switch is set to safe (extended) position, aft of door 6 (A1-F18AE-LWS-000).		
NOTE Gun safety handle may not go completely to the locked position until aircraft power is applied.		

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
i. Make sure gun hold-back mechanism handle is set to cleared; gun hold-back handle indicator (extended) (A1-F18AE-LWS-000). j. Close hooks on Bomb Ejector Racks BRU-32( ) for station used to ID HARM and set ground safety handle to LOCKED (A1-F18AE-LWS-000). 2. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2. b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003). 3. PROCEDURE. a. Open door 14R (A1-F18AC-LMM-010).		

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. On Armament Computer CP-1342/AYQ-9(V), do substeps below: (1) Set ARMAMENT switches to 64 for station used to ID HARM in step 1.j. (2) For remaining stations, set switches to 00, except stations with tank installed set switches to 01. c. Connect ground intercommunications (A1-F18AC-LMM-000). d. Apply electrical power (A1-F18AC-LMM-000).		
e. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
f. On SNSR pod control box panel assembly, make sure RADAR switch is set to OFF.		
NOTE If a malfunction occurs during this test, make sure circuit breakers are closed; ON 163427 THRU 165206 (A1-F18AE-740-200, WP011 00) or ON 165207 AND UP (A1-F18AH-740-200, WP006 00).		
g. On MC/HYD ISOL control panel assembly, set MC switch to NORM.		
h. After 80 to 180 seconds, select A/G master mode button.	A/G master mode button lights.	Make sure enough time has elapsed for SMS to complete self test, do steps 3.d. through 3.h.

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
i. On flaps, landing gear and stores panel assembly, select station used to ID HARM.	Selected station light comes on.	ON 163427 THRU 165206, do table 1 (A1-F18AE-740-200, WP017 00). ON 165207 AND UP, do table 1 (A1-F18AH-740-200, WP015 00).
j. On left hand vertical console control panel, move JETT select switch from SAFE to STORES.	In avionics bay door 13R, the CLC cooling fan comes on.	ON 163427 THRU 165206, replace CLC (A1-F18AE-740-300, WP011 00). ON 165207 AND UP, replace CLC (A1-F18AH-740-300, WP008 00).
NOTE MLVS display consists of two lines of text. The steps below indicate text display for each step.		
k. On MLVS, do substeps below:		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE See WP003 00 to verify program load CONFIG/IDENT number. For an operational CLC, both HARM operational program (PGM) and Electronic Intelligence (ELINT) files must be loaded. PGM file must be loaded before ELINT file.		
(2) If PGM is to be loaded into CLC do step 3.j.(3). If ELINT is to be loaded into CLC do step 3.j.(9). (3) Set UP/VRFY/DOWN switch in the UP position		
NOTE The displays below are in the format: n = slote number, f = file name, and x = file name extension (optional entry).		
(4) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(5) Momentarily press EXEC until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The letters (p, a) shown in the below steps are address numbers which increment as the load proceeds.		
(6) On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffff.xxx Upload pppp aaaa (2) FILE:n ffffff.xxx Verify pppp aaaa (3) FILE:n ffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(7) If ELINT is to be loaded into CLC, do step 3.j.(9). If not go to step 4.		
NOTE PGM file must be loaded before ELINT file.		
(8) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format n = file number, f = file name, and x = file name extension (optional entry).		
(9) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(10) Momentarily press EXEC until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The letters (p, a) shown in the below steps are address numbers which increment as the load proceeds.		
(11) On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffff.xxx Upload pppp aaaa (2) FILE:n ffffff.xxx Verify pppp aaaa (3) FILE:n ffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
4. TURN OFF. a. On GND PWR control panel assembly, set 3 switch to AUTO.		

Table 10. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. If system OFP loading is complete, on MLVS set PWR switch to OFF. c. Do table 2, WP004 00, to verify correct program identification. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 11. FLIR Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Forward Looking Infrared Receiver AN/AAS-38B (FLIR) **Related Systems Required** Avionics Cooling System Electrical System		

Table 11. FLIR Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
Support Equipment Required **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
NOTE To verify that cable is correctly seated when connecting flex-type cables, make sure that red line on the MLVS connector is not visible.		
a. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect data cable W3 connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable W2 connector P2 to utility power receptacle (1J-G089).		

Table 11. FLIR Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Connect data cable W3 connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
c. On MLVS, set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		

Table 11. FLIR Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
d. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On SNSR pod control box panel assembly, set FLIR switch to STBY.		
f. On MLVS, do the substeps below:		
(1) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(2) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(3) Repeat step f.(2) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.FLR EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
g. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: Waiting 01 Waiting 02 Waiting 03 FILE:n ffffffff.FLR Upload xxxx FILE:n ffffffff.FLR Upload Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 11. FLIR Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
3. SHUTDOWN. a. If system OFP loading is complete, on MLVS set PWR switch to OFF.		
b. On SNSR pod control box panel assembly, set FLIR switch to OFF. Wait 2 minutes for the FLIR to power down.	On the FLIR pod, air vent door shuts when FLIR powers down.	
c. On GND PWR control panel assembly, set 2 switch to AUTO.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
d. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number.		
e. Remove electrical power (A1-F18AC-LMM-000).		
f. On MLVS, do substeps below:		
(1) Disconnect data cable W3 connector P1 from connector J2.		
(2) Disconnect power cable W2 connector P1 from connector J1.		
g. In aircraft nose wheelwell, do substeps below:		
(1) Disconnect data cable W3 connector P2 from MUX test connector (83J-G003).		
(2) Disconnect power cable W2 connector P2 from utility power receptacle (1J-G089).		

Table 12. CIT Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Combined Interrogator Transponder AN/APX-111(V) (CIT) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below:		

Table 12. CIT Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step. Both MC1 and MC2 must be OFF (GND PWR control panel assembly switch 1 must be set to AUTO) when loading the CIT OFP.		
c. On MLVS, do substeps below:		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	

Table 12. CIT Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
d. On GND PWR control panel assembly, set and hold 2 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
f. On UFC, turn on IFF system. Allow 30 seconds for IFF to complete power on BIT.	Make sure IFF BIT status is go on BIT display.	Do table 2, WP004 00.

Table 12. CIT Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
g. Do substeps below: (1) On MLVS, press and hold EXEC. (2) On GND PWR control panel assembly, set 1 switch to AUTO. (3) Release EXEC switch on MLVS.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Waiting 02 01 Waiting 02 01 (2) FILE:n ffffffff.xxx Upload pppp aaaa (3) FILE:n ffffffff.xxx Verify pppp aaaa (4) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN. a. If system OFP loading is complete, on MLVS set PWR switch to OFF. b. On GND PWR control panel assembly, set 2 switch to AUTO.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number.		

Table 12. CIT Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 13. DCS Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Digital Communications System RT-1824(C)/ARC (DCS) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None		

Table 13. DCS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003).		
2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

Table 13. DCS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step. Both MC1 and MC2 must be OFF (GND PWR control panel assembly switch 1 must be set to AUTO) when loading the CIT OFP.		
c. On MLVS, do substeps below:		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 13. DCS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(4) Repeat step c.(3) until correct program load CONFIG/ IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
d. On GND PWR control panel assembly, set and hold 2 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On UFC, turn volume control knob (on COMM 2) fully clockwise to turn on DCS.		
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
f. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Waiting 05 01 Waiting 02 01 (2) FILE:n ffffffff.xxx Upload pppp aaaa (3) FILE:n ffffffff.xxx Verify pppp aaaa (4) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN. a. If system OFP loading is complete, on MLVS set PWR switch to OFF. b. On GND PWR control panel assembly, set 2 switch to AUTO.		

Table 13. DCS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On UFC, turn volume control knob (on COMM 2) fully counterclockwise to turn off DCS.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
d. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. e. Remove electrical power (A1-F18AC-LMM-000). f. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. g. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 14. EIBU Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Enhanced Interference Blanker Unit MX-11741A (EIBU) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below:		

Table 14. EIBU Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003).		
NOTE Both the upload and verify procedures must be completed for the EIBU to be operational.		
2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step. Both MC1 and MC2 must be OFF (GND PWR control panel assembly switch 1 must be set to AUTO) when loading the CIT OFP.		
c. On MLVS, do substeps below:		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 14. EIBU Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program load CONFIG/ IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
d. On GND PWR control panel assembly, set and hold 3 switch to A or B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).

Table 14. EIBU Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
e. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Waiting 05 01 Waiting 02 01 (2) FILE:n ffffffff.xxx Upload pppp aaaa (3) FILE:n ffffffff.xxx Verify pppp aaaa (4) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN. a. If system OFP loading is complete, on MLVS set PWR switch to OFF. b. On GND PWR control panel assembly, set 3 switch to AUTO.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2.		

Table 14. EIBU Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 15. CPS Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Computer Power Supply CP-1325/APG-65 (CPS)

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/USQ-131	Memory Loader-Verifier Set

Materials Required

None

Table 15. CPS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003).		
2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

Table 15. CPS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
NOTE The GND PWR control panel assembly switches 1, 3, and 4 must be set to AUTO when loading the radar OFP.		
c. On GND PWR control panel assembly, set and hold 2 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
d. On SNSR pod control box panel assembly, set RADAR switch to STBY.		
e. On MLVS, do substeps below:		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2 . If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	

Table 15. CPS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step e.(3) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
(5) On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
f. On SNSR pod control box panel assembly, set RADAR switch to OFF.		
3. SHUTDOWN.		
a. On MLVS, set PWR switch to OFF.		

Table 15. CPS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. On GND PWR control panel assembly, set 2 switch to AUTO.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 16. MC1 Boot Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Digital Data Computer No. 1 CP-2360/AYK-14 (MC1)

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/USQ-131	Memory Loader-Verifier Set

Materials Required

None

NOTE

For Component Locator, refer to WP005 00.

For Test Equipment Hookup, refer to WP007 00.

1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below:		
(1) Connect power cable connector P1 to connector J1.		
(2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below:		

Table 16. MC1 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET. c. On MLVS, do substeps below:		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	

Table 16. MC1 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program boot CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
d. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below: Turn On MC1	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
e. On MC/HYD ISOL control panel assembly, set MC switch to 2 OFF position and hold during step f.		
f. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).

Table 16. MC1 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
3. SHUTDOWN.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE Placing the MC/HYD ISOL switch to the center position before 1 switch on GND PWR control panel assembly is set to AUTO may corrupt MC2 OFP.		
a. On GND PWR control panel assembly, set 1 switch to AUTO. b. If system boot loading is complete, on MLVS, set PWR switch to OFF.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power-up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000).		

Table 16. MC1 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089). f. Do displays test below: ON F/A-18C 163427 THRU 163782, A1-F18AC-745-200, WP004 00. ON F/A-18D 163434 THRU 163778, A1-F18AC-745-200, WP005 00. ON F/A-18C 163985 AND UP, A1-F18AG-745-200, WP004 00. ON F/A-18D 163986 AND UP, A1-F18AG-745-200, WP005 00. g. Enter stored data variation into mission computer memory as required.		

Table 17. MC2 Boot Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Digital Data Computer No. 2 CP-2360/AYK-14 (MC2) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below:		

Table 17. MC2 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET. c. On MLVS, do substeps below:		
NOTE MLVS display is made up of two lines of text. The steps below indicate the text display for each step.		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	

Table 17. MC2 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program boot CONFIG/ IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
d. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below: Turn on MC2	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
e. On MC/HYD ISOL control panel assembly, hold MC switch to 1 OFF position.		
f. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).

Table 17. MC2 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The letters (p, a) in the steps below are address numbers which increment as the load proceeds.		
3. SHUTDOWN.	MLVS screen displays the below In sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE Placing the MC/HYD ISOL switch to the center position before 1 switch on GND PWR control panel assembly is set to AUTO may corrupt MC1 OFP.		
a. On GND PWR control panel assembly, set 1 switch to AUTO. b. If system boot loading is complete, on MLVS, set PWR switch to OFF.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below:		

Table 17. MC2 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089). g. Do displays test below: ON F/A-18C 162427 THRU 163782, A1-F18AC-745-200, WP004 00. ON F/A-18D 163434 THRU 163778, A1-F18AC-745-200, WP005 00. ON F/A-18C 163985 AND UP, A1-F18AG-745-200, WP004 00. ON F/A-18D 163985 AND UP, A1-F18AG-745-200, WP005 00.		

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

AVIONICS LOAD/VERIFICATION PROCEDURES USING AN/USQ-131 LOADER-VERIFIER SET

EFFECTIVITY: F/A-18A AND F/A-18B

Reference Material

Airborne Weapons/Stores Loading Manual	A1-F18AE-LWS-000
Line Maintenance Procedures	A1-F18AC-LMM-000
Multipurpose Display Group	A1-F18AC-745-200
Displays Test F/A-18A	WP004 00
Displays Test F/A-18B	WP005 00
Software Configuration Manual	A1-F18AC-SCM-000
Program Load Versions	WP003 00
Program Load CONFIG/IDENT Verification	WP004 00
Component Locator	WP005 00
Test Equipment Hookup Locator	WP007 00
Memory Loader-Verifier Set AN/USQ-131	NAVAIR 16-30USQ131-1
Extended BIT	WP003 00

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Record of Applicable Technical Directives

None

1. INTRODUCTION.

2. This work package includes procedures for loading operational flight programs (OFP) using the Memory Loader-Verifier Set AN/USQ-131 (MLVS).

3. Each programmable WRA has a separate table. The WRAs include:

a. Digital Data Computer No. 1 and No. 2 (MC1, MC2).

b. Armament Computer CP-1342/AYQ-9(V) (SMS).

c. Command Launch Computer CP-1001( )/AWG (CLC).

d. Receiver CN-1694(V)4/ASN-172(V) (EGI).

e. Digital Communications System RT-1824(C)/ARC (DCS).

f. Enhanced Interference Blanker Unit MX-11741A (EIBU).

g. Combined Interrogator Transponder AN/APX-111(V) (CIT).

h. Computer Power Supply CP-1325/APG-65 (CPS).

Table 1. MC1 Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Digital Data Computer No. 1 (MC1) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect data cable W3 connector P1 to connector J2. b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable W2 connector P2 to utility power receptacle (1J-G089). (2) Connect data cable W3 connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET. c. On MLVS, do substeps below:		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) If not in UP position, set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.MC1 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
d. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below: FILE:n ffffffff.MC1 Turn On MC1	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
e. On MC/HYD ISOL control panel assembly, set MC switch to 2 OFF position and hold during step f.		
f. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).

NOTE

The letters (p, a) shown in the steps below are address numbers which increment as the load continues.

	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.MC1 Upload pppp aaaa (2) FILE:n ffffffff.MC1 pppp aaaa (3) FILE:n ffffffff.MC1 Upload Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN.		

NOTE

Placing the MC/HYD ISOL switch to the center position before 1 switch on GND PWR control panel assembly is set to AUTO may corrupt MC2 OFP.

a. On GND PWR control panel assembly, set 1 switch to AUTO.		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. If system OFP loading is complete, on MLVS, set PWR switch to OFF.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power-up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable W3 connector P1 from connector J2. (2) Disconnect power cable W2 connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable W3 connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable W2 connector P2 from utility power receptacle (1J-G089). g. Do displays test below: ON F/A-18A, A1-F18AC-745-200, WP004 00. ON F/A-18B, A1-F18AC-745-200, WP005 00.		

Table 1. MC1 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
h. Enter stored data variation into mission computer memory as required.		

Table 2. MC2 Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Digital Data Computer No. 2 (MC2) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect data cable W3 connector P1 to connector J2. b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable W2 connector P2 to utility power receptacle (1J-G089). (2) Connect data cable W3 connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET. c. On MLVS, do substeps below:		

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE MLVS display is made up of two lines of text. The steps below indicate the text display for each step.		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) If not in UP position, set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.MC2 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
d. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below: FILE:n ffffffff.MC2 Turn on MC2	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
e. On MC/HYD ISOL control panel assembly, set MC switch to 1 OFF position and hold during step f.		
f. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
NOTE The letters (p, a) in the steps below are address numbers which increment as the load continues.		
	MLVS screen displays the below In sequence: (1) FILE:n ffffffff.MC2 Upload pppp aaaa (2) FILE:n ffffffff.MC2 Verify pppp aaaa (3) FILE:n ffffffff.MC2 Upload Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN.		
NOTE Placing the MC/HYD ISOL switch to the center position before 1 switch on GND PWR control panel assembly is set to AUTO may corrupt MC1 OFP.		
a. On GND PWR control panel assembly, set 1 switch to AUTO.		

Table 2. MC2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. If system OFP loading is complete, on MLVS, set PWR switch to OFF.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable W3 connector P1 from connector J2. (2) Disconnect power cable W2 connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable W3 connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable W2 connector P2 from utility power receptacle (1J-G089). g. Do displays test below: ON F/A-18A, A1-F18AC-745-200, WP004 00. ON F/A-18B, A1-F18AC-745-200, WP005 00.		

Table 3. SMS Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Armament Computer CP-1342/AYQ-9(V) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors. **NOTE** To verify that the cable is correctly seated when connecting flex-type cables make sure that red line on the MLVS connector is not visible.		
a. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect data cable W3 connector P1 to connector J2.		

Table 3. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable W2 connector P2 to utility power receptacle (1J-G089). (2) Connect data cable W3 connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
c. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
d. On MLVS, do substeps below:	MLVS screen has displays listed below.	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 3. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) If not in UP position, set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Wait 1 minute after SMS power up before doing step d.(5).		
(4) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 3. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(5) Repeat step d.(4) until correct SMS boot load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.SMS EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
e. On GND PWR control panel assembly, set 3 switch to AUTO and then back to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
f. On MLVS, do substeps below:		
(1) Wait 1 minute after SMS power up before doing step f.(2).		
(2) Momentarily press EXEC until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.SMS EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(3) Press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.SMS Upload pppp aaaa (2) FILE:n ffffffff.SMS Verify pppp aaaa (3) FILE:n ffffffff.SMS Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN.		
a. If system OFP loading is complete, on MLVS, set PWR switch to OFF.		
b. On GND PWR control panel assembly, set 3 switch to AUTO.		

Table 3. SMS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable W3 connector P1 from connector J2. (2) Disconnect power cable W2 connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable W3 connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable W2 connector P2 from utility power receptacle (1J-G089).		

Table 4. CLC Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Command Launch Computer CP-1001( )/AWG (CLC) **Related Systems Required** Avionics Cooling System Electrical System		

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
Support Equipment Required **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. STORES SAFETY INSPECTION (A1-F18AC-LWS-000).		
WARNING To prevent injury or death of personnel, all live weapons and explosive cartridges must be removed from aircraft and gun must be safetied before doing this test.		
a. Make sure electrical power is off (A1-F18AC-LMM-000). b. Make sure all weapons are removed from aircraft (A1-F18AC-LWS-000). c. Make sure all explosive cartridges are removed from cartridge chambers for all Aircraft Bomb Ejection Racks BRU-32( ) installed on aircraft (A1-F18AC-LWS-000).		

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
d. Make sure all explosive cartridges are removed from cartridge chambers on Aircraft Bomb Ejector Racks BRU-33 ( ) if installed on aircraft (A1-F18AC-LWS-000).		
e. Make sure all explosive cartridges are removed from cartridge chambers on Aircraft Guided Missile Launcher LAU-116/A AIM-7 fuselage stations if installed on aircraft (A1-F18AC-LWS-000).		
f. Make sure all Aircraft Guided Missile Launcher LAU-116/A hooks are closed and SAFETY RELEASE knob is rotated clockwise.		
g. Make sure all explosives are removed from breeches on Multiple Ejector Racks (MER) if installed on aircraft (A1-F18AC-LWS-000).		
h. Make sure gun electrical signal safety switch is set to safe (extended) position, aft of door 6 (A1-F18AC-LWS-000).		
NOTE Gun safety handle may not go completely to the locked position until aircraft power is applied.		
i. Make sure gun hold-back mechanism handle is set to cleared; gun hold-back handle indicator (extended) (A1-F18AC-LWS-000).		
j. Close hooks on Bomb Ejector Racks BRU-32( ) for station used to ID HARM (L OUTBD) and set ground safety handle to LOCKED (A1-F18AC-LWS-000).		

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
2. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up the connector keys with the mating keyways before mating the connectors.		
a. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect data cable W3 connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable W2 connector P2 to utility power receptacle (1J-G089). (2) Connect data cable W3 connector P2 to MUX test connector (83J-G003).		
3. PROCEDURE. a. Open door 14R (A1-F18AC-LMM-010).		
b. On Armament Computer CP-1342/AYQ-9(V), do substeps below: (1) Set ARMAMENT switches to 64 for station used to ID HARM in step 1.j. (2) For remaining stations, set switches to 00, except stations with tank installed set switches to 01.		

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. Connect ground intercommunications (A1-F18AC-LMM-000).		
d. Apply electrical power (A1-F18AC-LMM-000).		
e. On GND PWR control panel assembly, set and hold 3 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
f. On SNSR pod control box panel assembly, make sure RADAR switch is set to OFF.		
NOTE If a malfunction occurs during this test, make sure circuit breakers are closed (A1-F18AC-740-200, WP004 00).		
g. On MC/HYD ISOL control panel assembly, set MC switch to NORM.		
h. After 80 to 180 seconds, select A/G master mode button.	A/G master mode button lights.	Make sure enough time has elapsed for SMS to complete self test, do steps 3.d. through 3.h.
i. On flaps, landing gear and stores panel assembly, select station used to ID HARM.	Selected station light comes on.	Do table 2 (A1-F18AC-740-200, WP013 00).
j. On left hand vertical console control panel, move JETT select switch from SAFE to STORES.	In avionics bay door 13R, the CLC cooling fan comes on.	Replace CLC (A1-F18AC-740-300, WP010 00).
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
k. On MLVS, do substeps below:		

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number. For an operational CLC, both HARM operational program (PGM) and Electronic Intelligence (ELINT) files must be loaded. PGM file must be loaded before ELINT file.		
(2) If PGM is to be loaded into CLC do step 3.k.(3). If ELINT is to be loaded into CLC do step 3.k.(8). (3) If not in UP position, set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = file name, and x = file name extension (optional entry).		
(4) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(5) Momentarily press EXEC until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffff.HRM EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The letters (p, a) shown in the below steps are address numbers which increment as the load continues.		
(6) On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffff.HRM Upload pppp aaaa (2) FILE:n ffffff.HRM Verify pppp aaaa (3) FILE:n ffffff.HRM Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(7) If ELINT is to be loaded into CLC, do step 3.k.(8). If not go to step 4.		
NOTE PGM file must be loaded before ELINT file.		
(8) If not in UP position, set UP/VRFY/DOWN switch in the UP position.		

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The displays below are in the format n = slot number, f = file name, and x = file name extension (optional entry).		
(9) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(10) Momentarily press EXEC until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffff.HRM EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The letters (p, a) shown in the below steps are address numbers which increment as the load continues.		
(11) On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffff.HRM Upload pppp aaaa (2) FILE:n ffffff.HRM Verify pppp aaaa (3) FILE:n ffffff.HRM Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
4. SHUTDOWN. a. On left hand vertical console control panel, move JETT select from STORES to SAFE. b. On GND PWR control panel assembly, set 3 switch to AUTO. c. If system OFP loading is complete, on MLVS, set PWR switch to OFF.		

Table 4. CLC Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
d. Do table 2, WP004 00, to verify correct program identification.		
e. Remove electrical power (A1-F18AC-LMM-000).		
f. On MLVS, do substeps below:		
(1) Disconnect data cable W3 connector P1 from connector J2.		
(2) Disconnect power cable W2 connector P1 from connector J1.		
g. In aircraft nose wheelwell, do substeps below:		
(1) Disconnect data cable W3 connector P2 from MUX test connector (83J-G003).		
(2) Disconnect power cable W2 connector P2 from utility power receptacle (1J-G089).		

Table 5. EGI Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Receiver CN-1694(V)4/ASN-172(V) (EGI) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below:		

Table 5. EGI Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step. Both MC1 and MC2 must be OFF (GND PWR control panel assembly switch 1 must be set to AUTO) when loading the EGI OFP.		
c. On MLVS, do substeps below: (1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	

Table 5. EGI Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program load CONFIG/IDENT number is displayed (EGI BOOT OFP).	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
d. On GND PWR control panel assembly, set and hold 2 switch to A or B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On SNSR panel, set INS control mode switch to GND.		

Table 5. EGI Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
f. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
g. Set PWR switch to OFF.		
h. Set UP/VRFY/DOWN switch in the VRFY position.		
i. Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	

Table 5. EGI Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
j. Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
k. Repeat step j. until correct program load CONFIG/IDENT number is displayed (EGI BOOT OFP).	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
l. Press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Verify Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
m. Set PWR switch to OFF.		
n. Set UP/VRFY/DOWN switch in the UP position.		
o. Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 5. EGI Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
p. Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
q. Repeat step p. until correct program load CONFIG/IDENT number is displayed (EGI OFP).	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
r. Press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
s. Set PWR switch to OFF.		
t. Set UP/VRFY/DOWN switch in the VRFY position.		

Table 5. EGI Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
u. Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
v. Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
w. Repeat step v. until correct program load CONFIG/IDENT number is displayed (EGI OFP).	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 5. EGI Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
x. Press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Verify Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN. a. If system OFP loading is complete, on MLVS, set PWR switch to OFF. b. On GND PWR control panel assembly, set 2 switch to AUTO. c. On SNSR panel, set INS control mode switch to OFF.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
d. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. e. Remove electrical power (A1-F18AC-LMM-000). f. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1.		

Table 5. EGI Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
g. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 6. DCS Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Digital Communications System RT-1824(C)/ARC (DCS) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below:		

Table 6. DCS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step. Both MC1 and MC2 must be OFF (GND PWR control panel assembly switch 1 must be set to AUTO) when loading the CIT OFP.		
c. On MLVS, do substeps below:		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	

Table 6. DCS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
d. On GND PWR control panel assembly, set and hold 2 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On UFC, turn volume control knob (on COMM 2) fully clockwise to turn on DCS.		

Table 6. DCS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
f. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Waiting 05 01 Waiting 02 01 (2) FILE:n ffffffff.xxx Upload pppp aaaa (3) FILE:n ffffffff.xxx Verify pppp aaaa (4) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN. a. If system OFP loading is complete, on MLVS set PWR switch to OFF. b. On GND PWR control panel assembly, set 2 switch to AUTO. c. On UFC, turn volume control knob (on COMM 2) fully counter-clockwise to turn off DCS.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
d. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. e. Remove electrical power (A1-F18AC-LMM-000).		

Table 6. DCS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
f. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. g. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 7. EIBU Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Enhanced Interference Blanker Unit MX-11741A (EIBU) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None		

Table 7. EIBU Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003).		
NOTE Both the upload and verify procedures must be completed for the EIBU to be operational.		
2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

Table 7. EIBU Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step. Both MC1 and MC2 must be OFF (GND PWR control panel assembly switch 1 must be set to AUTO) when loading the CIT OFP.		
c. On MLVS, do substeps below:		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 7. EIBU Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(4) Repeat step c.(3) until correct program load CONFIG/ IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
d. On GND PWR control panel assembly, set and hold 3 switch to A or B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
e. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Waiting 05 01 Waiting 02 01 (2) FILE:n ffffffff.xxx Upload pppp aaaa (3) FILE:n ffffffff.xxx Verify pppp aaaa (4) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN. a. If system OFP loading is complete, on MLVS set PWR switch to OFF. b. On GND PWR control panel assembly, set 3 switch to AUTO.		

Table 7. EIBU Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 8. CIT Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Combined Interrogator Transponder AN/APX-111(V) (CIT) **Related Systems Required** Avionics Cooling System Electrical System		

Table 8. CIT Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
Support Equipment Required **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003).		
2. PROCEDURE.		

Table 8. CIT Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step. Both MC1 and MC2 must be OFF (GND PWR control panel assembly switch 1 must be set to AUTO) when loading the CIT OFP.		
c. On MLVS, do substeps below:		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		

Table 8. CIT Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
d. On GND PWR control panel assembly, set and hold 2 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
f. On UFC, turn on IFF system. Allow 30 seconds for IFF to complete power on BIT.	Make sure IFF BIT status is GO on BIT display.	Do table 1, WP004 00.

Table 8. CIT Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
g. Do substeps below:		
(1) On MLVS, press and hold EXEC.		
(2) On GND PWR control panel assembly, set 1 switch to AUTO.		
(3) Release EXEC switch on MLVS.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Waiting 02 01 Waiting 02 01 (2) FILE:n ffffffff.xxx Upload pppp aaaa (3) FILE:n ffffffff.xxx Verify pppp aaaa (4) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN.		
a. If system OFP loading is complete, on MLVS set PWR switch to OFF.		
b. On GND PWR control panel assembly, set 2 switch to AUTO.		
c. On UFC, turn off IFF system.		

Table 8. CIT Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
d. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. e. Remove electrical power (A1-F18AC-LMM-000). f. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. g. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 9. CPS Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Computer Power Supply CP-1325/APG-65 (CPS) **Related Systems Required** Avionics Cooling System Electrical System		

Table 9. CPS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
Support Equipment Required **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable connector P1 to connector J1. (2) Connect data cable connector P1 to connector J2. b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable connector P2 to utility power receptacle (1J-G089). (2) Connect data cable connector P2 to MUX test connector (83J-G003).		
2. PROCEDURE.		

Table 9. CPS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
NOTE The GND PWR control panel assembly switches 1, 3, and 4 must be set to AUTO when loading the radar OFP.		
c. On GND PWR control panel assembly, set and hold 2 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
d. On SNSR pod control box panel assembly, set RADAR switch to STBY.		
e. On MLVS, do substeps below:		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 9. CPS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2 . If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) Set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step e.(3) until correct program load CONFIG/ IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load proceeds.		
(5) On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.xxx Upload pppp aaaa (2) FILE:n ffffffff.xxx Verify pppp aaaa (3) FILE:n ffffffff.xxx Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 9. CPS Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
f. On SNSR pod control box panel assembly, set RADAR switch to OFF. 3. SHUTDOWN. a. On MLVS, set PWR switch to OFF. b. On GND PWR control panel assembly, set 2 switch to AUTO.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below: (1) Disconnect data cable connector P1 from connector J2. (2) Disconnect power cable connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable connector P2 from utility power receptacle (1J-G089).		

Table 10. MC1 Boot Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Digital Data Computer No. 1 CP-2360/AYK-14 (MC1) **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect data cable W3 connector P1 to connector J2. b. In aircraft nose wheelwell, do substeps below:		

Table 10. MC1 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Connect power cable W2 connector P2 to utility power receptacle (1J-G089). (2) Connect data cable W3 connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET. c. On MLVS, do substeps below:		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	

Table 10. MC1 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) If not in UP position, set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program boot CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.MC1 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
d. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below: FILE:n ffffffff.MC1 Turn On MC1	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
e. On MC/HYD ISOL control panel assembly, set MC switch to 2 OFF position and hold during step f.		
f. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).

Table 10. MC1 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The letters (p, a) shown in the steps below are address numbers which increment as the load continues.		
3. SHUTDOWN.	MLVS screen displays the below in sequence: (1) FILE:n ffffffff.MC1 Upload pppp aaaa (2) FILE:n ffffffff.MC1 pppp aaaa (3) FILE:n ffffffff.MC1 Upload Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE Placing the MC/HYD ISOL switch to the center position before 1 switch on GND PWR control panel assembly is set to AUTO may corrupt MC2 OFP.		
a. On GND PWR control panel assembly, set 1 switch to AUTO. b. If system OFP loading is complete, on MLVS, set PWR switch to OFF.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power-up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000).		

Table 10. MC1 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
e. On MLVS, do substeps below: (1) Disconnect data cable W3 connector P1 from connector J2. (2) Disconnect power cable W2 connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable W3 connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable W2 connector P2 from utility power receptacle (1J-G089). g. Do displays test below: ON F/A-18A, A1-F18AC-745-200, WP004 00. ON F/A-18B, A1-F18AC-745-200, WP005 00. h. Enter stored data variation into mission computer memory as required.		

Table 11. MC2 Boot Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Digital Data Computer No. 2 CP-2360/AYK-14 (MC2)

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/USQ-131	Memory Loader-Verifier Set

Materials Required

None

NOTE

For Component Locator, refer to WP005 00.

For Test Equipment Hookup, refer to WP007 00.

1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
a. On MLVS, do substeps below:		
(1) Connect power cable W2 connector P1 to connector J1.		
(2) Connect data cable W3 connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below:		

Table 11. MC2 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Connect power cable W2 connector P2 to utility power receptacle (1J-G089). (2) Connect data cable W3 connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET. c. On MLVS, do substeps below:		
NOTE MLVS display is made up of two lines of text. The steps below indicate the text display for each step.		
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	

Table 11. MC2 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
(2) If not in UP position, set UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until correct program boot CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.MC2 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
d. On MLVS, press and hold EXEC for 3 seconds.	MLVS screen displays the below: FILE:n ffffffff.MC2 Turn on MC2	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
e. On MC/HYD ISOL control panel assembly, set MC switch to 1 OFF position and hold during step f.		
f. On GND PWR control panel assembly, set and hold 1 switch to B ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).

Table 11. MC2 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The letters (p, a) in the steps below are address numbers which increment as the load continues.		
3. SHUTDOWN.	MLVS screen displays the below In sequence: (1) FILE:n ffffffff.MC2 Upload pppp aaaa (2) FILE:n ffffffff.MC2 Verify pppp aaaa (3) FILE:n ffffffff.MC2 Upload Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE Placing the MC/HYD ISOL switch to the center position before 1 switch on GND PWR control panel assembly is set to AUTO may corrupt MC1 OFP.		
a. On GND PWR control panel assembly, set 1 switch to AUTO. b. If system boot loading is complete, on MLVS, set PWR switch to OFF.		
NOTE Electrical power must be removed before CONFIG/IDENT verification or system operation to make sure of correct power up sequencing in mission computer.		
c. Do table 2, WP004 00 to verify correct program load CONFIG/IDENT number. d. Remove electrical power (A1-F18AC-LMM-000). e. On MLVS, do substeps below:		

Table 11. MC2 Boot Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Disconnect data cable W3 connector P1 from connector J2. (2) Disconnect power cable W2 connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable W3 connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable W2 connector P2 from utility power receptacle (1J-G089). g. Do displays test below: ON F/A-18A, A1-F18AC-745-200, WP004 00. ON F/A-18B, A1-F18AC-745-200, WP005 00.		

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

EW LOAD/VERIFICATION PROCEDURES USING AN/USQ-131 LOADER-VERIFIER SET

EFFECTIVITY: F/A-18C AND F/A-18D

Reference Material

Airborne Weapons/Stores Loading Manual	A1-F18AE-LWS-000
Line Maintenance Procedures	A1-F18AC-LMM-000
Multipurpose Display Group	A1-F18AC-745-200
Displays Test F/A-18C	WP004 00
Displays Test F/A-18D	WP005 00
Multipurpose Display Group	A1-F18AG-745-200
Displays Test F/A-18C	WP004 00
Displays Test F/A-18D	WP005 00
Software Configuration Manual	A1-F18AC-SCM-000
Program Load Versions	WP003 00
Program Load CONFIG/IDENT Verification	WP004 00
Component Locator	WP005 00
Test Equipment Hookup Locator	WP007 00
Memory Loader-Verifier Set AN/USQ-131	NAVAIR 16-30USQ131-1
Extended BIT	WP003 00

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Record of Applicable Technical Directives

None

1. INTRODUCTION.

2. This work package includes procedures for loading operational flight programs (OFP) using the Memory Loader-Verifier Set AN/USQ-131 (MLVS).

3. Each programmable WRA has a separate table. The WRAs include:

a. Radar Warning Receiver CP-1239C/ALR-67(V)2 (RWR)

b. Receiver Transmitter RT-1079/ALQ-126B (ECS)

c. Countermeasures Dispensing System Programmer CD-45/ALE-47 (CMDS)

d. Airborne Self Protect Jammer AN/ALQ-165 (ASPJ)

Table 1. ALR-67(V)2 Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Radar Warning Computer CP-1239C/ALR-67(V)2 **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		

Table 1. ALR-67(V)2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE To verify that the cable is properly seated when connecting flex-type cables make sure that red line on the MLVS connector is not visible.		
a. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect data cable W3 connector P1 to connector J2. b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable W2 connector P2 to utility power receptacle (1J-G089). (2) Connect data cable W3 connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
c. On MLVS, do substeps below:	MLVS screen has displays listed below.	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		

Table 1. ALR-67(V)2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
(2) If not in UP position, place UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step c.(3) until proper RWR program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.67 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The displays in the step below come and go quickly.		

Table 1. ALR-67(V)2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(5) Press and hold EXEC for 3 seconds.	MLVS screen displays the below: FILE:n ffffffff.67 Waiting 03 Waiting 02 Waiting 01 FILE:n ffffffff.67 CFG: xxx xxx LOAD ?	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE CFG information indicates the file that exists in the system. If the loaded file is the correct file the upload may be aborted by holding the MLVS EXEC to SKIP for 3 seconds.		
(6) Monitor CFG information on MLVS display. If existing CFG file number is the correct file, abort upload by pressing and holding EXEC switch to SKIP for 3 seconds and going to step 3.		
(7) Momentarily press EXEC.	1. Forward Azimuth Indicator displays MLV ACTIVE during the load/verify portion of the load process. 2. MLVS screen displays the below: FILE:n ffffffff.67 Upload xxxx yyyy FILE:n ffffffff.67 Verify xxxx yyyy	
NOTE A "/" will rotate clockwise to indicate burn in is in progress.		
	Burn in / XX XX FILE:n ffffffff.67 Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 1. ALR-67(V)2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(8) Momentarily press EXEC switch to SKIP.	MLVS screen displays the below: FILE:n ffffffff.67 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
d. On GND PWR control panel assembly, set and hold 3 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On Control-Indicator, press and release POWER ON switch.	1. Control-Indicator buttons light up with descriptions. 2. Power button displays ALR-67 in red and ON in green.	Do table 1, (A1-F18AE-760-200, WP045 00).
f. On Control-Indicator, press the POWER ON switch to turn off ALR-67 power.	1. Control Indicator buttons disappear. 2. Forward Azimuth Indicator display disappears.	
3. SHUTDOWN.		
a. On GND PWR control panel assembly, set 3 switch to AUTO.		
b. If system OFP loading is complete, on MLVS set PWR switch to OFF.		
c. Remove electrical power (A1-F18AC-LMM-000).		
d. On MLVS, do substeps below:		
(1) Disconnect data cable W3 connector P1 from connector J2.		
(2) Disconnect power cable W2 connector P1 from connector J1.		

Table 1. ALR-67(V)2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
e. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable W3 connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable W2 connector P2 from utility power receptacle (1J-G089).		

Table 2. ALR-67(V)2 Stand Alone Verify Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Radar Warning Computer CP-1239C/ALR-67(V)2 **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY		

Table 2. ALR-67(V)2 Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
NOTE To verify that the cable is properly seated when connecting flex-type cables make sure that red line on the MLVS connector is not visible.		
a. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect data cable W3 connector P1 to connector J2. b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable W2 connector P2 to utility power receptacle (1J-G089). (2) Connect data cable W3 connector P2 to MUX test connector (83J-G003). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		

Table 2. ALR-67(V)2 Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On MLVS, set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
d. On GND PWR control panel assembly, set and hold 3 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On the Control Indicator, press the POWER ON switch.	Observe the displays listed below: 1. Control-Indicator buttons light up with descriptions. 2. Power button displays ALR-67 in red and ON in green.	Do table 1, (A1-F18AE-760-200, WP045 00).
f. On MLVS: (1) If not in VRFY position, Place UP/VRFY/DOWN switch in the VRFY position.	″MLV ACTIVE″ display on.	

Table 2. ALR-67(V)2 Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(2) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Verify	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(3) Repeat step f.(2) until proper RWR program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.67 EXEC to Verify	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The displays in the step below come and go quickly.		
(4) Press and hold EXEC for 3 seconds.	MLVS screen displays the below: FILE:n ffffffff.67 Waiting 03 Waiting 02 Waiting 01 FILE:n ffffffff.67 Verify xxxx yyyy FILE:n ffffffff.67 Verify Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(5) On Control-Indicator, press the POWER ON switch to turn off ALR-67 power.	Observe the displays listed below: 1. Control-Indicator buttons disappear. 2. Forward Azimuth Indicator display goes out.	
3. SHUTDOWN. a. On GND PWR control panel assembly, set 3 switch to AUTO. b. If system OFP loading is complete, on MLVS set PWR switch to OFF.		

Table 2. ALR-67(V)2 Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. Remove electrical power (A1-F18AC-LMM-000). d. On MLVS, do substeps below: (1) Disconnect data cable W3 connector P1 from connector J2. (2) Disconnect power cable W2 connector P1 from connector J1. e. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable W3 connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable W2 connector P2 from utility power receptacle (1J-G089).		

Table 3. ALQ-126B Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Receiver Transmitter RT-1079/ALQ-126 **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None		

Table 3. ALQ-126B Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
NOTE To verify that the cable is properly seated when connecting flex-type cables make sure that red line on the MLVS connector is not visible.		
a. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect data cable W3 connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable W2 connector P2 to utility power receptacle (1J-G089). (2) Connect data cable W3 connector P2 to MUX test connector (83J-G003).		
2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

Table 3. ALQ-126B Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
c. On MLVS, set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
d. On GND PWR control panel assembly, set and hold 3 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).

Table 3. ALQ-126B Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
e. On ECM control panel assembly, set the ECM mode switch to STBY.	In cockpit, on LH advisory and threat warning indicator panel, STBY comes on for 3 to 4 minutes and then goes out.	1. ON F/A-18C, if STBY light did not come on, do table 1 (A1-F18AE-760-200, WP016 00). 2. ON F/A-18D, if STBY light did not come on, do table 1 (A1-F18AE-760-200, WP016 01). 3. ON F/A-18C, if STBY light did not go off, do table 2 (A1-F18AE-760-200, WP016 00). 4. ON F/A-18D, if STBY light did not go off, do table 2 (A1-F18AE-760-200, WP016 00).
f. On MLVS: (1) If not in UP position, place UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(2) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(3) Repeat step f.(2) until proper ALQ-126B program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.126 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The displays in the step below come and go quickly.		

Table 3. ALQ-126B Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(4) Press and hold EXEC for 3 seconds.	MLVS screen displays the below: FILE:n ffffffff.126 Waiting 03 Waiting 02 Waiting 01 FILE:n ffffffff.126 CFG: xxxx xxxx LOAD ?	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(5) Momentarily press EXEC switch to EXEC.	MLVS screen displays the below: FILE:n ffffffff.126 Waiting 03 Waiting 02 Waiting 01 FILE:n ffffffff.126 Upload xxxx yyyy FILE:n ffffffff.126 Upload Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(6) Momentarily press EXEC switch to SKIP.	MLVS screen displays the below: FILE:nfffffffff.126 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(7) On ECM Control Panel Assembly, set ECM switch to OFF.	All lights off.	Replace ECM control panel assembly (A1-F18AE-760-300, WP007 00).
3. SHUTDOWN.		
a. On GND PWR control panel assembly, set 3 switch to AUTO.		
b. If system OFP loading is complete, on MLVS set PWR switch to OFF.		
c. Remove electrical power (A1-F18AC-LMM-000).		
d. On MLVS, do substeps below:		
(1) Disconnect data cable W3 connector P1 from connector J2.		

Table 3. ALQ-126B Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Disconnect power cable W2 connector P1 from connector J1. e. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable W3 connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable W2 connector P2 from utility power receptacle (1J-G089).		

Table 4. ALQ-126B Stand Alone Verify Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Receiver-Transmitter RT-1079/ALQ-126

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/USQ-131	Memory Loader-Verifier Set

Materials Required

None

NOTE

For Component Locator, refer to WP005 00.

For Test Equipment Hookup, refer to WP007 00.

Table 4. ALQ-126B Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
NOTE To verify that the cable is properly seated when connecting flex-type cables make sure that red line on the MLVS connector is not visible.		
a. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect data cable W3 connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable W2 connector P2 to utility power receptacle (1J-G089). (2) Connect data cable W3 connector P2 to MUX test connector (83J-G003).		
2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

Table 4. ALQ-126B Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
c. On MLVS, set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right. (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
d. On GND PWR control panel assembly, set and hold 3 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).

Table 4. ALQ-126B Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
e. On ECM control panel assembly, set the ECM mode switch to STBY.	In cockpit, on LH advisory and threat warning indicator panel, STBY comes on for 3 to 4 minutes and then goes out.	1. ON F/A-18C, if STBY light did not come on, do table 1 (A1-F18AE-760-200, WP016 00). 2. ON F/A-18D, if STBY light did not come on, do table 1 (A1-F18AE-760-200, WP016 01). 3. ON F/A-18C, if STBY light did not go off, do table 2 (A1-F18AE-760-200, WP016 00). 4. ON F/A-18D, if STBY light did not go off, do table 2 (A1-F18AE-760-200, WP016 00).
f. On MLVS: (1)If not in VRFY position, place UP/VRFY/DOWN switch in the VRFY position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry). After EXEC is pressed, either ″MODE NOT ALLOWED″ or ″EXEC TO VERIFY″ may be displayed on MLVS second line of text.		
(2) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Verify	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(3) Repeat step f.(2) until proper UDF file is displayed.	MLVS screen displays the below: FILE:n ffffffff.126 EXEC to Verify	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The displays in the step below come and go quickly.		

Table 4. ALQ-126B Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(4) Press and hold EXEC for 3 seconds.	MLVS screen displays the below: FILE:n ffffffff.126 Waiting 03 Waiting 02 Waiting 01 FILE:n ffffffff.126 CFG: xxxx xxxx Load ? FILE:n ffffffff.126 Verify Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(5) Momentarily press EXEC switch to EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx Waiting 03 Waiting 02 Waiting 01 FILE:n ffffffff.xxx Verify xxxx yyyy FILE:n ffffffff.xxx Verify Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(6) Momentarily press EXEC switch to SKIP.	MLVS screen displays the below: FILE:nfffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(7) On ECM Control Panel assembly, set ECM switch to OFF.	All lights off.	Replace ECM control panel assembly (A1-F18AE-760-300, WP007 00).
3. SHUTDOWN.		
a. On GND PWR control panel assembly, set 3 switch to AUTO.		
b. If system OFP loading is complete, on MLVS, set PWR switch to OFF.		
c. Remove electrical power (A1-F18AC-LMM-000).		
d. On MLVS, do substeps below:		
(1) Disconnect data cable W3 connector P1 from connector J2.		

Table 4. ALQ-126B Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Disconnect power cable W2 connector P1 from connector J1. e. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable W3 connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable W2 connector P2 from utility power receptacle (1J-G089).		

Table 5. ALE-47 Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication

System Required Components

Countermeasures Dispensing System Programmer CD-45/ALE-47

Related Systems Required

Avionics Cooling System
Electrical System

Support Equipment Required

Part Number or Type Designation	Nomenclature
AN/USQ-131	Memory Loader-Verifier Set

Materials Required

None

NOTE

For Component Locator, refer to WP005 00.

For Test Equipment Hookup, refer to WP007 00.

Table 5. ALE-47 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
1. PRELIMINARY		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
NOTE To verify that the cable is properly seated when connecting flex-type cables make sure that red line on the MLVS connector is not visable.		
a. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect data cable W3 connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable W2 connector P2 to utility power receptacle (1J-G089). (2) Connect data cable W3 connector P2 to MUX test connector (83J-G003).		
2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

Table 5. ALE-47 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On GND PWR control panel assembly, set and hold 3 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
d. On ECM control panel, set DISPENSER switch to ON.		

NOTE

MLVS display is made up of two lines of text. The steps below indicate text display for each step.

e. On MLVS, set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

NOTE

The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.

(4) 1 MMDDYY 2 MMDDYY EXEC to continue	

NOTE

See WP003 00 to verify program load CONFIG/IDENT number.

For an operational ALE-47, both operational program (OFP) and Mission Data (MDF) files must be loaded. OFP file must be loaded before MDF file.

Table 5. ALE-47 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) If OFP is to be loaded into ALE-47, do step 2..f.(1). If MDF is to be loaded into ALE-47, do step 2.g.(1).		
f. On MLVS:		
(1) If not in UP position, place UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(2) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(3) Repeat step f.(2) until proper ALE-47 program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.47 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The displays in the step below come and go quickly.		
(4) Press and hold EXEC for 3 seconds.	MLVS screen displays the below: Waiting 02 Waiting 01 FILE:n ffffffff.47 CFG: xxxx xxxx Load ?	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 5. ALE-47 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(5) Momentarily press EXEC switch to EXEC.	FILE:n ffffffff.47 Waiting 04 Waiting 03 Waiting 02 Waiting 01 FILE:n ffffffff.47 Upload XXXX FILE:n ffffffff.47 Verify XXXX FILE:n ffffffff.47 Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(6) Momentarily press EXEC switch to SKIP.	MLVS screen displays the below: FILE:n ffffffff.47 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(7) If MDF is to be loaded into ALE-47, do step 2.g.(1). If not , go to step 2.h.		
NOTE OFP file must be loaded before MDF file.		
g. On MLVS:		
(1) If not in UP position, place UP/VRFY/DOWN switch in the UP position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(2) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).

Table 5. ALE-47 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(3) Repeat step g.(2) until proper ALE-47 program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.47 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The displays in the step below come and go quickly.		
(4) Press and hold EXEC for 3 seconds.	MLVS screen displays the below: Waiting 02 Waiting 01 FILE:n ffffffff.47 CFG: xxxx xxxx Load ?	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(5) Momentarily press EXEC switch to EXEC.	FILE:n ffffffff.47 Waiting 04 Waiting 03 Waiting 02 Waiting 01 FILE:n ffffffff.47 Upload XXXX FILE:n ffffffff.47 Verify XXXX FILE:n ffffffff.47 Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(6) Momentarily press EXEC switch to SKIP.	MLVS screen displays the below: FILE:n ffffffff.47 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
h. On ECM control panel, set DISPENSER switch to OFF.		
3. SHUTDOWN.		
a. On GND PWR control panel assembly, set 3 switch to AUTO.		
b. If system OFP loading is complete, on MLVS, set PWR switch to OFF.		

Table 5. ALE-47 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. Remove electrical power (A1-F18AC-LMM-000). d. On MLVS, do substeps below: (1) Disconnect data cable W3 connector P1 from connector J2. (2) Disconnect power cable W2 connector P1 from connector J1. e. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable W3 connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable W2 connector P2 from utility power receptacle (1J-G089).		

Table 6. ALQ-165 Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Airborne Self Protect Jammer AN/ALQ-165 **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None		

Table 6. ALQ-165 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY.		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors.		
NOTE To verify that the cable is properly seated when connecting flex-type cables make sure that red line on the MLVS connector is not visible.		
a. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect data cable W3 connector P1 to connector J2.		
b. In aircraft nose wheelwell, do substeps below: (1) Connect power cable W2 connector P2 to utility power receptacle (1J-G089). (2) Connect data cable W3 connector P2 to MUX test connector (83J-G003).		
2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000).		
b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

Table 6. ALQ-165 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On GND PWR control panel assembly, set and hold 3 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
d. On ECM control panel assembly, set the ECM mode switch to STBY.	In cockpit, on LH advisory and threat warning indicator panel, STBY illuminates for 3 to 4 minutes and then goes out.	1. ON F/A-18C AND F/A-18D 163434 THRU 163778, if STBY light did not come on, do table 1 (A1-F18AE-760-200, WP019 00). 2. ON F/A-18D 163986 AND UP, if STBY light did not come on, do table 2 (A1-F18AE-760-200, WP016 01). 3. ON F/A-18C AND F/A-18D 163434 THRU 163778, if STBY light did not go off, do table 1 (A1-F18AE-760-200, WP018 00). 4. ON F/A-18D 163986 AND UP, if STBY light did not go off, do table 1 (A1-F18AE-760-200, WP018 00).
e. On MLVS, do substeps below:	MLVS screen has displays listed below.	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		

Table 6. ALQ-165 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVSRev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
	NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.	
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
	NOTE See WP003 00 to verify program load CONFIG/IDENT number.	
(2) If not in up position, place UP/VRFY/DOWN switch in the UP position.		
	NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).	
(3) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(4) Repeat step e.(3) until proper ALQ-165 program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.165 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
	NOTE The displays in the following step come and go quickly.	

Table 6. ALQ-165 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(5) Press and hold EXEC for 3 seconds.	MLVS screen displays the below: FILE:n ffffffff.165 Waiting 05 Waiting 04 Waiting 03 Waiting 02 Waiting 01 FILE:n ffffffff.165 CFG: xxxx xxxx LOAD ?	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(6) Momentarily press EXEC switch to EXEC.	MLVS screen displays the below: FILE:n ffffffff.165 Waiting 05 Waiting 04 Waiting 03 Waiting 02 Waiting 01 FILE:n ffffffff.165 Upload xxx yyy FILE:n ffffffff.165 Verify xxxx yyyy FILE:n ffffffff.165 Upload verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(7) Momentarily press EXEC switch to SKIP.	MLVS screen displays the below: FILE:nfffffffff.165 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
3. SHUTDOWN.		
a. On ECM control panel assembly, set ECM switch to OFF.		
b. On GND PWR control panel assembly, set 3 switch to AUTO.		
c. If system OFP loading is complete, on MLVS set PWR switch to OFF.		
d. Remove electrical power (A1-F18AC-LMM-000).		
e. On MLVS, do substeps below:		

Table 6. ALQ-165 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Disconnect data cable W2 connector P1 from connector J2. (2) Disconnect power cable W2 connector P1 from connector J1. f. In aircraft nose wheelwell, do substeps below: (1) Disconnect data cable W3 connector P2 from MUX test connector (83J-G003). (2) Disconnect power cable W2 connector P2 from utility power receptacle (1J-G089).		

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

EW LOAD/VERIFICATION PROCEDURES USING AN/USQ-131 LOADER-VERIFIER SET

EFFECTIVITY: F/A-18A AND F/A-18B

Reference Material

Airborne Weapons/Stores Loading Manual	A1-F18AE-LWS-000
Line Maintenance Procedures	A1-F18AC-LMM-000
Multipurpose Display Group	A1-F18AC-745-200
Displays Test F/A-18A	WP004 00
Displays Test F/A-18B	WP005 00
Software Configuration Manual	A1-F18AC-SCM-000
Program Load Versions	WP003 00
Program Load CONFIG/IDENT Verification	WP004 00
Component Locator	WP005 00
Test Equipment Hookup Locator	WP007 00
Memory Loader-Verifier Set AN/USQ-131	NAVAIR 16-30USQ131-1
Extended BIT	WP003 00

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Record of Applicable Technical Directives

None

1. INTRODUCTION.

2. This work package includes procedures for loading operational flight programs (OFP) using the Memory Loader-Verifier Set AN/USQ-131 (MLVS).

3. Each programmable WRA has a separate table. The WRAs include:

a. Radar Warning Receiver CP-1239C/ALR-67(V)2 (RWR)

b. Receiver Transmitter RT-1079/ALQ-126B (ECS)

Table 1. ALR-67(V)2 Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Radar Warning Computer CP-1239C/ALR-67(V)2 **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY. a. Open doors 13L and 14R (A1-F18AC-LMM-010).		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors. **NOTE** To verify that the cable is properly seated when connecting flex-type cables make sure that red line on the MLVS connector is not visible.		

Table 1. ALR-67(V)2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. On ALR-67(V)2 do substeps below: (1) Disconnect cable from connector J2. (2) Connect W1J1 of tee cable to the cable disconnected from J2 in step b. (1). (3) Connect P2 of tee cable to connector J2. c. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect MLVS tee cable P1 to connector J2. d. In aircraft nose wheelwell, connect power cable W3 connector P2 to utility power receptacle (1J-G089). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		

Table 1. ALR-67(V)2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On MLVS, set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
d. On GND PWR control panel assembly, set and hold 3 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).

Table 1. ALR-67(V)2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
e. On Control-Indicator, press and release POWER ON switch.	1. POWER ON light comes on.	Do table 1, (A1-F18AC-760-200, WP033 00).
	2. BIT, OFFSET, DISPLAY, and SPECIAL lights come on.	Replace Control-Indicator C-10250/ALR-67(V) (A1-F18AC-760-300, WP056 00).
	3. Forward Azimuth Indicator has status and emitter display.	1. No display on Azimuth Indicator. Do table 2, (A1-F18AC-760-200, WP033 00). 2. Priority display cycles A to N continuously. Do substeps below: a. Remove Radar Receiver R-2055A/ALR-67(V) (A1-F18AC-760-300, WP054 00). b. On Radar Receiver, if circuit breaker CB1 is in OFF position (tripped), do step c. If CB1 is in ON position, do table 5, (A1-F18AC-760-200, WP034 00). c. Reset CB1 and install Radar Receiver R-2055A/ALR-67(V) (A1-F18AC-760-300, WP054 00). If malfunction still exists, do table 5, (A1-F18AC-760-200, WP034 00). 3. Azimuth Indicator displays flashing B. Do table 5, (A1-F18AC-760-200, WP033 00).
	4. On F/A-18B, rear Azimuth Indicator has status and emitter display.	1. No display on rear Azimuth Indicator. Do table 5, (A1-F18AC-760-200, WP033 00). 2. Rear Azimuth Indicator displays flashing B. Do table 5, (A1-F18AC-760-200, WP033 00).
f. On MLVS: (1) Place UP/VRFY/DOWN switch in the UP position.		

Table 1. ALR-67(V)2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(2) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(3) Repeat step f.(2) until proper RWR program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.67 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The displays in the following step come and go quickly.		
(4) Press and hold EXEC for 3 seconds.	MLVS screen displays the below: FILE:n ffffffff.67 Waiting 03 Waiting 02 Waiting 01 FILE:n ffffffff.67 CFG: xxxx xxxx LOAD ?	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE CFG information indicates the current file already exists in the system. If the loaded file is the correct file the upload may be aborted by holding the MLVS EXEC		
(5) Monitor CFG information on MLVS display. If existing CFG file number is the correct file, abort upload by pressing and holding EXEC switch to SKIP for 3 seconds and go to step 3.		

Table 1. ALR-67(V)2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(6) Momentarily press EXEC.	Azimuth Indicator displays "MLV ACTIVE", MLVS screen displays the below: FILE:n ffffffff.67 Upload XXXX YYYY FILE:n ffffffff.67 Verify XXXX YYYY Burn in XXXX FILE:n ffffffff.67 Upload Verified	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(7) Momentarily press EXEC switch to SKIP.	MLVS screen displays the below: FILE:n ffffffff.67 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(8) On Control-Indicator, press the POWER ON switch to turn off ALR-67 power.	1. Control-Indicator buttons disappear. 2. Forward Azimuth Indicator display goes off.	
3. SHUTDOWN. a. On GND PWR control panel assembly, set 3 switch to AUTO. b. If system OFP loading is complete, on MLVS, set PWR switch to OFF. c. Remove electrical power (A1-F18AC-LMM-000). d. In aircraft nose wheelwell, disconnect power cable W2 connector P2 from utility power receptacle (1J-G089). e. On MLVS, do substeps below:		

Table 1. ALR-67(V)2 Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(1) Disconnect MLVS tee cable W1 connector P1 from connector J2. (2) Disconnect power cable W2 connector P1 from connector J1. f. On ALR-67(V)2, do the substeps below: (1) Disconnect MLVS tee cable W1 connector J1 from the ALR-67 cable. (2) Disconnect MLVS tee cable W1 connector P2 from ALR-67 connector J2. (3) Reconnect ALR-67 cable to ALR-67 connector J2. g. Close doors 13R and 14L (A1-F18AC-LMM-010).		

Table 2. ALR-67(V)2 Stand Alone Verify Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Radar Warning Computer CP-1239C/ALR-67(V)2 **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY. a. Open doors 13L and 14R (A1-F18AC-LMM-010).		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors. **NOTE** To verify that the cable is properly seated when connecting flex-type cables make sure that red line on the MLVS connector is not visible.		

Table 2. ALR-67(V)2 Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. On ALR-67(V)2, do substeps below: (1) Disconnect cable from connector J2. (2) Connect W1J1 of the tee cable to the cable disconnected from J2 in step b. (1). (3) Connect P2 of tee cable to connector J2. c. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect MLVS tee cable P1 to connector J2. d. In aircraft nose wheelwell, connect power cable W2 connector P2 to utility power receptacle (1J-G089). 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		

Table 2. ALR-67(V)2 Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
c. On MLVS, set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
NOTE See WP003 00 to verify program load CONFIG/IDENT number.		
d. On GND PWR control panel assembly, set and hold 3 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).

Table 2. ALR-67(V)2 Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
e. On Control-Indicator, press and release POWER ON switch.	1. POWER ON light comes on.	Do table 1, (A1-F18AC-760-200, WP033 00).
	2. BIT, OFFSET, DISPLAY, and SPECIAL lights come on.	Replace Control-Indicator C-10250/ALR-67(V) (A1-F18AC-760-300, WP056 00).
	3. Forward Azimuth Indicator has status and emitter display.	1. No display on Azimuth Indicator. Do table 2, (A1-F18AC-760-200, WP033 00). 2. Priority display cycles A to N continuously. Do substeps below: a. Remove Radar Receiver R-2055A/ALR-67(V) (A1-F18AC-760-300, WP054 00). b. On Radar Receiver, if circuit breaker CB1 is in OFF position (tripped), do step c. If CB1 is in ON position, do table 5, (A1-F18AC-760-200, WP034 00). c. Reset CB1 and install Radar Receiver R-2055A/ALR-67(V) (A1-F18AC-760-300, WP054 00). If malfunction still exists, do table 5, (A1-F18AC-760-200, WP034 00). 3. Azimuth Indicator displays flashing B. Do table 5, (A1-F18AC-760-200, WP033 00).
	4. On F/A-18B, rear Azimuth Indicator has status and emitter display.	1. No display on rear Azimuth Indicator. Do table 5, (A1-F18AC-760-200, WP033 00). 2. Rear Azimuth Indicator displays flashing B. Do table 5, (A1-F18AC-760-200, WP033 00).
f. On MLVS: (1) If not in VRFY position, place UP/VRFY/DOWN switch in the VRFY position.		

Table 2. ALR-67(V)2 Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(2) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Verify	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(3) Repeat step f.(2) until proper RWR program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.67 EXEC to Verify	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The displays in the following step come and go quickly.		
(4) Press and hold EXEC for 3 seconds.	MLVS screen displays the below: FILE:n ffffffff.67 Waiting 03 Waiting 02 Waiting 01 FILE:n ffffffff.67 Verify xxxx yyyy FILE:n ffffffff.67 Verify Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(5) On Control-Indicator, press the POWER ON switch to turn off ALR-67 power.	1. Control-Indicator buttons disappear. 2. Forward Azimuth Indicator display goes off.	
3. SHUTDOWN. a. On GND PWR control panel assembly, set 3 switch to AUTO. b. If system OFP loading is complete, on MLVS set PWR switch to OFF. c. Remove electrical power (A1-F18AC-LMM-000).		

Table 2. ALR-67(V)2 Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
d. In aircraft nose wheelwell, disconnect power cable W2 connector P2 from utility power receptacle (1J-G089). e. On MLVS, do substeps below: (1) Disconnect MLVS tee cable W1 connector P1 from connector J2. (2) Disconnect power cable W2 connector P1 from connector J1. f. On ALR-67(V)2, do the substeps below: (1) Disconnect MLVS tee cable W1 connector J1 from the ALR-67 cable. (2) Disconnect MLVS tee cable W1 connector P2 from ALR-67 connector J2. (3) Reconnect ALR-67 cable to ALR-67 connector J2. g. Close doors 13R and 14L (A1-F18AC-LMM-010).		

Table 3. ALQ-126B Load/Verification Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Receiver-Transmitter RT-1079/ALQ-126B **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY. a. Open doors 13L and 14R (A1-F18AC-LMM-010).		
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors. **NOTE** To verify that the cable is properly seated when connecting flex-type cables make sure that red line on the MLVS connector is not visible.		

Table 3. ALQ-126B Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
b. On ALR-67(V)2 do substeps below: (1) Disconnect cable from connector J2. (2) Connect W1J1 of the tee cable to the cable disconnected from J2 in step b.(1). (3) Connect P2 of tee cable W1 to connector J2. c. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect MLVS tee cable W1 connector P1 to connector J2. d. In aircraft nose wheelwell, connect power cable W2 connector P2 to utility power receptacle (1J-G089). e. On ALQ-126B, do substeps below: (1) Disconnect connector 3J21. (2) Connect ALQ-126B jumper plug to connector 3J21. 2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		

Table 3. ALQ-126B Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
c. On MLVS, set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
d. On GND PWR control panel assembly, set and hold 3 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).
e. On the ECM Control Panel, set the ECM switch to STBY.	On LH advisory and threat warning indicator panel, STBY comes on for 3 to 4 minutes and then goes off.	1. If light did not come on, do table 1 (A1-F18AC-760-200, WP016 00). 2. If light did not go off, do table 2 (A1-F18AC-760-200, WP016 00).
f. On MLVS: (1) If not in UP position, place UP/VRFY/DOWN switch in the UP position.		

Table 3. ALQ-126B Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(2) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(3) Repeat step f.(2) until proper ALQ-126 UDF program load CONFIG/IDENT number is displayed.	MLVS screen displays the below: FILE:n ffffffff.126 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The displays in the following step come and go quickly.		
(4) Press and hold EXEC for 3 seconds.	MLVS screen displays the below: FILE:n ffffffff.126 Waiting 03 Waiting 02 Waiting 01 FILE:nfffffffff.126 CFG: xxx xxx LOAD ?	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(5) Momentarily press EXEC switch to EXEC.	MLVS screen displays the below: FILE:n ffffffff.126 Waiting 03 Waiting 02 Waiting 01 FILE:nfffffffff.126 Upload Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(6) Momentarily press EXEC switch to SKIP.	MLVS screen displays the below: FILE:nfffffffff.126 EXEC to Upload	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(7) In cockpit, on ECM Control Panel Assembly, set ECM switch to OFF. 4. SHUTDOWN.		

Table 3. ALQ-126B Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
a. On GND PWR control panel assembly, set 3 switch to AUTO. b. If system OFP loading is complete, on MLVS set PWR switch to OFF. c. Remove electrical power (A1-F18AC-LMM-000). d. In aircraft nose wheelwell, disconnect power cable W2 connector P2 from utility power receptacle (1J-G089). e. On MLVS, do substeps below: (1) Disconnect MLVS tee cable W1 connector P1 from connector J2. (2) Disconnect power cable W2 connector P1 from connector J1. f. On ALR-67(V)2, do the substeps below: (1) Disconnect MLVS tee cable W1 connector J1 from the ALR-67 cable. (2) Disconnect MLVS tee cable W1 connector P2 from ALR-67 connector J2. (3) Reconnect ALR-67 cable to ALR-67 connector J2. g. Close door 13R (A1-F18AC-LMM-010). h. On ALQ-126B, do the substeps below: (1) Disconnect jumper plug from connector 3J21.		

Table 3. ALQ-126B Load/Verification Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Reconnect ALQ-126B cable to connector 3J21. i. Close door 14L (A1-F18AC-LMM-010).		

Table 4. ALQ-126B Stand Alone Verify Procedure

Procedure	Normal Indication	Remedy for Abnormal Indication
System Required Components Receiver-Transmitter RT-1079/ALQ-126B **Related Systems Required** Avionics Cooling System Electrical System **Support Equipment Required** **Part Number or Type Designation** — **Nomenclature** AN/USQ-131 — Memory Loader-Verifier Set **Materials Required** None **NOTE** For Component Locator, refer to WP005 00. For Test Equipment Hookup, refer to WP007 00.		
1. PRELIMINARY. a. Open doors 13L and 14R (A1-F18AC-LMM-010).		

Table 4. ALQ-126B Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
CAUTION To prevent damage to connector pins when connecting MLVS cables, visually line up connector keys with mating keyways before mating connectors. **NOTE** To verify that the cable is properly seated when connecting flex-type cables make sure that red line on the MLVS connector is not visible.		
b. On ALR-67(V)2, do substeps below: (1) Disconnect cable from connector J2. (2) Connect W1J1 of the tee cable to the cable disconnected from J2 in step b.(1). (3) Connect P2 of tee cable W1 to connector J2. c. On MLVS, do substeps below: (1) Connect power cable W2 connector P1 to connector J1. (2) Connect MLVS tee cable W1 connector P1 to connector J2. d. In aircraft nose wheelwell, connect power cable W2 connector P2 to utility power receptacle (1J-G089). e. On ALQ-126B, do substeps below: (1) Disconnect connector 3J21. (2) Connect ALQ-126B jumper plug to connector 3J21.		

Table 4. ALQ-126B Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
2. PROCEDURE. a. Apply electrical power (A1-F18AC-LMM-000). b. On GND PWR control panel assembly, set EXT PWR switch to RESET.		
NOTE MLVS display is made up of two lines of text. The steps below indicate text display for each step.		
c. On MLVS, set PWR switch to ON.	On MLVS, display screen displays the below in sequence: (1) two rows of solid blocks sweeping left to right (2) Basic BIT Success All Tests Passed (3) MLVS Rev X.XX MMM DD HH:MM:SS	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The display below is in the format MMDDYY = the date of memory card currently installed in slot 1 and slot 2. If no memory card is in slot 1 or slot 2 the message 1 Empty and/or 2 Empty is displayed.		
	(4) 1 MMDDYY 2 MMDDYY EXEC to continue	
d. On GND PWR control panel assembly, set and hold 3 switch to A ON for 3 seconds.	Switch remains on (latched).	1. If switch unlatches in 10 to 30 seconds, apply external cooling air to aircraft (A1-F18AC-LMM-000). 2. If switch does not remain on, do Ground Power Switching System Test (A1-F18AC-420-200, WP006 00).

Table 4. ALQ-126B Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
e. On the ECM Control Panel, set the ECM switch to STBY.	On LH advisory and threat warning indicator panel, STBY comes on for 3 to 4 minutes and then goes off.	1. If light did not come on, do table 1 (A1-F18AC-760-200, WP016 00). 2. If light did not go off, do table 2 (A1-F18AC-760-200, WP016 00).
f. On MLVS: (1) If not in VRFY position, place UP/VRFY/DOWN switch in the VRFY position.		
NOTE The displays below are in the format: n = slot number, f = program load CONFIG/IDENT number, and x = file extension (optional entry).		
(2) Momentarily press EXEC.	MLVS screen displays the below: FILE:n ffffffff.xxx EXEC to Verify	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(3) Repeat step f.(2) until proper user data file (UDF) is displayed.	MLVS screen displays the below: FILE:n ffffffff.126 EXEC to Verify	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
NOTE The displays in the following step come and go quickly.		

Table 4. ALQ-126B Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(4) Press and hold EXEC for 3 seconds.	MLVS screen displays the below: FILE:n ffffffff.126 Waiting 03 Waiting 02 Waiting 01 CFG: XXXX XXXX LOAD	
(5) Press EXEC switch to EXEC momentarily.	FILE:n ffffffff.126 Waiting 03 Waiting 02 Waiting 01 FILE:n ffffffff.126 Verify xxxx yyyy FILE:n ffffffff.126 Verify Done	Do Extended BIT Procedure (NAVAIR 16-30USQ131-1, WP003 00).
(6) In cockpit, on ECM Control Panel Assembly, set ECM switch to OFF.		
(7) On GND PWR control panel assembly, set GND PWR switch to AUTO.		
4. SHUTDOWN.		
a. On GND PWR control panel assembly, set 3 switch to AUTO.		
b. If system UDF loading is complete, on MLVS, set PWR switch to OFF.		
c. Remove electrical power (A1-F18AC-LMM-000).		
d. In aircraft nose wheelwell, disconnect power cable W2 connector P2 from utility power receptacle (1J-G089).		
e. On MLVS, do substeps below:		
(1) Disconnect MLVS tee cable W1 connector P1 from connector J2.		

Table 4. ALQ-126B Stand Alone Verify Procedure (Continued)

Procedure	Normal Indication	Remedy for Abnormal Indication
(2) Disconnect power cable W2 connector P1 from connector J1. f. On ALR-67(V)2, do the substeps below: (1) Disconnect MLVS tee cable W1 connector J1 from the ALR-67 cable. (2) Disconnect MLVS tee cable W1 connector P2 from ALR-67 connector J2. (3) Reconnect ALR-67 cable to ALR-67 connector J2. g. Close door 13R (A1-F18AC-LMM-010). h. On ALQ-126B do the substeps below: (1) Disconnect jumper plug from connector 3J21. (2) Reconnect ALQ-126B cable to connector 3J21. i. Close door 14L (A1-F18AC-LMM-010).		

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

TEST EQUIPMENT HOOKUP LOCATOR

Title	WP Number
Test Equipment Hookup Locator Using Computer Memory Loader-Verifier Test Set AN/ASM-607(V)5 and Advanced Memory Loader-Verifier Test Set AN/ASM-687	007 01
Test Equipment Hookup Locator Using Memory Loader-Verifier Set AN/USQ-131	007 02

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

TEST EQUIPMENT HOOKUP LOCATOR USING COMPUTER MEMORY LOADER-VERIFIER TEST SET AN/ASM-607(V)5 AND ADVANCED MEMORY LOADER-VERIFIER TEST SET AN/ASM-687

This WP supersedes WP007 01, dated 1 September 1995.

Reference Material

None

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Record of Applicable Technical Directives

None

1. INTRODUCTION.

2. This work package includes test equipment hookup locator used for Load/Verification Procedures Using Computer Memory Loader-Verifier Test Set (MLV) (WP006 01 and WP006 02) and Load/Verification Procedures Using Advanced Memory Loader-Verifier Test Set AN/ASM-687 (AMLV)(WP006 03).

3. REMOVAL OF TAPE TRANSPORT CARTRIDGE.

CAUTION

To prevent foreign object damage to the tape transport cartridge (TTC) and MLV or AMLV TTC reader, the TTC should only be removed/installed in the shop.

4. On the MLV or AMLV front panel, loosen captive fasteners and raise the DATA CARTRIDGE ACCESS cover (figure 1).

5. Push the TTC latch in the direction of the arrow until the latch pops forward.

CAUTION

To prevent damage of TTC or TTC latch do not carry the TTC by the TTC latch.

6. Grasp latch and firmly pull TTC from slot.

7. INSTALLATION OF TAPE TRANSPORT CARTRIDGE.

Support Equipment Required

Part Number or Type Designation	Nomenclature
-	Torque Wrench, 0 to 10 Inch-Pounds

Materials Required

None

CAUTION

To prevent foreign object damage to the tape transport cartridge (TTC) and MLV or AMLV TTC reader, the TTC should only be removed/installed in the shop.

NOTE

The avionic TTC should always be installed in the TTC slot number 1. The EW TTC should always be installed in the TTC slot number 2.

8. On the MLV or AMLV front panel, loosen captive fasteners and raise the DATA CARTRIDGE ACCESS cover (figure 1).

9. With the latch pulled away from the body of the TTC, insert the (avionic TTC into slot 1, EW TTC into slot 2) of the tape transport cartridge dual cradle assembly.

10. Secure the TTC by gently pushing its latch into the locked position.

11. Close DATA CARTRIDGE ACCESS cover (figure 1) and tighten captive fasteners to between 3 and 5 in.lbs.

12. Check for full contact of DATA CARTRIDGE ACCESS cover gasket with front panel. If gap exists do substeps below:

a. Loosen screws in hinge at top of DATA CARTRIDGE ACCESS cover.

b. Adjust DATA CARTRIDGE ACCESS cover for proper contact position (no gaps exist between gasket and front panel).

c. Torque hinge screws in DATA CARTRIDGE ACCESS cover to between 1.75 and 2.25 in.lbs.

13. On AMLV, do AMLV self test procedure, WP006 03.

14. ILLUSTRATED PARTS BREAKDOWN.

15. This illustrated parts breakdown has data required for identifying and ordering parts. The manual introduction has more information on IPB data.

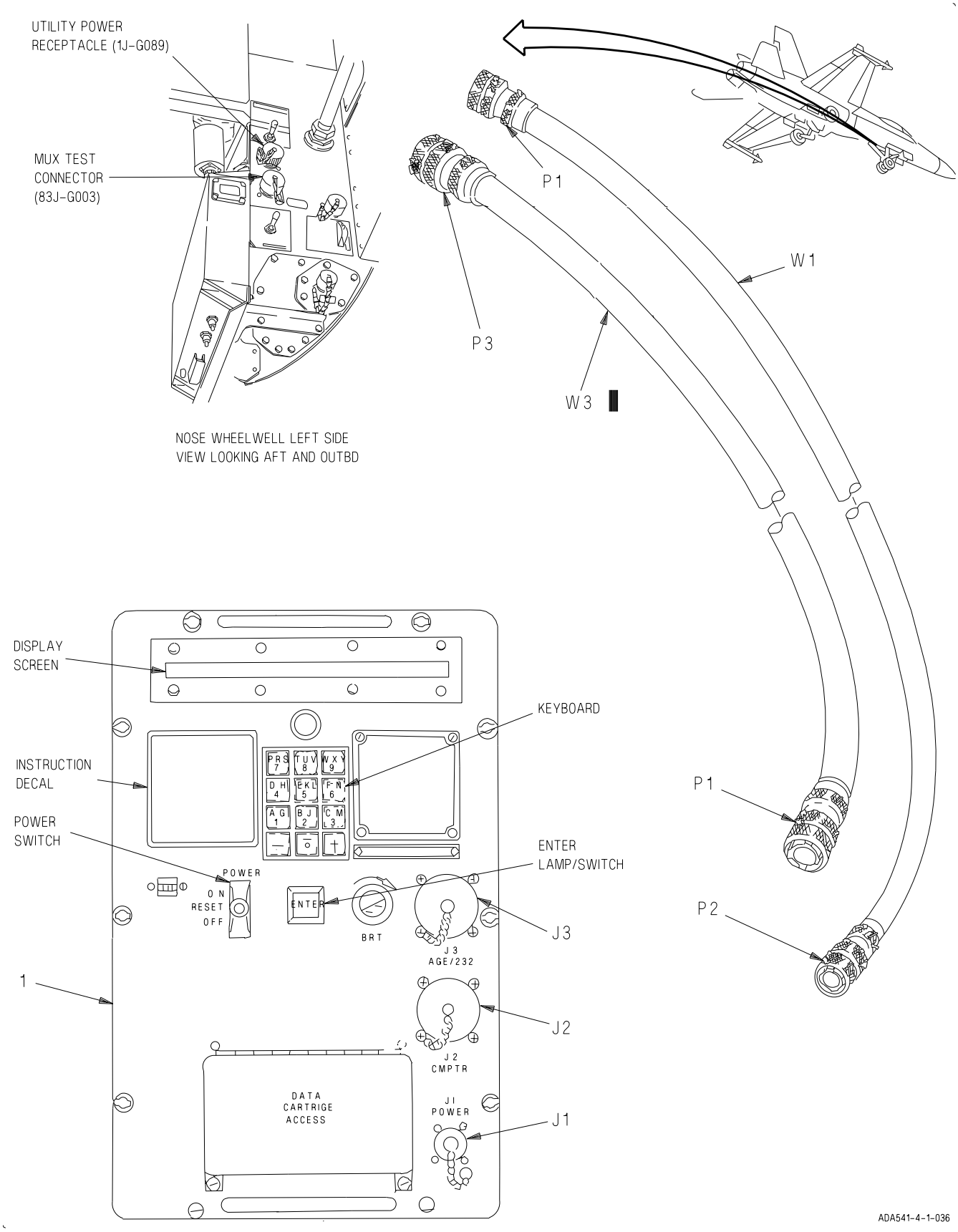


Figure 1. Test Equipment Hookup Locator (Sheet 1)

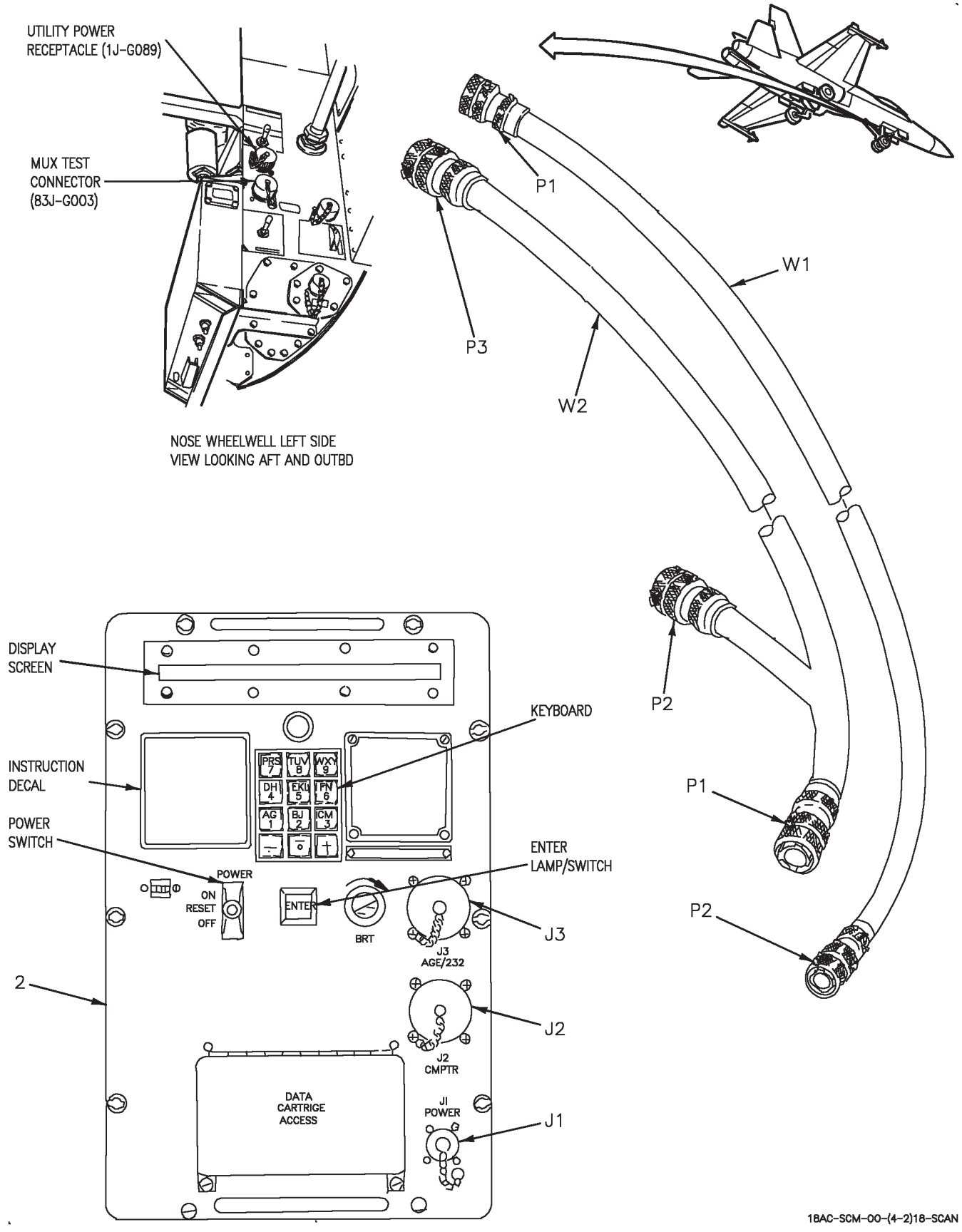
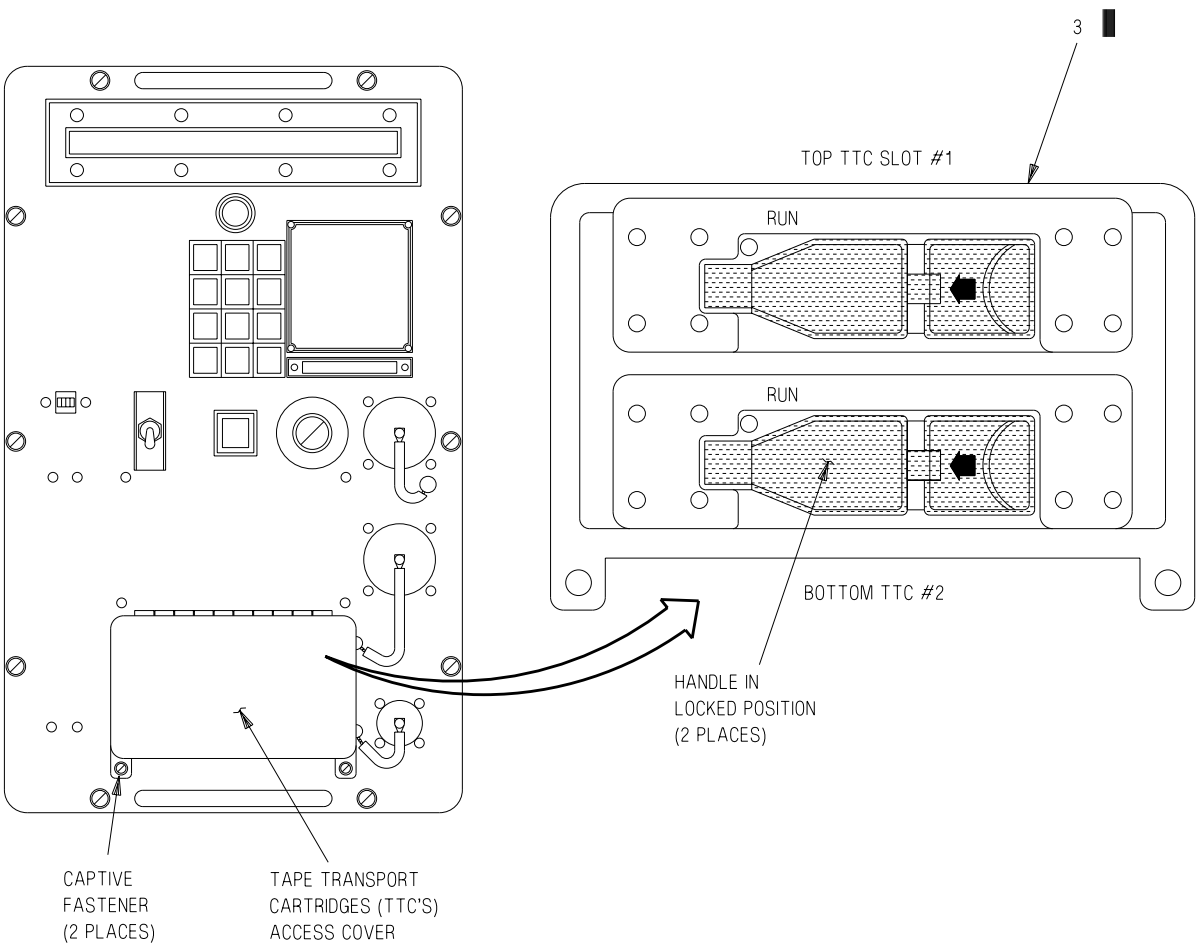
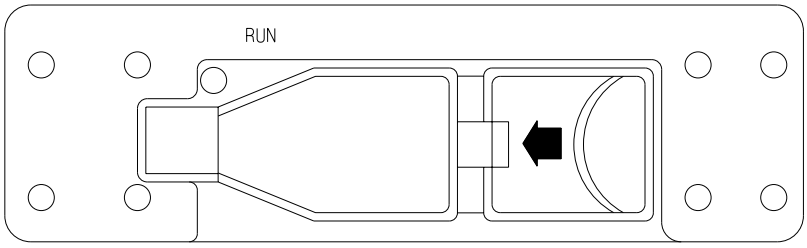
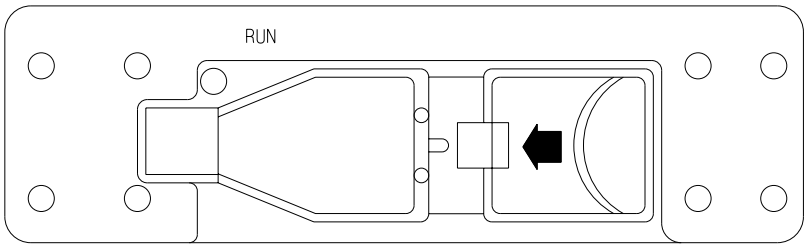


Figure 1. Test Equipment Hookup Locator (Sheet 2)





LATCH CLOSED



LATCH OPEN

ADA541-4-3-036

Figure 1. Test Equipment Hookup Locator (Sheet 3)

INDEX NO.	PART NUMBER	DESCRIPTION	UNITS PER ASSY	USE ON CODE	SM&R CODE
		TEST EQUIPMENT HOOKUP LOCATOR			
1	AN/ASM - 607(V)5	TEST SET, COMPUTER MEMORY LOADER - VERIFIER (96214) (K0289)	1	A	PEOGD
2	AN/ASM - 687	TEST SET, ADVANCED MEMORY LOADER - VERIFIER (96214)	1	B	PEOGD
3	107045-101	DUAL CRADLE ASSY, TAPE TRANSPORT CARTRIDGE (92059)	1		XBGZZ

CODE	USABLE ON	MODEL
A	161353 THRU 163985	F/A-18A/B/C/D
B	163427 AND UP	F/A-18C/D

Figure 1. Test Equipment Hookup Locator (Sheet 4)

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

TEST EQUIPMENT HOOKUP LOCATOR USING MEMORY LOADER-VERIFIER SET AN/USQ-131

Reference Material

None

Alphabetical Index

Subject	Page No.
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Parts List	5
Installation of Memory Card	2
Introduction	1
Removal of Memory Card	1
Test Equipment Hookup Locator, Figure 1	3

Record of Applicable Technical Directives

None

1. INTRODUCTION.

2. This work package includes test equipment hookup locator used for Load/Verification Procedures Using Memory Loader-Verifier Set AN/USQ-131 (MLVS) (WP006 04).

3. REMOVAL OF MEMORY CARD

CAUTION

To prevent foreign object damage to the memory card and MLVS memory card drive the memory card should only be removed/installed in the shop.

4. On the MLVS front panel loosen captive fasteners and raise the memory card access cover (figure 1).

5. Grasp memory card and pull memory card from slot.

6. INSTALLATION OF MEMORY CARD

CAUTION

To prevent foreign object damage to the memory card and MLVS memory card drive the memory card should only be removed/installed in the shop.

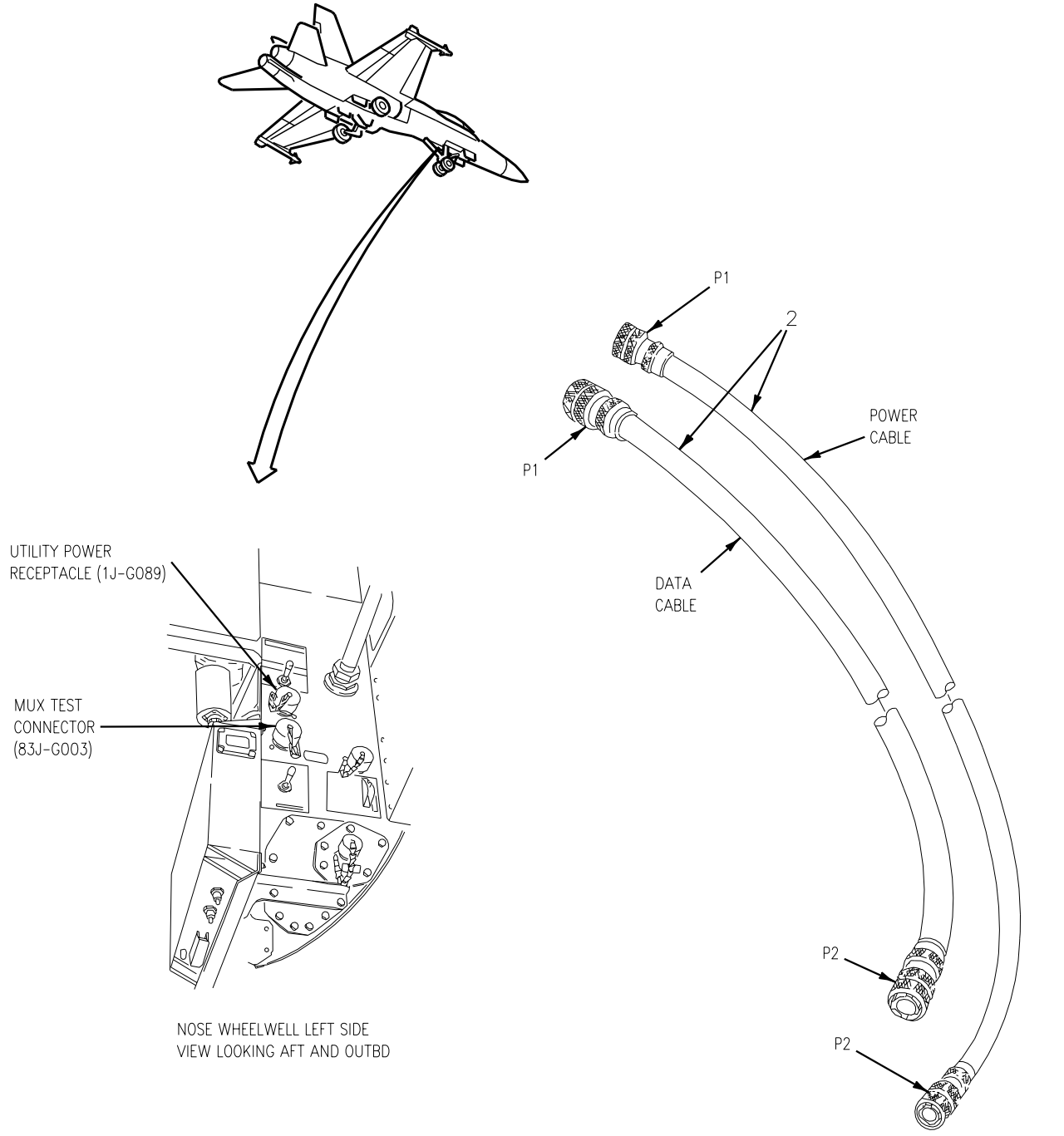
7. On the MLVS front panel, loosen captive fasteners and raise the memory card access cover (figure 1).

8. Insert the memory card(s) into slot 1 and/or slot 2 of the MLVS.

9. Close memory card access cover and hand tighten captive fasteners.

11. ILLUSTRATED PARTS BREAKDOWN.

12. This illustrated parts breakdown has data required for identifying and ordering parts. The manual introduction has more information on IPB data.



18AC-SCM-00-(8-1)30-SCAN

Figure 1. Test Equipment Hookup Locator (Sheet 1)

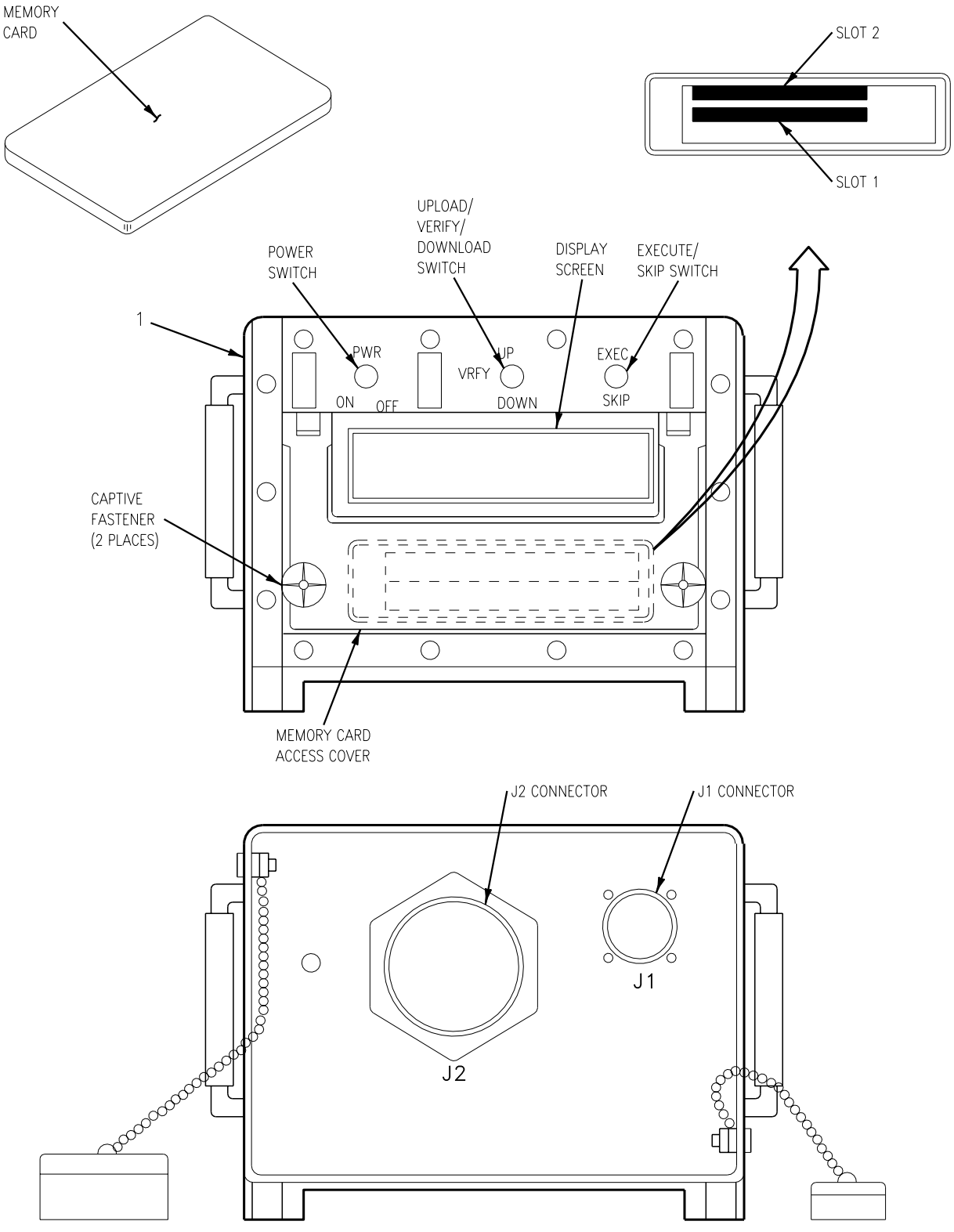


Figure 1. Test Equipment Hookup Locator (Sheet 2)

INDEX NO.	PART NUMBER	DESCRIPTION	UNITS PER ASSY	USE ON CODE	SM&R CODE
		TEST EQUIPMENT HOOKUP LOCATOR			
1	3359AS1000	MEMORY LOADER - VERIFIER SET AN/USQ-131 (30003)	1		PEOGD
1	3359AS853	CABLE SET, MLVS (02387)	1		PEOGG

Figure 1. Test Equipment Hookup Locator (Sheet 3)

ORGANIZATIONAL MAINTENANCE

SOFTWARE CONFIGURATION MANUAL

SCHEMATIC - MUX TEST CONNECTOR (83J-G003) INTERCONNECT

This WP supersedes WP008 00, dated 1 September 1995.

Reference Material

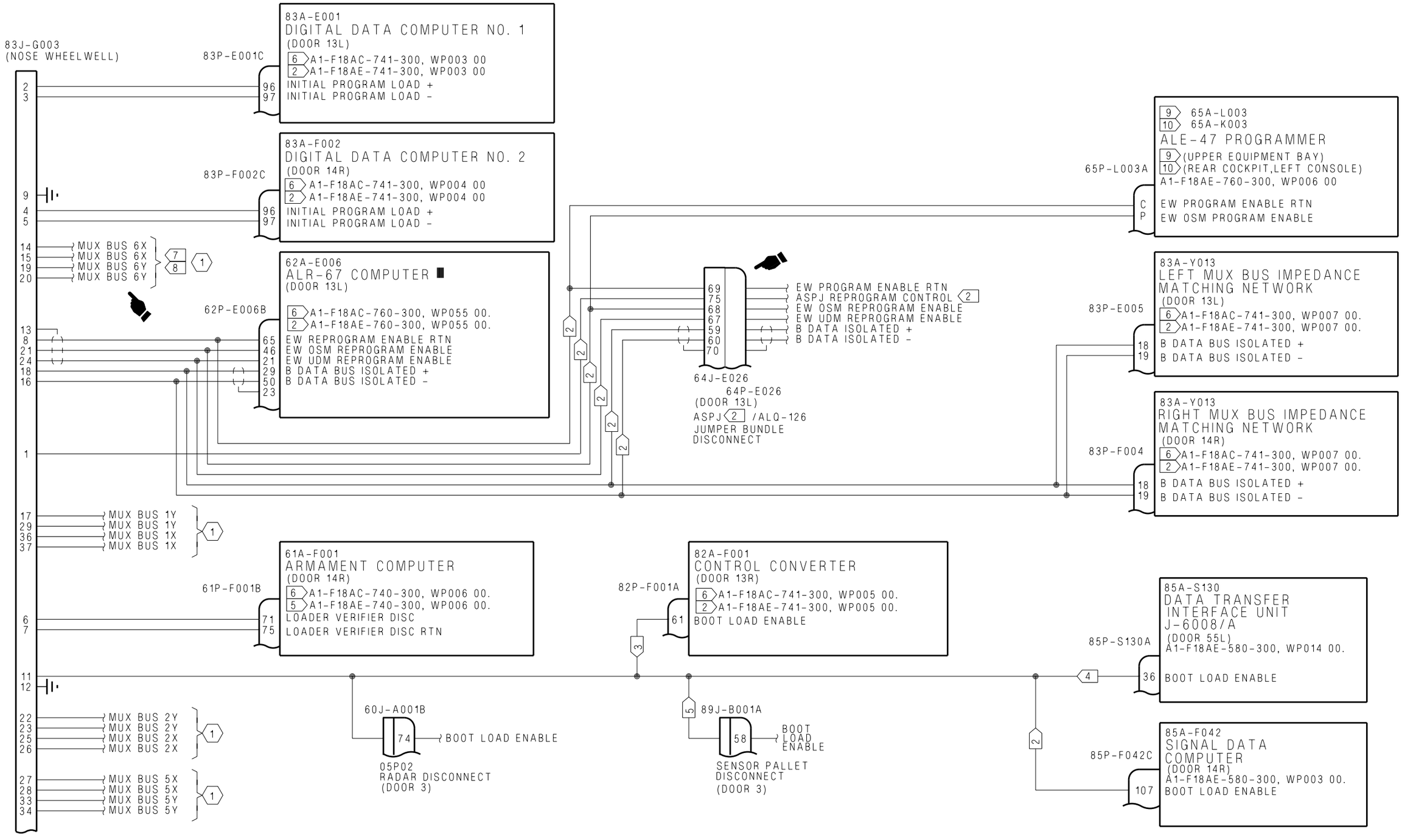
None

Alphabetical Index

Subject	Page No.
MUX Test Connector (83J-G003) Interconnect Schematic, Figure 1	2

Record of Applicable Technical Directives

None



ADA541-7-1-035

Figure 1.

Figure 1. MUX Test Connector (83J-G003) Interconnect Schematic (Sheet 1)

Figure 1.

LEGEND

⟨1⟩ SEE APPLICABLE AVIONIC MUX CHANNEL SCHEMATIC, A1-F18AC-741-500, WP001 00 OR A1-F18AE-741-500, WP001 00.

2 F/A-18C AND F/A-18D.

3 163985 AND UP.

4 164279 AND UP.

5 F/A-18D 164279 AND UP.

6 F/A-18A AND F/A-18B.

7 F/A-18A AND F/A-18B; F/A-18C AND F/A-18D 163427 THRU 164980.

8 165171 AND UP.

9 F/A-18C 165171 AND UP.

10 F/A-18D 165409 AND UP.

11. TO ISOLATE DEFECTIVE AIRCRAFT WIRING, SEE A1-F18A( )-WDM-000.

Figure 1. MUX Test Connector (83J-G003) Interconnect Schematic (Sheet 2)